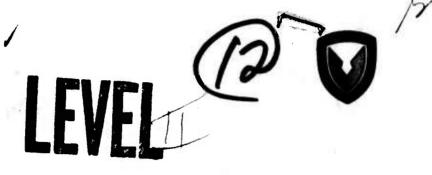
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PRELIMINARY DESIGN STUDY OF A TAIL ROTOR BLADE JETTISON CONCEPT

Robert A. Selleck Sikorsky Aircraft Division of United Technologies Corp. Main Street Stratford, Conn. 06602

JULY 1978

Final Report

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APPLIED TECHNOLOGY LABORATORY U. S. ARMY RESEARCH AND TECHNOLOGY LABORATORIES (AVRADCOM)

Fort Eustis, Va. 23604

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APPLIED TECHNOLOGY LABORATORY POSITION STATEMENT

This report provides the results of a preliminary concept study which indicates that a tail rotor blade jettison system can be developed for four-bladed tail rotor systems, which will allow the controlled jettison of a damaged blade and its opposing blade and allow continued flight with the remaining two blades. A prototype system was developed that evaluated system performance characteristics throughout the maximum/minimum tail rotor speeds anticipated and determined the resultant effects on structural integrity, tail rotor stability, and handling qualities using both analytical and simulation modeling techniques. The analyses and evaluation tests conducted showed that the prototype blade jettison system developed meets the performance requirements of the UH-60A helicopter.

Results of this contractual effort are still preliminary, and additional effort is required to improve and validate the survivable characteristics of the design.

Mr. Harold W. Holland of the Aeronautical Systems Division served as technical monitor for this effort.

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was developed and its performance evaluated to determine the capability of the system to jettison rotor blades in a manner that would prevent secondary damage. Additionally, analyses were conducted to determine the dynamic stability characteristics of the UH-60A tail rotor in a two-bladed configuration and the ability of the helicopter to accommodate the loads developed during transition from four to two blades. Residual helicopter performance and the capability of the helicopter to be retrimmed following jettison of two opposing tail rotor blades was analyzed using the GENERAL HELICOPTER FILCHT DYNAMIC MODEL programmed on a PDP-10 Hybrid Computer. Handling qualities to be expected following the loss of opposing tail rotor blades were examined by integrating the computer model and two-blade subroutine with a flight simulator. The analyses and evaluation tests conducted show that the prototype blade jettison system developed meets the performance requirements of the UH-60A helicopter.

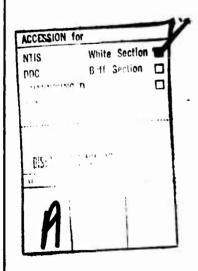


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INTRODUCTION

Loss of a significant portion of a tail rotor blade will result in severe tail rotor imbalance that can lead to secondary damage to the helicopter and/or injury to the occupants. Continued use of the helicopter in hostile environments is anticipated, and with this the possibility of exposure of the helicopter to increased ballistic threat levels beyond the normally survivable 7.62mm projectile. The 14.5nm, 23mm, 30mm, and 37mm high explosive incendiary rounds are capable of inflicting such severe damage that if a tail rotor blade sustains a hit, blade loss can be expected to occur. The development of tail rotor blades that can tolerate the magnitude of damage that these rounds can cause must therefore be substantially larger and will, in all likelihood, have an undesirable increase in centrifugal force resulting in significant weight penalties in the rotor hub, drive shaft, and supporting structure as well as increasing power requirements. This escalation of component size is particularly unsuitable for the small, reconnaissance-type helicopters.

As an alternative to the development of ballistically tolerant blades to prevent rotor imbalance of four-bladed tail rotor systems, the imbalance forces caused by the loss of a part or all of a blade can be overcome by the controlled jettison of the residual portion of the damaged blade and its opposing blade, to allow continued flight with the remaining two blades. The intent of this effort was to develop a working prototype system to evaluate system performance characteristics throughout the maximum/minimum tail rotor speeds anticipated and to determine the resultant effects on structural integrity, tail rotor stability, and handling qualities using both analytical and simulation modeling techniques.

The prototype system relates closely to a concept defined in a study program previously conducted by Sikorsky Aircraft under U. S. Army Contract DAADO5-73-C-0523, where the feasibility of applying the opposing blade jettison concept to the main rotor system of a four-bladed helicopter was determined. The results of the program are included in a report (Reference 1) distributed by USA Ballistic Research Laboratories, Aberdeen Proving Ground, Maryland. In accordance with the requirements of that program, a conceptual design was established for damage detection and controlled rotor blade jettison that is the basis for the prototype design of this present contract. Although the basic concept is essentially the same, many circuitry improvements and refinements have been incorporated.

The prototype development and associated analyses were based on the physical and performance characteristics of the UH-60A BLACK HAWK helicopter (Figure 1), a candidate for initial application of the system. Additionally, complete performance information and simulation programs were readily available for this effort in accordance with the contract. The system design lends itself to kit-type installation, and the electrical portion of the system can be modified for other four-bladed helicopters with only minor changes. Pyro-

J. W. Johnson, R. A. Selleck, Rotor Balance Restoration Study, Sikorsky Aircraft, U. S. Army Ballistic Research Laboratories Contract Report #197, December, 1974.

technic blade severing devices, damage detectors, and supporting hardware must be sized for the particular helicopter application.

During the system design portion of this program, emphasis was placed on maximizing system reliability. To this end, the system employs state-of-the-art components, fully encapsulated assemblies, solid-state design, no moving parts in contact, and the latest design techniques and safeguards to prevent inadvertent system initiation. The capability of the prototype system developed under this contract to effect blade jettison within the jettison window established for the UH-60A helicopter was evaluated by installing the system in an Engineering rotary test stand, capable of incremental rotational speed adjustment from well below to well above the UH-60A design rotor speeds.

Concurrent with prototype system development, the dynamic stability characteristics and handling qualities that can be anticipated for the UH-60A helicopter with two opposing tail rotor blades jettisoned were determined. The dynamic analysis included an investigation of the two-bladed stability, vibration, and rotor hub load levels for speeds to 150 knots. Additionally, the ability of the UH-60A helicopter to structurally accommodate the centrifugal loads from loss of a full tail rotor blade until rebalance by opposing blade jettison was examined. The handling qualities were examined through the use of the UH-60A version of General Helicopter Simulation Program and blade transition subroutines to determine trim characteristics. The pilot work load was determined by integration of the Simulation Program with a flight simulator to obtain qualitative pilotin-the-loop assessment of flight attitude recoverability following transition from the normal four-bladed mode to the emergency two-bladed mode.



Figure 1. UH-60A BLACK HAWK Helicopter Baseline

PART I CONCEPT ANALYSIS

Loss of a significant portion of tail rotor blade is sensed by the interruption of detector circuits located along the leading and trailing edges of the blade spar. Interruption of both of the detector circuits results in initiation of pyrotechnic linear-shaped charges (LSC) located adjacent to the rotor hub that sever both the residual portion of the damaged blade and its opposing blade. The specific point of rotor rotation at which the separation occurs is controlled to prevent secondary damage to the main rotor blades or the vehicle proper.

Successful operation of the system on the UH-60A helicopter requires the analysis of the effects of the system on the aircraft and its occupants both during and after blade jettison has occurred. It is necessary to define an available window for blade jettison that will avoid secondary damage and to determine the residual performance capability of the helicopter following the jettison of the two rotor blades. Included in the analyses is a determination of the resultant vibration levels and loads to be expected and a discussion of the ability of the UH-60A to structurally accommodate the reaction loads during the period of rotor imbalance. This work is presented in the following subtasks.

BLADE SEPARATION WINDOW

Following loss of a portion of all of the damaged rotor blade, rebalance of the rotor can be effected by either simultaneous jettison of the residual portion of the damaged blade and its opposing blade or by sequential jettison of the blades. The latter, sequential, approach is selected for the UH-60A for the following reasons: The available window for simultaneous blade jettison is established by the need for adequate clearance with the main rotor blades for a forward jettisoned blade and the horizontal stabilator for an aftward jettisoned blade. The available window must accommodate the variances in the point of blade jettison due to the variations of component function times and the range of rotor speeds considered for the design of the system. Of greater importance, however, is the potential hazard of a vertically jettisoned tail rotor blade to the helicopter in the event of system operation in a hover, or near hover, flight mode. Selection of a sequential blade jettison capability overcomes both of these difficulties. Figure 2 shows the window established for use in the design of the UH-60A blade jettison system.

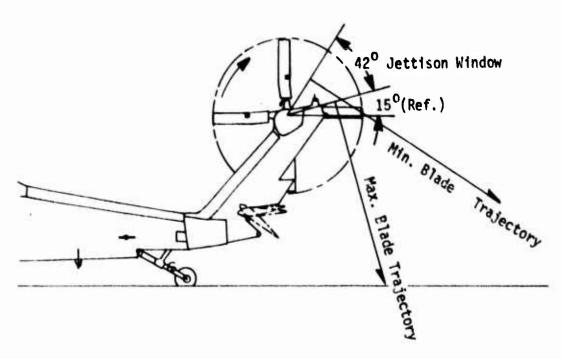


Figure 2. Acceptable Window for Sequential Rotor Blade Jettison

Although the sequential jettison of blades would normally allow a much greater window than that shown in Figure 2, the upper limit has been constrained to minimize any potential hazard that jettisoned blades might present to other aircraft that may be flying formation with the damaged aircraft. Further, the lower window limit is less than that available to reduce the possibility of secondary damage to the remaining rotor blades by a rebounding blade if the system is operated while the aircraft is on, or near the ground. The 42-degree window has been verified by analysis to be adequate for rotor rotations from 70% NR (830 RPM) to 150% NR (1877 RPM). Additionally, the 42-degree window will accommodate a worst case stack-up of component function times at the 150% NR rotor speed.

TWO BLADE STABILITY, VIBRATION, AND LOAD LEVEL

Analytical investigations were performed to evaluate the dynamic stability characteristics of the two-bladed UH-60A cross beam tail rotor, and the vibration and load levels in the aircraft in the forward flight regime up to 150 knots. The analytical results indicate that the two-bladed rotor system mounted on the flexible pylon is stable up to the maximum speed investigated of 150 knots.

A normal modes rotor aeroelastic analysis was employed to calculate the steady and 2 and 4/rev vibratory loads at the hub originating from the two-bladed rotor system. These loads were then used in conjunction with the tail rotor pylon modes to evaluate the vibration levels experienced at the hub and gearbox. It was found that the vibration and load levels

increase with forward speed. At 150 knots the highest speed investigated, the maximum 2/rev vibration levels result in a roll moment at the gearbox-pylon attachment of 20 percent of the ultimate value. This level is acceptable to maintain flight and to land within the required one-half hour design criterion established for the UH-60A helicopter.

DYNAMIC STABILITY ANALYSIS RESULTS

The dynamic stability characteristics of the two-bladed UH-60A tail rotor system were investigated with a Floquet analysis. The analysis uses blade flapping and lead-lag degrees-of-freedom and up to ten air-frame modes. The stiffness, damping, and mass matrices are evaluated for all the blades from the initial conditions specified for blade flap, lead-lag, and pitch motions at various azimuthal positions and integrated for one rotor revolution. The eigenvalues are then calculated and the system stability determined from an inspection of the real part of the eigenvalues. The UH-60A tail rotor blade characteristics and the pylon modes used in the Floquet analysis are summarized in Tables 1 and 2 respectively. The axes sign convention employed throughout this study is shown in Figure 3. The analysis also includes the effect of rotor inflow, blade pitch and twist, and linear aerodynamic characteristics.

Table 1. UH-60A Tail Rotor Blade Characteristics

Parameter	Units	Quantity
Radius	ft	5.5
Radial Location Where Blade Bending Starts	ft	0.3333
Outboard Blade Chord	ft	0.8125
Number of Blades	-	2.0
Rotor Speed (100% N _R)	rpm	1215.0
Weight of One Blade	1bs	19.0
Blade First Inertia Moment	slug-ft	1.2332
Blade Second Inertia Moment	slug-ft ²	4.1266
Structural Damping	percent	0.50
Equivalent Linear Twist (from center of rotation to blade tip)	deg	- 20.0
Outboard Blade Airfoil Section	-	SC-1095
Tip Loss Factor	•	0.97
First Flatwise Frequency/Rotor Speed	-	1.152
First Edgewise Frequency/Rotor Speed	-	1.696
Air Density	slug/ft ³	0.002175
Speed of Sound	ft/sec	1102.0
Pitch-flap Coupling, δ_3	deg	35.0
Airfoil Lift Curve Slope	-	6.30
Airfoil Drag Coefficient	-	0.007

Table 2. Tail Rotor Pylon Modes Without Axial Modal Components

Mode	MG	ωG	$\frac{\zeta_{G}}{}$	ϕ_{x}	ϕ_y	$\phi_{\mathbf{e}\mathbf{x}}$	$\phi_{m{\theta}}$
No.	lb sec ² /in	rad/sec	%	in/in	in/in	1/in	1/in
1.	1.3146	41.15	4.5	1.0	.2105	0.	017
2.	1.1255	198.13	3.6	285	1.0	095	068
3.	9.9661	395.42	1.8	1.0	.7714	0.	.0

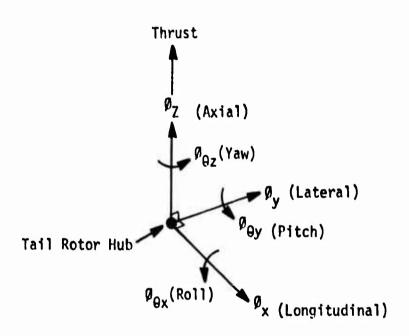


Figure 3. Axes Sign Convention

The results from the Floquet analysis are presented in Figures 4 through 6, for a range in forward speeds from zero to 150 knots. From Figure 4 it is seen that all rotating system (blade) and fixed system (tail pylon) modes are stable in the speed range investigated. The least stable modes are the blade lead-lag mode and the third fixed system mode, both showing a damping level of about one-half percent. The effect of forward speed is not significant except near 150 knots when two of the fixed system modes indicate a degradation in damping level. It is noted that the response of the two blades is not identical due to the presence of the fixed system modes and their interaction with each blade. Exclusion of

the fixed system modes from the analysis simulates a completely rigid hub.

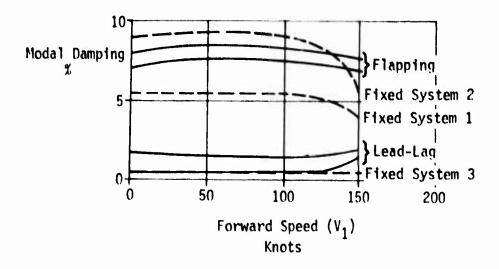


Figure 4. Modal Damping Variation with Forward Speed Including Tail Rotor Hub Flexibility

The damping associated with the blade flapping and lead-lag motions without tail rotor hub flexibility is presented in Figure 5. The results indicate that the blade lead-lag damping is lowered slightly by the flexibility of the hub while the flapping mode shows very little change in stability. Each blade now behaves exactly the same.

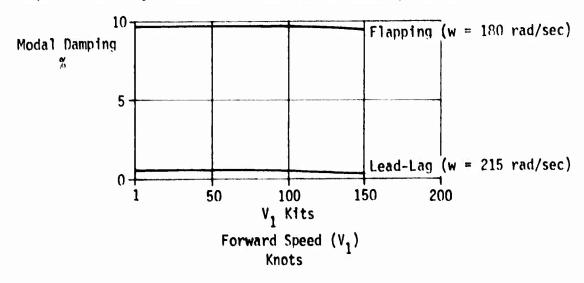


Figure 5. Modal Damping Variation with Forward Speed without Tail Rotor Hub Flexibility

The frequency of the blade and fixed system modes with forward speed is illustrated in Figure 6. This figure shows that the modal frequency is not influenced significantly by forward speed. The blade flapping mode and one of the fixed system modes have frequencies close to each other, especially at 150 knots. The coupling between these two modes results in the damping degradation seen in Figure 4. The frequencies of the blade flapping and lead-lag motions without tail rotor hub flexibility are 180 (1.42/rev) and 215 (1.69/rev) rad/sec, respectively. The increase in the blade flapping frequency over the uncoupled value of 1.152/rev given in Table 1 is due to the pitch-flap coupling of 35 degrees.

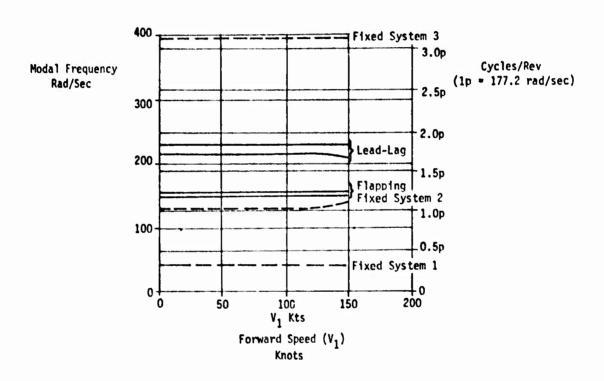


Figure 6. Modal Frequency Variation with Forward Speed Including Tail Rotor Hub Flexibility

TAIL ROTOR HUB LOADS AND VIBRATION RESULTS

The aeroelastic behavior of the two-bladed UH-60A tail rotor was investigated in hover and forward speeds up to 150 knots using the time history analysis discussed in Reference 2. The analysis describes the aeroelastic response of the rotor blade by a "normal modes" technique. Five rotor blade degrees-of-freedom (three flatwise, one edgewise, and one torsional) are employed to describe the aeroelastic characteristics

Arcidiacono, P. J., Prediction of Rotor Instability at High Forward Speeds, Volume I; Sikorsky Aircraft; USAAVLABS Technical Report 68-18A, U. S. Army Aviation Materiel Laboratories, Fort Eustis, Virginia, February, 1969, AD-685860.

of the rotor system. Once the rotor flight condition is prescribed (as given by blade collective pitch input, rotor inflow, and thrust), the integration of the equations of motion proceeds around the rotor azimuth at specified intervals for a number of rotor revolutions. The displacements and velocities of all blade modes are then checked at the beginning and at the end of each complete rotor revolution. For a stable condition, this procedure usually takes up to ten rotor revolutions before the modal displacements and velocities repeat themselves within a specified tolerance for a "converged" solution. Once convergence has been reached, the rotor hub loads can be calculated both in a rotating axis system and in a fixed axis system. The hub shears and moments are then harmonically analyzed. For a two-bladed rotor, the hub loads present in the fixed axis system are the steady loads and are at frequencies that are multiples of twice the rotor speed.

The tail rotor thrust and collective pitch at the 75-percent radial location are presented as a function of forward speed in Figure 7. The thrust-pitch relation is consistent with the main rotor torque requirements for level flight operation. It should be noted that at forward speeds greater than 120 knots, the tail rotor operates increasingly in the blade stall region, resulting in degradation of rotor performance and substantial increases in rotor hub loads.

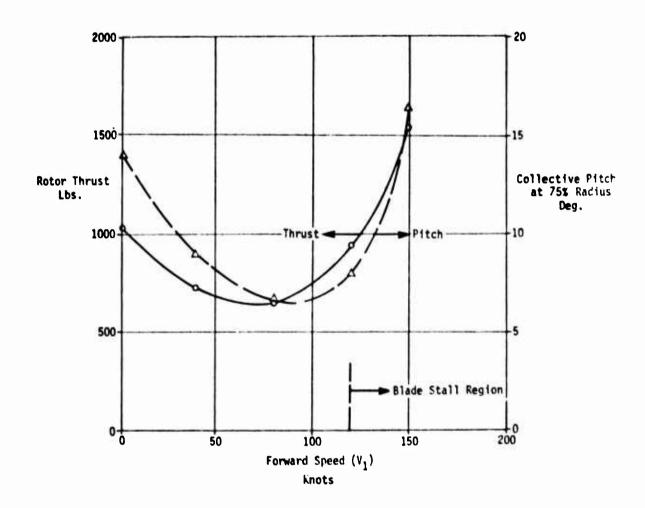


Figure 7. Tail Rotor Thrust and Collective Pitch Variations with Forward Speed

The steady and 2 and 4/rev vibratory hub shears (Figures 8, 9, and 10) and moments (Figures 11, 12, and 13) in the fixed system have been calculated for the forward speed range up to 150 knots. Higher harmonics are not presented since they are small in comparison to the 2 and 4/rev components. The positive directions of the hub shears and moments were previously illustrated in Figure 3. The results presented in Figures 8 through 13 indicate that the rotor hub loads generally increase with

forward speed. All shears and moments except the steady yaw moment increase rapidly at forward speeds greater than 120 knots as the rotor operates increasingly in the blade stall environment.

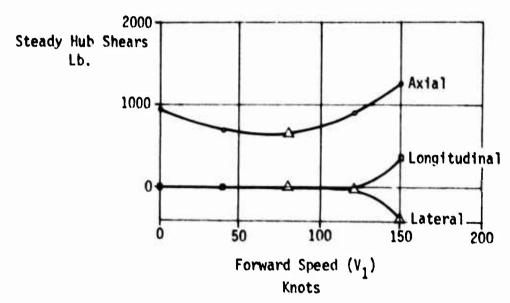


Figure 8. Tail Rotor Hub Shear Load Variations with Forward Speed (Steady Component)

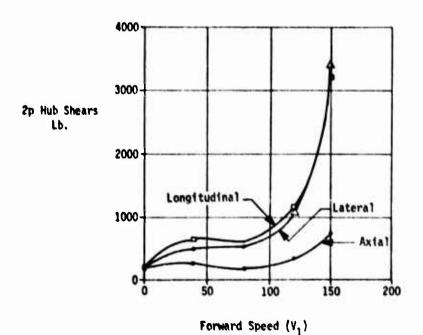


Figure 9. Tail Rotor Hub Shear Load Variations with Forward Speed (2 Cycles/Rev Component)

Knots

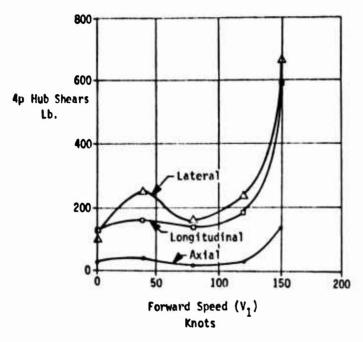


Figure 10. Tail Rotor Hub Shear Load Variations with Forward Speed (4 Cycles/Rev Component)

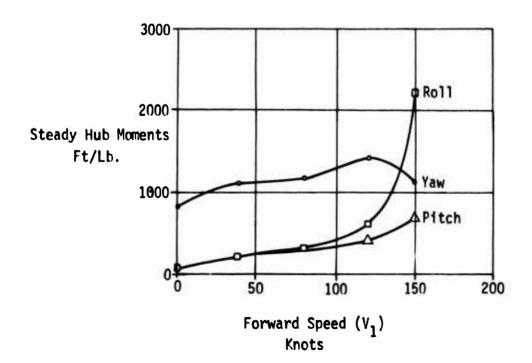


Figure 11. Tail Rotor Hub Moment Variations with Forward Speed (Steady Component)

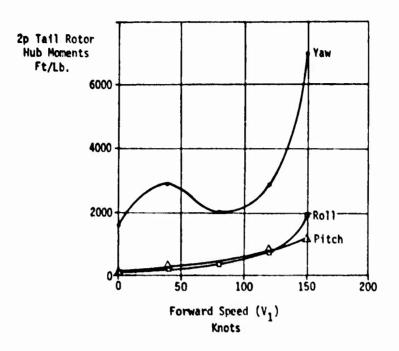
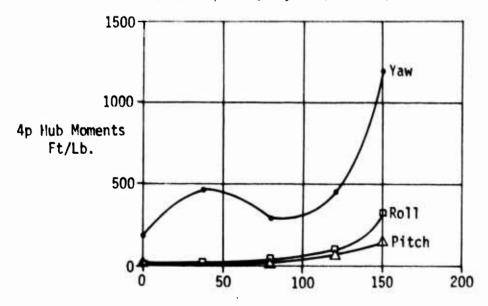


Figure 12. Tail Rotor Hub Moment Variations with Forward Speed (2 Cycles/Rev Component)



Forward Speed (V₁)
Knots

Figure 13. Tail Rotor Hub Moment Variations with Forward Speed (4 Cycles/Rev Component)

The vibratory hub loads are used in conjunction with the tail rotor pylon modes (Reference 3) presented in Table 3 to calculate the accelerations present at the tail rotor hub and gearbox.

Table 3. UH-60A Tail Rotor Pylon Modes Including Axial Modal Components

Mode	MG	^ω G	$\frac{\zeta_{G}}{}$	$\frac{\phi_{x}}{}$	φ _y _	$\frac{\phi_z}{\phi_{\text{ex}}}$	ϕ_{ey}
No.	lb sec ² /in	rad/sec	A)	in/in	in/in	in/in 1/in	1/in
1.	0.400	38.32	4.5	.264	.056	1.0 0.	004
2.	1.438	182.20	3.6	285	1.0	519095	068
3.	11.980	389.53	1.8	1.167	0.9	1.0 0.	0.

Three linear and two rotational accelerations are computed and plotted in Figures 14 and 15 for harmonic loads of two and four times the rotor speed respectively. The accelerations generally increase with forward speed as expected from the behavior exhibited by the hub shears and moments from Figures 8 through 13. Significant increases in hub and gearbox accelerations are shown for forward speeds greater than 120 knots. It is noted that the 2/rev accelerations are much higher than the 4/rev accelerations.

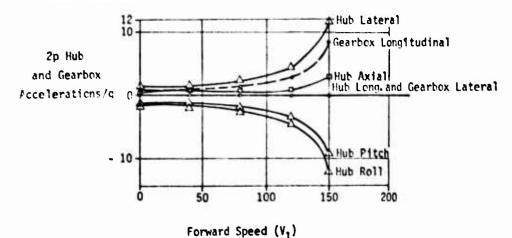


Figure 14. Tail Rotor Hub and Gearbox Accelerations with Forward Speed (2 Cycles/Rev Component)

Knots

UTTAS Aeroelastic Stability Analysis, Sikorsky Aircraft, SER-70545, Revision 2, May 1978.

(b) Cycles/Rev Component

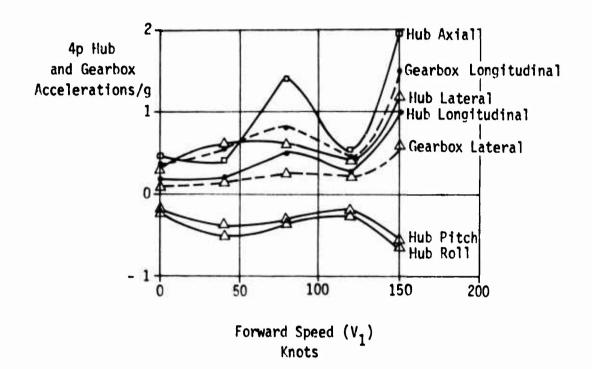


Figure 15. Tail Rotor Hub and Gearbox Accelerations with Forward Speed (4 Cycles/Rev Component)

The accelerations from Figures 14 and 15 are used to calculate the roll moment experienced at the gearbox-pylon attachment, which is a critical stress region. The steady and 2 and 4/rev roll moments at the gearbox-pylon attachment are shown in Figure 16. The highest loaded conditions occur at the maximum forward speed investigated, 150 knots. The 2/rev vibratory response is much greater than the 4/rev response. When added to the steady roll moment, the total 2/rev moment at 150 knots is approximately 20 percent of the ultimate roll moment value of 13,750 foot-pounds. This load level is acceptable to maintain a level flight operation and to conduct a safe landing to meet the required one-half hour criterion.

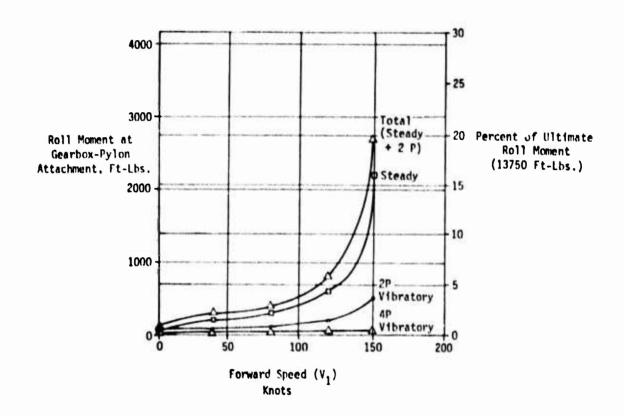


Figure 16. Variation of Roll Moment at Gearbox-Pylon Attachment with Forward Speed

STRUCTURAL ACCOMMODATION

The ability of the UH-60A helicopter to structurally accommodate the load caused by tail rotor blade loss has been examined. The analysis is based on the assumption that ballistic damage is limited to the complete removal of the outer 90% of one rotor blade or that area outboard of the back-to-back hub plates and that the load application time is limited to 360 degrees by removal of the opposing rotor blade. The effect of the centrifugal load on the empennage was examined using the UH-60A NASTRAN Structural Analysis Program to determine the deflections of the center of the rotor that can be expected to occur. Figure 17

shows a trace of the excursions of the rotor center through 360 degrees of the load application. With an imbalance load applied in the manner shown in the figure, the trace indicates that the rotor hub center can be expected to deflect approximately 2.2 inches rearward and 5.8 inches forward.

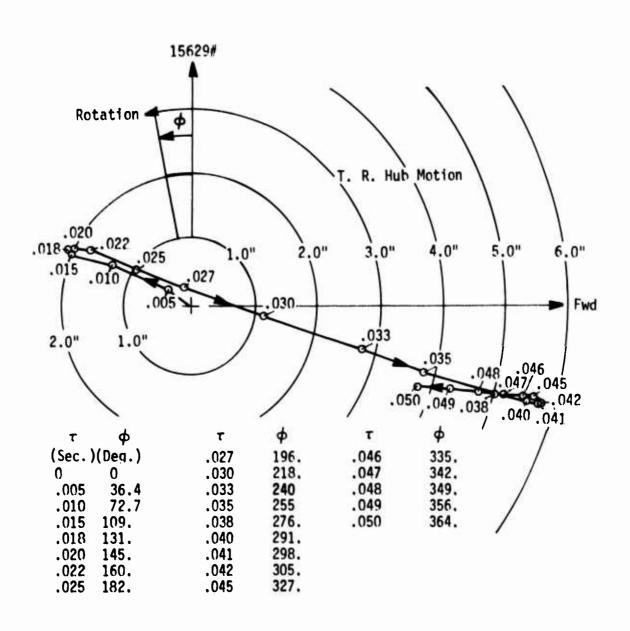


Figure 17. NASTRAN Plot of Tail Rotor Center Excursions

Due to the offset of the rotor hub center from the centerline of the tail cone and pylon, the unbalanced centrifugal load induces both torsion and vertical bending in the pylon and torsion, vertical, and horizontal bending in the tail cone. Examination of the analytical data verifies that these structural deflections all contribute to the motions of the rotor hub center. Quantifying the relative contributions of each of these structural deflections is beyond the scope of the contract. Preliminary examination indicates that the basic tail cone and pylon structure can sustain the unbalanced load over 360 degrees of rotor rotation.

The ability of the tail rotor drive shaft and tail rotor gearbox to accommodate the high centrifugal loads that accompany full blade loss has been examined. Comparing the centrifugal load against the design allowables that were established for these components of the UH-60A helicopter, the loads are expected to exceed the design limits, requiring redesign to accommodate full blade loss. It has been initially determined that the tail gearbox output can be sufficiently strengthened by a material change accompanied by the addition of a process change that will provide the necessary surface hardening characteristics. Further, it is expected that additional strength in the gearbox housing and local pylon structure can be incorporated to provide the necessary structural capability.

PITCH LINK ACCOMMODATION

Severing the tail rotor blade spar must be accompanied by severance of the pitch horn or link as well. The level of centrifugal load applied to the link attachment is a function of the weight of the blade remaining after ballistic damage has occurred. If the point of ballistic impact is well outboard on the rotor blade, the residual centrifugal force would be sufficient to effect automatic separation of the link as well. However, in the event that the point of ballistic damage occurred immediately outboard of the pitch horn, the resulting level of centrifugal force would be insufficient to be confident that the separation of the pitch horn would not be delayed causing the trajectory of the pitch horn to be toward the helicopter proper.

PART II SYSTEM DESIGN

The automatic tail rotor blade jettison system is designed to effect removal of the residual portion of a severed blade and, sequentially, its opposing blade within one full rotation of the rotor system. The system confines blade jettison to an aftward/downward direction to avoid possible secondary damage to the helicopter proper or its occupants. The system is fully solid-state to achieve maximum operational reliability and employs state-of-the-art techniques to prevent inadvertent actuation.

The system is comprised of four basic elements: an electrical power transfer to the rotor system, damage detectors that initiate system operation, a logic system that determines the blade pair to be jettisoned and provides means to control the direction of blade separation, and pyrotechnic devices that sever the blade spar and associated pitch horn for blade jettison. Figure 18 is a block diagram of the system showing the interrelationship of the basic elements. The prototype system designed for fabrication and evaluation testing differed from a fully productionized configuration in several respects. Development of the pyrotechnic devices was not required, allowing evaluation tests to be performed using an engineering-type rotary test stand previously developed for other programs. As a further expedient of the evaluation tests to be performed on the system, manual switching was incorporated to allow selectable faulting of the two portions of the logic circuitry.

The prototype system has been designed to be compatible with the electrical system, rotor system, tail rotor gearbox, and deicing kit of the UH-60A.

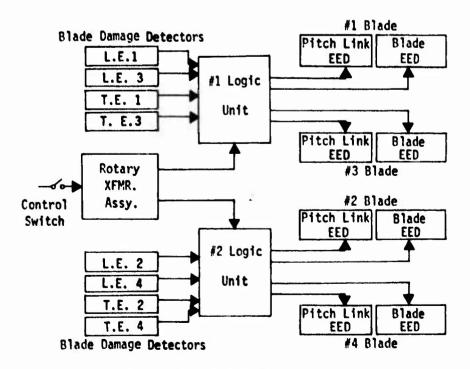


Figure 18. System Block Diagram

Figure 19 is the circuit diagram for the automatic blade jettison system. The diagram indicated the four basic elements of the system and the components that comprise those portions.

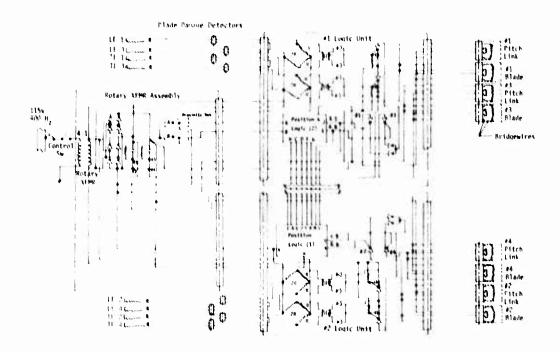


Figure 19. Tail Rotor Balance Restoration System Schematic.

ELECTRICAL POWER TRANSFER - Electrical power to operate the automatic blade jettison system is transferred from the nonrotating aircraft structure to the rotating rotor system by means of a rotary transformer. The transformer is comprised of two subassemblies: an inner frame with the primary windings of the transformer that mounts on the tail qearbox housing and an outer frame with secondary windings that is installed to rotate with the rotor system. The transformer is designed to accept 115 vac, 400 Hz aircraft power and to provide an output of 24 vdc that allows use of aircraft-type hardware. Unlike conventional transformers, the rotary transformer is designed to operate even though the components are moving with respect to each other and efficiency of the transformer's output is directly related to the narrowness of the gap between the induction plates of the two halves of the transformer. Although the output level of the rotary transformer is normally lower than that achievable with the conventional transformer, the output is more than adequate for the intended application.

The rotary transformer was selected as the electrical transfer device in lieu of the more conventional collector ring approach to minimize the requirement for field level maintenance. This not only reduces the operating costs but, perhaps more importantly, minimizes the possibility of field maintenance-induced malfunctions.

The 13-inch transformer for the UH-60A blade jettison system (see Figure 20) is the largest transformer of its type designed to date, the size of the device being dictated by a need for compatibility with the rotor blade deicing kit of the UH-60A. The transformer (part number 207073) was designed to meet Sikorsky performance requirements by Superior Electric Company of Bristol, Connecticut. Two proximity switches are assembled to the rotary transformer that gate the power output to the blade severing charges to control the direction of blade separation. These switches are installed with the transformer as a convenient and practical location to relate the gating function to rotor rotation and to maintain a kit-type approach for the installation of the blade jettison system. The proximity switches operate 90 degrees out of phase with respect to each other. Each switch is associated with one pair of opposing blade paddles. Each proximity sensor transitions from "closed" to "open" or the reverse as determined by the placement of metal targets that are installed in accordance with the blade jettison window requirement. The direction of blade jettison can easily be controlled by this design approach.

The proximity switches selected for the system are fully qualified with a mean-time-between-failures of 200,000 hours. The device, manufactured by Eldec Corporation of Lynnwood, Washington (part number SCD8-260), has a switching rate capability of greater than 20,000 cycles per minute and has an operating temperature range of -65° to +180°F.

Included also as a part of the rotary transformer assembly is power rectification and charge storage capacitors that provide sufficient output to initiate the electro-explosive detonators that, in turn, initiate the blade severing devices. The relatively high power requirement to

achieve reliable initiation of the four EED's for jettison of the blade pair is achieved by capacitance discharge. Two storage capacitors are installed that charge following arming of the system by the pilot.

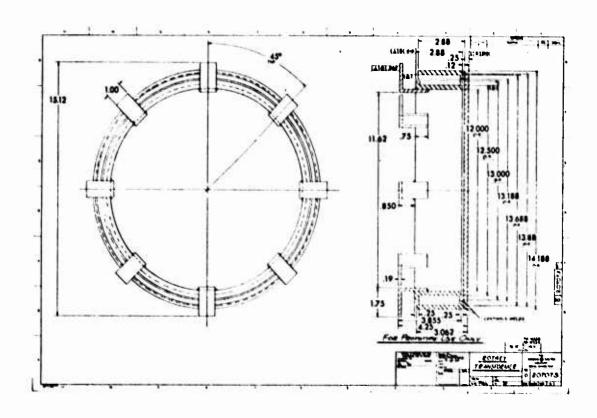


Figure 20. Prototype Rotary Transformer Fully Compatible with the UH-60A Tail Rotor Gearbox and Deicing Unit

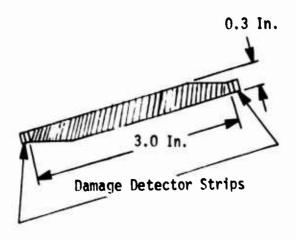
DAMAGE DETECTORS - Two sets of detectors are employed with each blade paddle to detect blade spar damage. Specifically, they are intended, by the manner in which they are installed, to differentiate between ballistic damage to the blade that is of a tolerable nature and that which will result in an intolerable rotor imbalance. This is achieved by locating independent detectors along both the leading edge and trailing edge of the blade spar. The logic system is designed to require an input from both detectors in order to activate the blade severing pyrotechnic devices. In this way, ballistic (or other) damage that does not cause spar separation will not result in blade jettison.

The detector assembly for the UH-60A is comprised of two thin strips of graphite epoxy, the same material from which the spars are fabricated, to obtain the same physical characteristics and thereby avoid differential elongations due to temperature and centrifugal force. The two halves of the detector assembly are insulated from each other and bonded together. For production installations, the detector assemblies are installed by bonding in position along the edges of the spar prior to application of the outer fiberglass wrap. For test applications where blade retrofit is required, the detector assemblies are potted in position along the spar edges following preparation of the blade by longitudinal saw cuts to a depth that is sufficient to expose the spar edges.

The electrical properties of the graphite epoxy allow the use of the material as a conductive element having a useful resistance level for the intended application. The detector assembly is essentially a circuit that serves to balance a conventional wheatstone bridge. As long as the circuit's resistance level remains within acceptable tolerance limits, the logic system remains inactive with regard to blade separation initiation.

In the event that the detector assembly circuitry is interrupted or the resistance level becomes significantly changed beyond the established tolerance limits, the bridge becomes unbalanced and the logic unit "senses" a fault. When both circuits exhibit faults, the logic unit has "sensed" a failed spar condition and blade jettison is initiated. In order to effect a positive indication of resistance level change, the detector assembly includes the installation of a resistor at the outer end of the spar; circuit interruption then results in significant resistance level change. The resistor further serves to allow tuning of the circuit, if required. The inner end of the detector assembly terminates in a small potted end fitting that has posts for attachment of the wires leading to the logic unit. This connection is located under the rubber boot near the root of the blade and is provided to allow rapid rotor blade change.

Figure 21 shows the general arrangement of the detector assembly. Detail design of the detector assembly was not required for the evaluation test work and was beyond the scope of the contract.



Graphite Epoxy Spar Cross Section

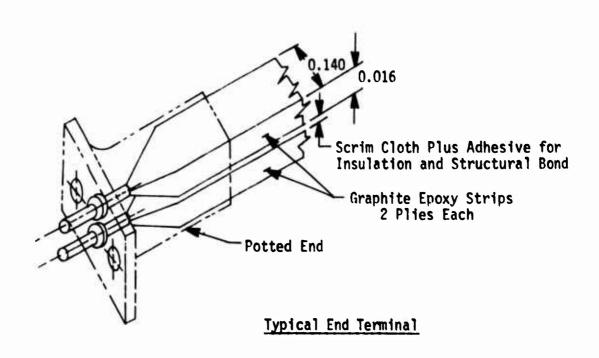
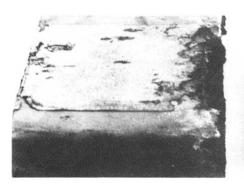


Figure 21. Damage Detector Assemblies (Fabricated from Spar Material for Elongation Compatibility)

LOGIC UNIT - This unit is comprised of a number of switches (normally open) that gate the power output to the proximity sensors. These series oriented switches change from "Open" to "Closed" when the associated detector assembly faults and its wheatstone bridge becomes unbalanced. Silicon controlled rectifiers (SCR) were selected to perform the switching function to achieve highly reliable, solid-state performance. The logic unit also includes position logic circuitry that allows two proximity sensors to perform the timing function for the entire rotor. This is achieved by means of two additional SCR's that switch over the proximity sensors to the affected blade pair. Also included in the logic unit are the remaining three legs of the wheatstone bridge. Two logic units are required for the four-bladed UH-60A tail rotor system.

ROTOR BLADE SEVERANCE - Pyrotechnic severing devices are installed at the root of each rotor blade and on the pitch horn to effect jettisoning of the blades. The devices are both comprised of linear shaped charges supported in silicon rubber and installed in fiberglass housings. The spar severing devices are installed adjacent to the spar, immediately outboard of the rotor hub plates under the existing rubber boot of the blade. The pitch horn severing devices are installed at the horn's smaller, outer end and are nested in the cavity of the component.

The development of the actual blade and pitch severing devices was not required by this contract. Certain preliminary work has been performed which verifies that the UH-60A graphite epoxy spar can be severed by a linear shaped charge of acceptable charge size to be compatible with the intended application. Figure 22 is a photograph of a segment of graphite epoxy spar assembly of the UH-60A material type and thickness that was successfully severed by 200 grains per foot charge loading in the cross-grain manner required for a blade separation. This test performed by Teledyne McCormick Selph, was a preliminary effort only to ascertain feasibility of severing graphite epoxy spars by pyrotechnic means and no attempt was made to optimize the charge size. It is estimated that the production severing charge would be lead-sheathed and would fall in the size range of 125 to 150 grains per foot using RDX (Cyclotrimethylenetrinitramine).



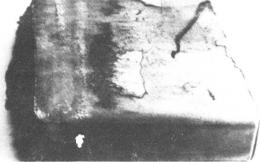


Figure 22. Graphite Epoxy Spar Severed by Linear-Shaped Charge

The chevron-shaped LSC's of both the blade severance assembly and the pitch horn severance assembly are initiated by electro-explosive detonators (EED) that are installed in the housings. Two approaches are considered to be acceptable for the severance assemblies. The first employs a single dual bridgewire EED for LSC initiation and achieves redundant initiation by means of the duality of the EED's bridgewires. The other approach, expected to be superior in terms of severance reliability, employs two individual EED's, each having single bridgewires. The EED's are physically located at opposite ends of the LSC to maximize the redundancy capability.

The blade and pitch horn severance assemblies are initiated simultaneously by the electrical power output from the logic unit. Quick disconnects are provided to accommodate blade removal/installation. The built-in test capability of the system allows preflight verification of circuit continuity up to, and including the bridgewires of the EED's to avoid a maintenance-induced error during blade replacement.

Figure 23 shows the general arrangement of the various components of the system installed on the UH-60A tail rotor.

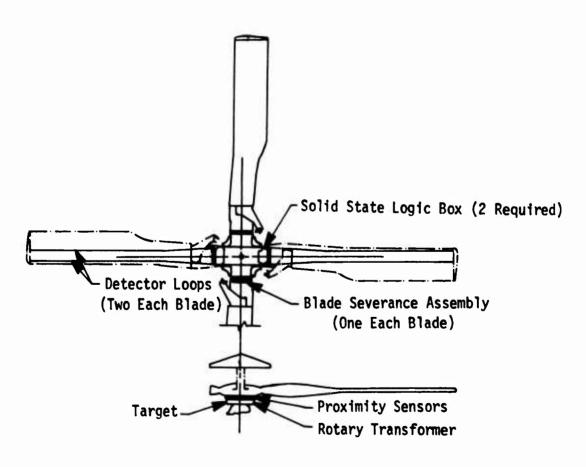


Figure 23. General Arrangement - Tail Rotor Blade Jettison Concept

SYSTEM SAFETY AND RELIABILITY CHARACTERISTICS - The following general discussion pertains to the characteristics that are, or can be, incorporated into a production version of the automatic blade jettison system that enhance reliability of operation and that enhance safety with use of the system.

RELIABILITY

- . Electrical components are selected to be working at approximately 50% of their rating. This practice was not employed during prototype development in order to allow any possible problem areas to surface.
- . Entire system circuitry can be preflight checked to verify operational status. All test functions, however, are removed from the airborne system in order to minimize system complexity and part count.

SAFETY

- . The system incorporates three SCR interlocks ahead of the firing of the EED's to prevent inadvertent blade jettison without prior loss of a blade or blade segment.
- . A control switch is employed to allow the system to be armed at the discretion of the pilot.
- . The storage capacitors are subjected to intentional discharge by bleed resistors following shutdown to maximize safety during performance of maintenance.

IMPACT DAMAGE

- . The logic units are fully encapsulated.
- . The rotary transformer assembly employs encapsulation.
- The blade damage detector circuits and terminal resistors are buried within the leading and trailing edges of the blades.
- The pyrotechnic linear shaped charges are insensitive to impact and are located such that the possibility of impact damage is remote.

HEAT DAMAGE

All components selected are compatible with the temperature requirements of the UH-60A.

LIGHTNING

. Tail rotor blades are covered with aluminum mesh that is grounded to the airframe. Lightning strikes will be conducted on the blade surface and not through the graphite epoxy detectors.

MAINTENANCE

- . The EED's are disarmed during maintenance when the test box harness is installed to prevent inadvertent system initiation during maintenance of the system.
- . Power levels employed to conduct circuit continuity checks are well below the level required to initiate the EED.
- . System design is such that two ground faults are required before the system can actuate; ground faults are detected in the preflight test box.

STATIC ELECTRICAL DISCHARGE

- . The aircraft has adequate static discharge wicks on the stabilator trailing edges to prevent static charge buildup.
- . Any small corona discharges that may develop from the tail rotor blades are insufficient to set off the blade severance system.

STRAY VOLTAGE OR INDUCED CURRENTS

- . All SCR's incorporate anode/gate shunt capacitors to prevent inadvertent change of switch state.
- . Wheatstone bridge resistors are shielded to prevent inadvertent change of switch state.
- . The blade severance system wiring is separated to the maximum extent possible from the tail rotor blade deicing system.

PART III SYSTEM FABRICATION AND EVALUATION

System performance evaluation has been completed on a prototype fabricated in accordance with the schematic shown in Figure 19. The prototype system includes a rotary transformer, electrical power rectifiers, two logic units, two proximity sensors and targets, and fuses to simulate initiation of EED's. The system evaluated includes a capacitance discharge concept to initiate the fuses that provide the required indications to verify proper system performance and timing.

Figure 24 shows the various components of the system with the covers removed. For the test program, the electronic components are attached to a mounting plate that secures to the rotating portion of the rotary transformer. The fuses used to provide initiation indication are installed in a light assembly at the four quadrants that represent the associated blade positions. A switch box is visible that allows either logic unit to be independently selected for testing. The slip ring assembly located at the center of the system is included as test equipment to transmit the system initiating faults and to pick off output signals that verify positional relationship of the initiated components during the tests.

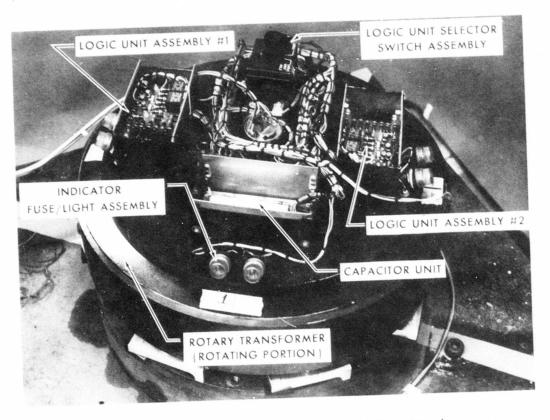


Figure 24. Prototype System Installed in Test Stand

The prototype system was installed on an engineering test stand previously developed for similar test work and upgraded to provide the high-speed capability consistent with a tail rotor application. The stand, shown in Figure 25, includes a protective drum around the upper area for safety reasons during the test operations. The test stand is belt-driven by a variable speed motor capable of rotating the system to speeds up to 1800 RPM. Also visible in Figure 25 is a small control panel containing the switches required to control the test stand drive motor to control power to the rotary transformer, and to introduce the faults that simulate interruption of the damage detectors.

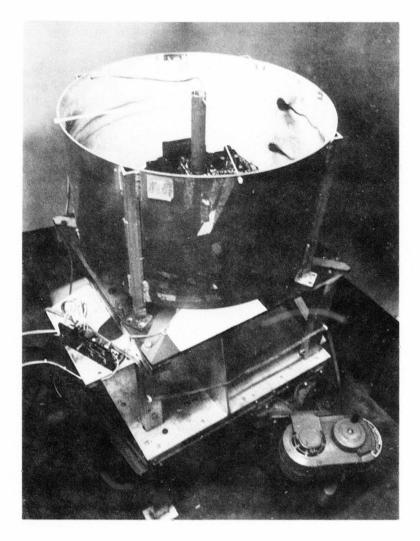


Figure 25. Engineering Rotary Test Stand used for Evaluation of Prototype System Performance Capability

EVALUATION TESTING - Tests were run to determine the positional relationship of certain components of the prototype system in order to verify that the system meets the jettison window requirements of the UH-60A helicopter. This was accomplished by displaying EED initiation (simulated by the use of fuses) and the open/closed status of the proximity switches using a Techtronix 654 scope with 4 trace memory. This approach allows the measurement of the time lapse between the change in state of the proximity sensor and the initiation of the EED. Further, the display provided visual verification that both blades would be jettisoned at the proper location as controlled by the location of the target. It should be noted that the EED's are initiated at a change in state of the proximity switch and the data recorded, therefore, indicates initiation only at the point where the proximity switch changes from open to closed state or the reverse.

In accordance with the system performance test plan, 28 test runs were performed using the engineering rotary test stand to verify proper prototype performance throughout the rotational speed range of 70% NR to 150% NR in accordance with:

%N R	Equivalent RPM	Number of Tests
70	834	2
80	954	2
90	1073	2
100	1192	12
110	1311	2
120	1430	2
130	1550	2
140	1669	2
150	1788	2

The prototype system successfully met the performance requirements in all tests performed, and compatibility with the UH-60A window for tail rotor blade jettison has been verified. Due to the repetitive nature of the data obtained, only the data for the 100% NR run is included here (see Figure 26); the remainder of the data is included as Appendix A to this report. Figure 26 is the photographic record of the Techtronix scope with the various elements defined. By using this data recording approach, it was possible to record data with the required degree of accuracy.

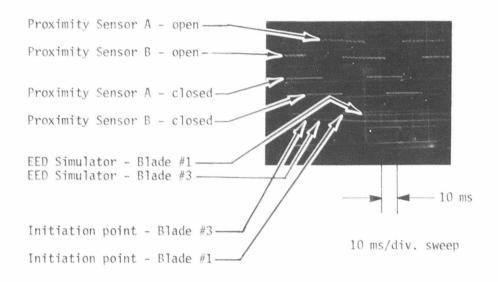


Figure 26. Prototype Performance Data - 100% NR (1192 RPM)

The testing performed in this manner confirmed that the prototype system components were operating properly, in correct sequence, and with sufficient speed to effect blade jettison within the window tolerance limits established. An additional test was needed to verify that the severance signal occurred at the proper rotational position with respect to the location of the window. This was verified by the use of a strobe light, triggered by the output signal from the appropriate proximity switch. Since the oscilloscope record verified that the response times of the components were proper, positional accuracy of the system is a function of the response time of the proximity sensor. By utilizing the output of the proximity sensor to trigger the strobe light, rotational indexing and the output variance throughout the speed range of 834 RPM to 1788 RPM could be determined. Using this technique, it was determined that the positional characteristics of the system were well within the tolerance limits of the UH-60A blade jettison window and that the point of blade jettison varies less than six degrees throughout the entire speed range. Since the 150% NR speed is well beyond the normal operating limits of the UH-60A helicopter, the variation is considered of negligible significance.

PART IV FLIGHT SIMULATION ANALYSIS

An extensive simulation analysis program was conducted to determine the effects that the programmed blade jettison can induce on the performance characteristics of the UH-60A helicopter and to assess the handling qualities to be expected. The Sikorsky General Helicopter (GEN HEL) UH-60A flight dynamic simulation program and PDP-10 Hybrid Computer were employed to study the changes in helicopter trim conditions. By coupling a new subroutine for the transitional, four, to three, to two-bladed rotor condition with the basic GEN HEL Program, it was possible to ascertain the effect of the imbalance loads on the helicopter. To ascertain the change in handling qualities to be anticipated with the blade jettison, the GEN HEL Program was interfaced with the Sikorsky cockpit simulator for qualitative pilot assessment in the normal four-blade mode followed by transition to the two-blade mode.

METHODOLOGY

The tail rotor blade severance scenario was considered as four consecutive, but distinct, handling qualities situations and the GEN HEL Program was employed to study all four of these situations:

- 1. Helicopter trim with four tail rotor blades
- 2. Severance transition from four-to three-to two tail rotor blades
- 3. Pilot response to loss of two opposing blades
- 4. Helicopter trim with two rotor blades, if possible

The basic GEN HEL Program was used to obtain UH-60A trim conditions with either four or two tail rotor blades for flight conditions and loading configurations that represent the most critical - but realistic - situations for tail rotor blade loss. A subroutine was written to program, in detail, the loads at the tail rotor during the imbalance period. This subroutine was coupled with the GEN HEL Program to produce the proper interplay between the response of the helicopter and the loads generated at the tail rotor. The UH-60A GEN HEL Program was interfaced with the cockpit simulator to determine how well the aircraft could be recovered from the loss of two opposing tail rotor blades and the flight conditions that would have to be assumed to continue flight following initial recovery from the helicopter's reaction to tail rotor blade loss. Table 4 presents the axis system, parametric definition, and sign conventions used in the program.

BASIC PROGRAM

The GEN HEL Program is an analytic helicopter model developed by Sikorsky Aircraft and is used as the primary handling qualities design analysis tool. This computer program is a fully coupled, nonlinear model of the helicopter, containing detailed descriptions of the components of the

helicopter that affect the handling qualities of the aircraft. GEN HEL is used for solving aircraft trims for a variety of flight conditions such as level flight, autorotation, and climb, and for solving the dynamic response of the helicopter to control inputs or aircraft disturbances.

The GEN HEL program is arranged in modular form, grouping descriptions of the basic helicopter elements such as the main rotor, tail rotor, and fuselage separately to allow uncomplicated modification of particular components, if necessary. One of these particular files, called the specific file, describes all components of the helicopter that are unique to a particular model. When the specific file describing the UH-60A descriptive input and control system is assembled with the rest of the general files of GEN HEL, the resulting program then models the UH-60A helicopter. This is the basic UH-60A simulation model around which this tail rotor blade severance handling qualities study was developed.

SEVERANCE TRANSITION ROUTINE

To simulate the buildup of centrifugal forces in the tail rotor during the three-bladed transition period and the transfer of this load imbalance to the rest of the helicopter, a special routine was incorporated into GEN HEL that calculated the inertial loads in the three remaining blades as the tail rotor proceeded to rotate in the discrete digital program solution. The inertial loads were summed for all three blades, and the resulting forces and moments imparted on the helicopter were solved. Additionally, the calculation of the aerodynamic thrust produced at the tail rotor existing in the basic version of GEN HEL was modified to reflect the proper decrease in thrust that occurred as the number of blades was reduced.

An associated logic routine was written into GEN HEL to distinguish when the program user elected to lose the first tail rotor blade due to ballistic damage during the dynamic simulation of the helicopter. This logic then routed the program through the additional imbalance equations and kept track of the location of the remaining unbalanced blade in order to determine when it entered the jettison envelope of tail rotor azimuth. Once the imbalanced blade reached this envelope, the program logic then left the tail rotor imbalance equations and proceeded with simulation of the helicopter using only the basic GEN HEL program again, but in the two tail rotor blade mode. Figure 27 is a flow diagram that depicts the logic procedure employed.

Because the tail rotor rotates at such a high speed, the duty cycle (simulated time between program updates) had to be made small enough to sample the tail rotor imbalance whenever the severance transition period was being studied. For simulation of the severance transition, a duty cycle was selected as a function of rotor speed to yield a change in tail rotor blade azimuth of 20 deg each duty cycle. This duty cycle was approximately seven times greater than the 1/50 second (50 cycles per second) normally used for dynamic simulation.

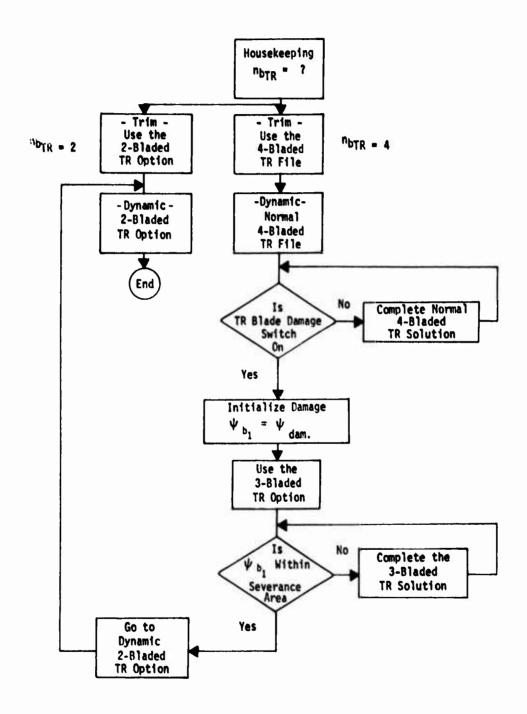


Figure 27. Flow Diagram of the Logic Modification used with GEN HEL (UH-60A) Tail Rotor File to Simulate Blade Severance Sequence

CASES STUDIED

Aircraft level flight trims and severance period transitions were studied with variations in aircraft gross weight, fuselage station cg location, density altitude, and rotor speed. These conditions were studied over a speed range from hover to 150 knots. Level flight was considered the most relevant flight condition since in all likelihood after losing tail rotor blades, the primary concern would be to return to base, not to continue the mission. Therefore, unusual or stringent flight conditions were not considered following the loss of two tail rotor blades.

Autorotation is not a demanding flight condition if two tail rotor blades are lost. Autorotation demands from directional control are not critical even when two tail rotor blades are lost because the directional control requirements are not near control system limits.

TWO-AND FOUR-BLADE TRIMS

Four trim cases were studied in accordance with the parametric mix shown below with trim data acquired at 0, 40, 60, 80, 100, 120, 140, and 150 knot speeds.

<u>Case</u>	^h Density	<u>G</u>	W, LB		FSCG	Rotor NR Speed
1	SLS	Low	(16450)	Aft	(360.2)	100%
2	SLS	High	(19900)	Aft	(360.2)	100%
3	10,000	High	(19900)	Aft	(360.2)	100%
4	10,000	High	(19900)	Fwd	(347)	100%

Baseline trims were gathered using GEN HEL for the UH-60A with four tail rotor blades. Trims were then attempted for the same conditions, only with two tail rotor blades.

The result of this exercise showed that the helicopter could be trimmed with two tail rotor blades throughout the speed range for low altitudes even at high gross weight. At higher altitudes, power requirements increase and there are conditions near hover and high speed that cannot be trimmed.

For all of the cases studied at nominal (100%) rotor speed, there always was some speed range at which the helicopter could be trimmed with only two tail rotor blades. This speed range, 60 to 100 knots, corresponds to the lower power requirements of the helicopter (see Figure 28). At rotor speeds of 95% NR and 110% NR (see Figure 29), the helicopter could be

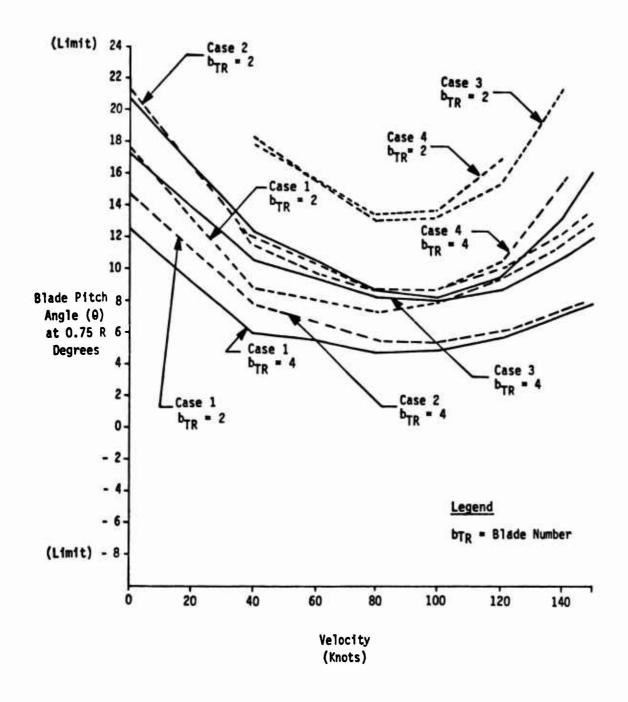


Figure 28. Two-and Four-Bladed Trims (100% NR)

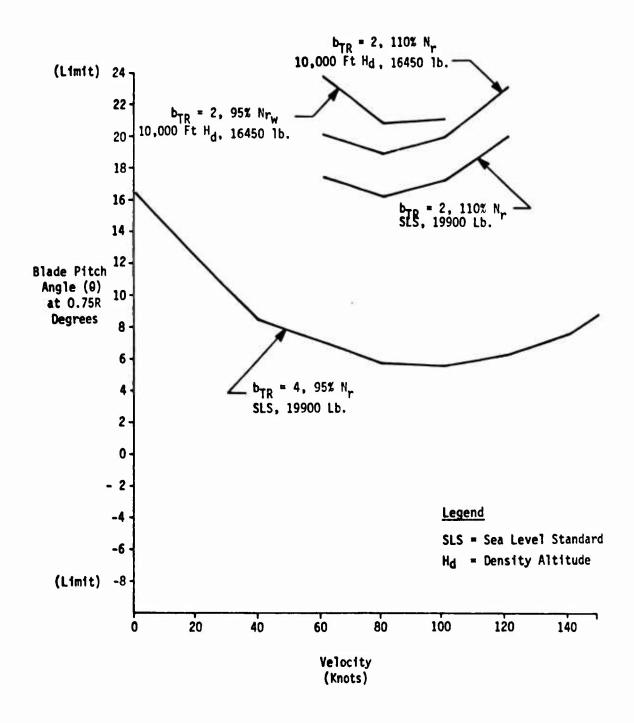


Figure 29 Two- and Four-Bladed Trims (95% NR and 110% NR)

trimmed at this same speed range with only two tail rotor blades. However, two blade trims become difficult (if not impossible) at any speed for low rpm at high altitudes, necessitating altitude reduction to achieve trim. Low rpm trims with two tail rotor blades are not advised because it becomes difficult to generate the required tail rotor thrust at reduced rotor speed.

These two tail rotor blade trim results indicate that for many flight conditions, the helicopter can be trimmed with only two tail rotor blades at any possible speed. At most flight conditions there exists a speed range (corresponding to the minimum power requirement) where the helicopter can be trimmed with only two tail rotor blades.

Naturally, trims with only two tail rotor blades provide less control margin and control sensitivity than the four-bladed situation, but under emergency conditions this situation could be considered acceptable.

For record, trim data printouts for the two-bladed and four-bladed conditions are provided in Appendix B.

AIRCRAFT "HANDS OFF" RESPONSE FOLLOWING BLADE LOSSES

The responses of the aircraft with no pilot correction for both the SAS OFF and SAS ON conditions are relatively mild, and should be acceptable to the pilot. These time history responses of the hands-off conditions, shown in Figures C-1 to C-12 (see Appendix C), demonstrate the response of the aircraft up to six seconds following blade loss. Actually, the pilot would react to the TR blade loss condition much earlier than six seconds and the response can be expected to be even less.

The aircraft responses vary as a function of the azimuth position where the initial blade is lost (Ψ damage). This is caused by the unbalanced blade (No. 2) traveling through different percentages of tail rotor azimuth before reaching the jettison envelope for different azimuth locations of the ballistic damage to the initial blade (No. 0). The centrifugal imbalance imparts different net impulses on the helicopter for the different proportions of the revolution for which the imbalanced blade is carried before reaching the jettison envelope. Therefore, different aircraft responses follow the event as a function of damage but all of the responses are mild and acceptable for the full sweep of values of Ψ damage. Table 4 presents data extracted from Figures C-1 through C-12 for: (a) the maximum roll ($\Psi_{\rm b}$), pitch ($\varphi_{\rm b}$), and yaw ($\Psi_{\rm b}$) angles achieved within 6 seconds after initiation of blade loss with SAS OFF; and (b) the roll, pitch, and yaw attitudes of the aircraft at 6 seconds following blade loss with SAS ON.

Table 4. Aircraft "Hands Off" Response Following Blade Losses

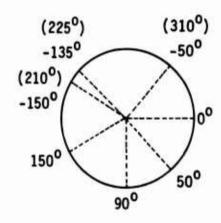
(a) A/C response within 6 seconds after severance (SAS OFF)

	Maximum Value Incurred Any Time During the Time History			
Ψ Damage	Φ _b Max	ϕ_{b} Max	$\Psi_{b \underline{Max}}$	
0	10	40	28	
50	14	38	28	
90	22	33	28	
150	31	28	28	
-135	30	28	29	
- 50	13	39	28	

(b) A/C response at 6 seconds after severance (SAS ON)

	Value at 6 Sec Point in the Time History			
ψ Damage	φ _b 6 Sec	φ ₆ 6 Sec	Ψ _b 6 Sec	
0	26	13	32	
50	27	13	33	
90	28	12	34	
150	28	11	34	
-135	28	11	34	
- 50	27	14	33	

SAS OFF - A/C response is low frequency, oscillatory deviation from trim.



SAS ON - A/C response is slow steady deviation from trim.

Trim condition = 16,450 Lb (Level Fit)
FSCG = 360.2
S.L.S.

STEPPED TRANSITION TIME HISTORIES

During the three-bladed transitional period, the unbalanced centrifugal force imparts loads on the tail rotor of the helicopter. A series of ten time history cases with parameters varied in accordance with the table below were performed to examine the helicopter response to the unbalanced load during the transition from four to two blades. The simulation runs, Figures D-1 through D-10, are included in Appendix D to this report.

Table 5. Time History Case Parameters

Figure	GW(Lb)	FSCG(In) V,Knots	Ω , z	Ψ dmg	SAS	Hd Alt.,Ft
D-1	16,450	360.2	100 Kts	100	- 150	Off	S.L.S.
D-2	19,900	360.2	100 Kts	100	- 150	0ff	S.L.S.
D-3	19,900	360.2	100 Kts	100	- 150	0ff	10,000
D-4	19,900	347	100 Kts	100	- 150	Off	10,000
D-5	19,900	347	150 Kts	100	- 150	Off	10,000
D-6	19,900	360.2	150 Kts	100	- 150	(On)	10,000
D-7	19,900	360.2	Hover	100	- 150	Off	10,000
*D-8	19,900	347	Hover	100	- 150	Off	10,000
D-9	19,900	347	Hover	95	- 150	Off	10,000
D-10	19,900	347	Hover	100	- 150	0ff	10,000

^{*}No print out

The ten aircraft configurations studied show that there are insignificant changes in the tail rotor load and aircraft response parameters with variations in gross weight, center-of-gravity location, airspeed, density altitude, or on/off condition of SAS. Two additional runs were conducted (Figures D-11 and D-12, Appendix D) to ascertain the significance of varying the rotational point at which blade loss occurs. These two cases illustrate that the loads developed at the tail rotor during the centrifugal imbalance period essentially are not dependent on the tail rotor blade azimuth location at which the first blade is lost. The centrifugal load in the unbalanced rotor blade does vary, as can be expected, with change in tail rotor speed. From the data, it can be determined that the centrifugal loads for 95% NR, 100% NR, and 110% NR are 24,000, 27,500, and 34,000 pounds, respectively.

It is anticipated that the loss of tail rotor blades is felt only as an impact load by the pilot since the transitional three-blade condition is of such short duration.

SIMULATOR INTERFACE

In order to qualitatively study pilot recovery from the tail rotor blade loss condition and retrimming of the helicopter, the basic GEN HEL program was coupled to a fixed base cockpit simulator (see Figure 30). Once the program is in a dynamic mode the program operator can select jettison of the two tail rotor blades. The program shifts from the four-bladed tail rotor solution to the two-bladed solution, imparting an impact-type response to the helicopter. Interpreting the blade loss condition from instrument readings and the visual display, the simulator pilot then proceeds to effect recovery of the helicopter from its response brought about by the forces from the blade loss and then trims the ship to the same or other flight conditions if necessary. The simulator pilot can proceed to a hover, if possible, and land, or conduct a run-on landing at low forward speed, if necessary.

The three-bladed severance transition portion of the solution is not used in the simulator study because real-time simulation requirements sample the tail rotor solution too infrequently to see a small enough change in tail rotor blade azimuth position. What transpires at the tail rotor during this very short imbalance period has negligible impact on helicopter handling qualities in a simulator study.

Nothing unusual or unexpected occurred during the piloted simulator portion of the handling qualities study. Even without external cues such as noise or impact motion, the simulator pilot was able to detect early the blade loss condition at the tail rotor. Yaw rates were not excessive for any of the severance cases studied using the simulator; qualitatively, these rates appeared to be between 10 deg/sec and 20 deg/ sec. Typical reaction was to put in the left pedal to slow or stop the yaw rate. If the loading condition and forward speed of the original trim did not permit the helicopter to be retrimmed, the simulator pilot would reduce speed until the slip rate could be reduced. This speed range corresponded to the low power requirement range: between 60 and 100 knots. Many configurations were flown back to a trimmed hover with the two remaining tail rotor blades, from which a landing could be negotiated. The higher power configurations (high weight, high altitude) could only be flown down to 30 or 40 knots before running out of tail rotor range. These cases represented loading configurations of the aircraft that would have to be landed at forward speed.



Interior View



Frontal View

Figure 30. Cockpit Simulator

RESULTS

DYNAMIC STABILITY ANALYSIS

Based on a Floquet stability analysis, a two-bladed, cross-beam tail rotor system is stable up to the maximum speed investigated (150 knots).

The least stable blade/pylon modes are the blade lead-lag mode and the third fixed system mode, both showing a damping level of approximately one-half percent. The effect of forward speed is not significant except near 150 knots where two of the fixed system modes indicate a degradation in damping level.

The blade lead-lag damping is lowered slightly by the flexibility of the hub while the flapping mode shows very little change in stability.

Modal frequency is not influenced significantly by forward speed. The blade flapping mode and one of the fixed system modes have frequencies close to each other, particularly at 150 knots, resulting in modal damping degradation. However, modal frequencies do not coalesce and vibration problems from modes interacting are not likely to occur.

Vibration and load level on the two-bladed rotor increase with forward speed, and at the highest speed investigated (150 knots) the maximum 2/rev vibration levels result in a roll moment at the gearbox-pylon attachment of 20 percent of the ultimate value. This level is acceptable to maintain flight for the 30-minute minimum following ballistic damage that has been established for the UH-60A helicopter, and to conduct a landing.

Rotor hub loads generally increase with forward speed. All shear and moment loads except the steady yaw moment load increase rapidly at forward speeds greater than 120 knots as the rotor operates increasingly in the blade stall environment. The highest roll moment loading of the gear-box-pylon attachment occurs at the 150 knot maximum forward speed investigated.

Significant increases in hub and gearbox accelerations are observed for forward speeds greater than 120 knots and the 2/rev accelerations are much higher than the 4/rev accelerations.

STRUCTURAL ACCOMMODATION

The centrifugal load resulting from loss of a full tail rotor blade applied for 0.05 second (360° rotation at 100% NR) will result in a torsion flexure of the tailcone of approximately 0.05 inch per inch, a level not considered excessive for a single load application.

PITCH LINK ACCOMMODATION

Severing of the tail rotor blade spar of the UH-60A must be accompanied by severing of the pitch horn fitting as well to achieve clean blade separation under all blade damage conditions.

PROTOTYPE SYSTEM PERFORMANCE

The prototype system successfully met the performance requirements in all tests performed, and compatibility with the blade jettison window for the UH-60A helicopter has been verified.

FLIGHT SIMULATION ANALYSIS

Helicopter trims gathered using the GEN HEL for the UH-60A with only two tail rotor blades indicated that the helicopter can be trimmed throughout the speed range for low altitudes even at high gross weights.

At higher altitudes, power requirements increase and there are conditions near hover and high speed that cannot be trimmed. At 100% NR there is always a speed range (between 60 and 100 knots) where helicopter trim can be effected with the two-bladed tail rotor.

In general, the aft cg location is more critical for trimming the helicopter with two tail rotor blades due to the shorter moment arm provided about the cq.

The trim cases studied at 95% NR and 110% NR indicate that the helicopter can be trimmed in the 60-to-100-knot speed range except for low rotor speed with high altitude conditions where altitude reduction may be necessary to achieve trim due to inadequate rotor thrust at the reduced rotor speed and density altitude.

The time history studies conducted indicate that responses of the aircraft for both the SAS OFF and the SAS ON conditions are relatively mild, and should be acceptable to the pilot.

Although the aircraft responses vary with respect to the azimuthal position at which the blade is lost (onset of the centrifugal load), all of the responses are mild and acceptable for the sweep of the values of damage examined.

FLIGHT SIMULATOR WITH PILOT-IN-THE-LOOP

Yaw rates were not excessive for any of the severance cases studied using the flight simulator, ranging between 10 and 20 degrees per second.

Following left pedal input to slow or stop the yaw rate where loading and forward speed conditions did not permit retrimming with the two-bladed tail rotor, yaw rate could be stopped and out-of-trim slip angles could be reduced by reducing speed to the low power requirement range of 60 to 100 knots.

The higher power configurations (high weight, high altitude) indicated a 30-to-40-knot minimum speed limitation, requiring run-on type landings.

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CONCLUSIONS

The prototype blade jettison system developed under this program has been verified by test to meet the rotational speed requirements of the UH-60A tail rotor and, when used in conjunction with blade severance assemblies, it will initiate tail rotor blade jettison within the blade jettison window established for that helicopter.

The dynamic stability analysis performed indicates that frequencies and amplitudes of the vibrations resulting from jettison of two opposing tail rotor blades will allow continued flight for a minimum of 30 minutes.

The simulation analysis performed to assess handling qualities indicates that the helicopter is controllable following the jettison of two opposing tail rotor blades and that the pilot work load to effect recovery is anticipated to be minimal.

An examination of the capability of the UH-60A helicopter to structurally accommodate the centrifugal force generated by the loss of an entire tail rotor blade for a period of 360° of rotor rotation indicates that redesign of the tail rotor drive shaft and the gearbox housing may be required.

For the UH-60A helicopter, the location of the pitch horn with respect to the point at which the blade spar is severed, necessitates that the pitch horn be severed simultaneously with the spar to prevent delayed blade jettison.

RECOMMENDATIONS

Based on the conclusions drawn from the results of the work performed under this contract, it is recommended that a follow-on ground test program be conducted that will demonstrate controlled jettison of tail rotor blades using a fully instrumented UH-60A tail cone and pylon and the prototype blade jettison system. The intent of the program would be to:

- . Verify proper blade severance/jettison using full length tail rotor blades.
- . Measure loads at the critical points to determine actual load levels achieved and to verify structural adequacy.
- Develop pyrotechnic devices, sized to sever UH-60A tail rotor blade spars and pitch horns.
- . Prepare a failure modes and effects analysis for the blade jettison system.
- Instrument the tail rotor assembly and conduct lightning tests of the blades. Measure induced voltage levels, if present, at the logic units.

REFERENCES

- 1. USA Ballistic Research Laboratories Contract Report #197, December, 1974, "Rotor Balance Restoration Study".
- 2. Arcidiacono, P. J., Prediction of Rotor Instability at High Forward Speeds, Volume I, Steady Flight Differential Equations of Motion for a Flexible Helicopter Blade with Chordwise Mass Unbalance, Sikorsky Aircraft, USAAVLABS Technical Report 68-18A, U. S. Army Aviation Materiel Laboratories, Fort Eustis, Virginia, February, 1969, AD-685860.
- 3. Sikorsky Aircraft Report, SER-70545, Revision 2, May 1978, "UTTAS Aeroelastic Stability Analysis".

APPENDIX A

PROTOTYPE PERFORMANCE DATA

This appendix contains the photographically recorded data obtained from the tests conducted on the prototype blade jettison system to determine the capability of the system to meet the performance requirements of the UH-60A helicopter. The data was secured by means of Polaroid photographic equipment attached to the oscilloscope. Each of the photographs includes the oscilloscope sweep rate used to obtain the data. Each of the data is in accordance with the following key:

Trace #1: Proximity Sensor A - open condition

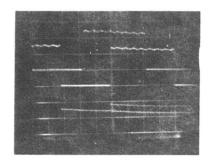
Trace #2: Proximity Sensor B - open condition

Trace #3: Proximity Sensor A - closed condition

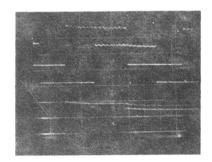
Trace #4: Proximity Sensor B - closed condition

Trace #5: EED Simulator - Blade #1 (left end is point of initiation)

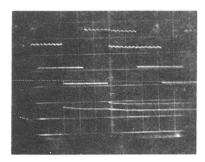
Trace #6: EED Simulator - Blade #3 (left end is point of initiation)



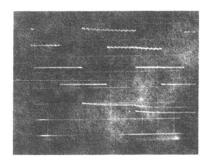
70% NR (834 RPM) 10 ms Sweep



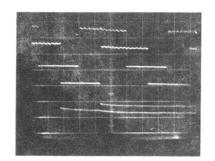
70% NR (834 RPM) 10 ms Sweep



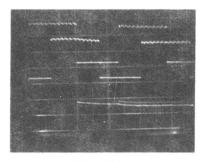
80% N_R (954 RPM) 10 ms Sweep



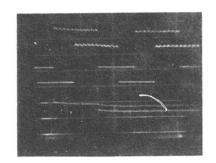
80% N_R (954 RPM) 10 ms Sweep



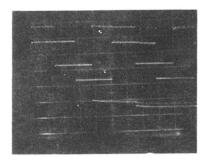
90% NR (1073 RPM) 10 ms Sweep

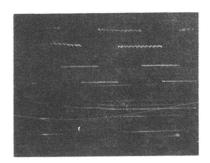


90% N $_{\mbox{\scriptsize R}}$ (1073 RPM) 10 ms Sweep

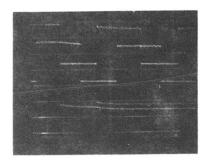


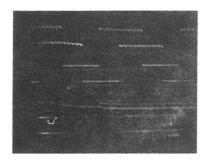
 $100\%~N_R~(1192~RPM)~10~ms~Sweep$ $100\%~N_R~(1192~RPM)~10~ms~Sweep$

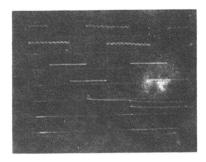




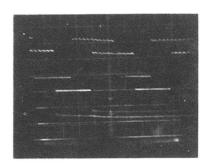
100% N_R (1192 RPM) 10 ms Sweep 100% N_R (1192 RPM) 10 ms Sweep



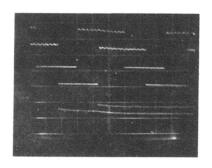




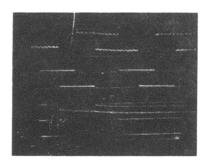
100% N_R (1192 RPM) 10 ms Sweep 100% N_R (1192 RPM) 10 ms Sweep



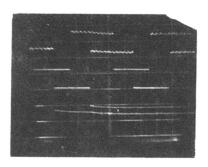
100% N_R (1192 RPM) 10 ms Sweep



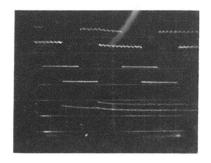
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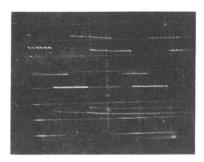
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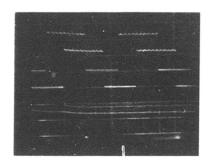
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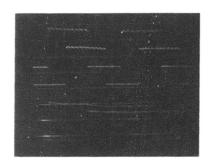
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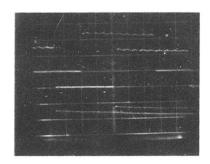
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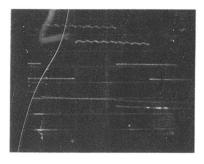
110% N_R (1311 RPM) 10 ms Sweep



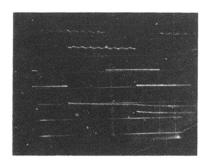
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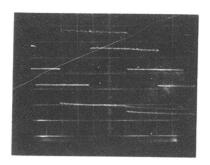


120% N_R (1430 RPM) 5 ms Sweep

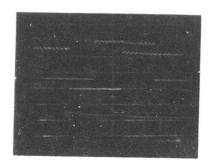


120% N_R (1430 RPM) 5 ms Sweep

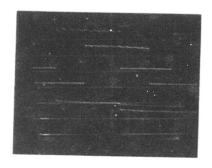




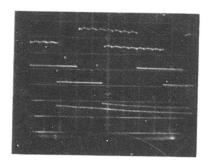
130% N_R (1550 RPM) 5 ms Sweep 130% N_R (1550 RPM) 5 ms Sweep



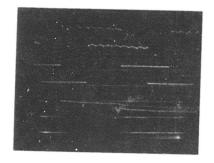
140% N_R (1669 RPM) 5 ms Sweep



140% N_R (1669 RPM) 5 ms Sweep



 $150\%~N_{\mbox{\scriptsize R}}$ (1788 RPM) 5 ms Sweep



150% N_R (1788 RPM) 5 ms Sweep

APPENDIX B

TWO-AND FOUR-BLADED TRIMS

The computer runs provided in this appendix are the two-and four-bladed trim data conducted to assess the ability of the UH-60A helicopter to achieve trim under varying density altitude, gross weight, and center-of-gravity conditions. Definitions of the symbols used are included.

SYMBOLS

WEIGHT	AIRCRAFT GROSS WEIGHT
IX	INERTIA ABOUT BODY X-AXIS LESS ROTOR, FT-LB-SEC ²
IY	INERTIA ABOUT BODY Y-AXIS LESS ROTOR, FT-LB-SEC ²
IZ	INERTIA ABOUT BODY Z-AXIS LESS ROTOR, FT-LB-SEC ²
OMEGMR	MAIN ROTOR ROTATIONAL SPEED, RAD/SEC
OMEGTR	TAIL ROTOR ROTATIONAL SPEED, RAD/SEC
KFR	ROTOR FILTER CONSTANT
FSHT	FUSELAGE STATION HORIZONTAL TAIL, IN.
LATSTK	LATERAL STICK POSITION, INTERMEDIATE CALCULATION, DEG
LNGSTK	LONGITUDINAL STICK POSITION, INTERMEDIATE CALCULATION, DEG
COLSTK	COLLECTIVE STICK POSITION, INTERMEDIATE CALCULATION, DEG
PEDAL	DIRECTIONAL CONTROL PEDAL POSITION, INTERMEDIATE CALCULATION, DEG
XAIN	LATERAL STICK POSITION, IN.
XBACTP	LONGITUDINAL STICK POSITION WITH BIAS ACTUATOR, %
VXB	X-AXIS
VYB	BODY AXIS TRANSLATIONAL Y-AXIS FT/SEC
VZB	VELOCITIES Z-AXIS
P	ROLL RATE, RAD/SEC
Q	PITCH RATE, RAD/SEC
R	YAW RATE, RAD/SEC
ALFWF	FUSELAGE ANGLE OF ATTACK, DEG
CHITPP	ROTOR DOWN WASH ANGLE, DEG

EKTR	DOWNWASH FACTOR OF TAIL ROTOR ON VERTICAL TAIL
QWF	DYNAMIC PRESSURE AT FUSELAGE, LB/FT ²
MUXS MUYS MUZS	NORMALIZED SHAFT AXIS TRANSLATIONAL VELOCITIES AT THE HUB, NORMALIZED BY AMR TIP SPEED
LAMBMR	MAIN ROTOR INFLOW RATIO
DWSHMR	MAIN ROTOR INDUCED DOWN WASH
XMR YMR ZMR	ROTOR BODY AXIS FORCES, LB
LMR	ROLL MOMENT, FT-LB
MMR	THE C.G. PITCHING, MOMENT FT-LB
NMR	YAWING MOMENT, FT-LB
XWF YWF ZWF	FUSELAGE BODY AXIS FORCES, LB
LWF	ROLL, FT-LB
MWF	FUSELAGE BODY AXIS MOMENTS PITCH, FT-LB
NWF	ABOUT THE C.G. YAW, FT-LB
XHT	
YHT	HORIZONTAL TAIL BODY AXIS
ZHT	FORCES, LB
FSCG	FUSELAGE STATION C.G., IN.
WLCG	WATER LINE STATION C.G., IN.
RHO	AIR DENSITY, SLUGS/FT ³
TIME	TIME INTERVAL BETWEEN ROTOR CALCULATIONS, SEC

NBSS NUMBER OF BLADES

NSSS NUMBER OF BLADE SEGMENTS

PASCNT NUMBER OF ROTOR CALCULATIONS FOR TRIM

SHT HORIZONTAL TAIL AREA, FT²

A1S LATERAL CYCLIC PITCH, DEG

BIS LONGITUDINAL CYCLIC PITCH, DEG

THETAD MAIN ROTOR COLLECTIVE PITCH AT CUFF, DEG

THETTR RAIL ROTOR IMPRESSED COLLECTIVE PITCH @ CENTER OF

ROTATION, DEG

XBIN LONGITUDINAL STICK POSITION, WITHOUT BIAS ACTUATOR, IN-

XBACTI LONGITUDINAL STICK POSITION, WITH BIAS ACTUATOR, IN-

THETAB AIRCRAFT PITCH ATTITUDE, DEG

PHIB AIRCRAFT ROLL ATTITUDE, DEG

BETAWF AIRCRAFT SIDESLIP ANGLE, DEG

GAMC CLIMB ANGLE, DEG

OMGRAT RATIO OF ACTUAL TO TRIMMED ROTOR SPEED

PSIDOT EULER ANGLE YAW RATE, DEG/SEC

EKTX MAIN ROTOR DOWNWASH FACTOR AT HORIZONTAL TAIL IN BODY

X-AXIS

EKTZ MAIN ROTOR DOWNWASH FACTOR AT HORIZONTAL TAIL IN BODY

Z-AXIS

EPSWT FUSELAGE DOWNWASH ANGLE AT HORIZONTAL TAIL, DEG

KOHT SQUARE ROOT OF DYNAMIC PRESSURE RATIO AT HORIZONTAL TAIL

CTSIG MAIN ROTOR THRUST COEFFICIENT SOLIDITY RATIO

CHSIG MAIN ROTOR H-FORCE COEFFICIENT SOLIDITY RATIO

CQHSIG MAIN ROTOR TORQUE COEFFICIENT SOLIDITY RATIO

NZ LOAD FACTOR ALONG THE AIRCRAFT Z-AXIS

VC	RATE OF CLIMB, FT/MIN
HBAR	MAIN ROTOR H-FORCE, LB
JBAR	MAIN ROTOR SIDEFORCE, LB
TBAR	MAIN ROTOR THRUST, LB
LBARH	MAIN ROTOR ROLL MOMENT, FT-LB
MBARH	MAIN ROTOR PITCH MOMENT, FT-LB
QBAR	MAIN ROTOR TORQUE MOMENT, FT-LB
XT	EMPENNAGE X-FORCE (VERTICAL + HORIZONTAL TAIL) IN BODY AXIS, LB
YT	EMPENNAGE Y-FORCE (VERTICAL + HORIZONTAL TAIL) IN BODY AXIS, LB
ZT	EMPENNAGE Z-FORCE (VERTICAL + HORIZONTAL TAIL) IN BODY AXIS, LB
LT	EMPENNAGE ROLL MOMENT (VERTICAL + HORIZONTAL TAIL) IN BODY AXIS, FT-LB
MT	EMPENNAGE PITCH MOMENT (VERTICAL + HORIZONTAL TAIL) IN BODY AXIS, FT-LB
NT	EMPENNAGE YAW MOMENT (VERTICAL + HORIZONTAL TAIL) IN BODY AXIS, FT-LB
XVT	VERTICAL TAIL X-FORCE IN BODY AXIS, LB
YVT	VERTICAL TAIL Y-FORCE IN BODY AXIS, LB
ZVT	VERTICAL TAIL Z-FORCE IN BODY AXIS, LB
V	AIRSPEED, KTS
DELS	SWASH PLATE ROTATION, DEG
VSOUND	SPEED OF SOUND, FT/SEC
DEL3MR	MAIN ROTOR DELTA 3 ANGLE, DEG
TWSTMR	MAIN ROTOR TWIST, DEG
TWSTTR	TAIL ROTOR TWIST, DEG
WLHT	WATER LINE HORIZONTAL TAIL, IN.

SVT VERTICAL TAIL AREA, FT²

IHT HORIZONTAL TAIL INCIDENCE, DEG

IS SHAFT ANGLE INCIDENCE, POSITIVE FWD, DEG

TH75MR MAIN ROTOR COLLECTIVE PITCH AT .75 RADIUS. DEG

TH75TR TAIL ROTOR COLLECTIVE PITCH AT .75 RADIUS, DEG

XCIN COLLECTIVE STICK POSITION, IN.

RSTR YAW ACCELERATION AT TAIL ROTOR, SHAFT AXIS, RAD/SEC²

AAØF FOURIER SERIES COEFFICIENT FOR MAIN ROTOR

AA1F FLAPPING, NEGATIVE SERIES, DEG

BB1F

3B1L

AAØL

AA1L LAGGING, NEGATIVE SERIES, DEG

EKWFX MAIN ROTOR DOWNWASH FACTOR AT FUSELAGE, BODY X-AXIS

FOURIER SERIES COEFFICIENT FOR MAIN ROTOR

EKWFZ MAIN ROTOR DOWNWASH FACTOR AT FUSELAGE, BODY Z-AXIS

SIGWT FUSELAGE SIDEWASH ANGLE AT VERTICAL TAIL, DEG

KQVT SQUARE ROOT OF DYNAMIC PRESSURE RATIO AT VERTICAL TAIL, DEG

TTR TAIL ROTOR THRUST, LB

HPMR HORSE POWER MAIN ROTOR

KTRBLK TAIL ROTOR BLOCKAGE FACTOR

VXBDOT BODY AXIS ACCELERATION AT C.G. IN X-AXIS, FT/SEC²

VYBDOT BODY AXIS ACCELERATION AT C.G. IN Y-AXIS, FT/SEC²

VZBDOT BODY AXIS ACCELERATION AT C.G. IN Z-AXIS, FT/SEC²

· PDOT AIRCRAFT ROLL ACCELERATION, RAD/SEC²

QUOT AIRCRAFT PITCH ACCELERATION, RAD/SEC²

RDOT AIRCRAFT YAW ACCELERATION, RAD/SEC²

XTR TAIL ROTOR FORCE IN X-AXIS, LB

YTR TAIL ROTOR FORCE IN Y-AXIS, LB

ZTR TAIL ROTOR FORCE IN Z-AXIS, LB

LTR TAIL ROTOR ROLL MOMENT ABOUT BODY X-AXIS, FT-LB

MTR TAIL ROTOR PITCH MOMENT ABOUT BODY Y-AXIS, FT-LB

NTR TAIL ROTOR YAW MOMENT ABOUT BODY Z-AXIS, FT-LB

ALFHTT LOCAL ANGLE OF ATTACK OF HORIZONTAL TAIL

ALFVTT LOCAL ANGLE OF ATTACK OF VERTICAL TAIL

AABB1F $\sqrt{(AA1F)^2 + (BB1F)^2}$

PSITR2 AZIMUTH POSITION OF TR BLADE NUMBER 2, DEG

VXSTR. LINEAR ACCELERATION AT TAIL ROTOR IN TAIL ROTOR SHAFT

X-AXIS, FT/SEC²

VYSTR. LINEAR ACCELERATION AT TAIL ROTOR IN TAIL ROTOR SHAFT

Y-AXIS, FT/SEC2

VZSTR. LINEAR ACCELERATION AT TAIL ROTOR IN TAIL ROTOR SHAFT

Z-AXIS, FT/SEC²

PSTR. AIRCRAFT ROLL ANGLULAR ACCELERATIONS AT TAIL ROTOR IN

SHAFT AXIS

WLVT WATERLINE VERTICAL TAIL

FSVT FUSELAGE STATION VERTICAL TAIL

XA LATERAL STICK POSITION, %

XB LONGITUDINAL STICK POSITION, %

XC COLLECTIVE STICK POSITION, %

XP PEDAL POSITION, %

XPIN PEDAL POSITION IN INCHES

PSTR AIRCRAFT ROLL ANGULAR ACCELERATIONS AT TAIL ROTOR IN TR

SHAFT AXIS, RAD/SEC²

AIRCRAFT PITCH ANGLULAR ACCELERATIONS AT TAIL ROTOR IN TR OSTR SHAFT AXIS, RAD/SEC2 AIRCRAFT YAW ANGULAR ACCELERATIONS AT TAIL ROTOR IN TR **RSTR** SHAFT AXIS, RAD/SEC2 TAIL ROTOR INERTIA THRUST (SHAFT AXIS), LB TITR HITR TAIL ROTOR INERTIA H-FORCE (SHAFT AXIS), LB JITR TAIL ROTOR INERTIA J-FORCE (SHAFT AXIS), LB TAIL ROTOR INERTIA HUB PITCHING MOMENT (MOMENT ABOUT SHAFT MHITR Y AXIS), FT-LB LHITR TAIL ROTOR INERTIA HUB ROLLING MOMENT (MOMENT ABOUT SHAFT X AXIS), FT-LB QHITR TAIL ROTOR INERTIA HUB YAW MOMENT (MOMENT ABOUT SHAFT Z AXIS), FT-LB XITR TR INERTIA X-FORCE (BODY AXIS), LB YITR TR INERTIA Y-FORCE (BODY AXIS), LB TR INERTIA Z-FORCE (BODY AXIS), LB ZITR TR INERTIA ROLL MOMENT (BODY AXIS), FT-LB LITR TR INERTIA PITCH MOMENT (BODY AXIS). FT-LB MITR TR INERTIA YAW MOMENT (BODY AXIS), FT-LB NITR LONGITUDINAL ACCELERATION AT PILOT'S LOCATION. FT/SEC2 **AXP** LATERAL ACCELERATION AT PILOT'S LOCATION. FT/SEC² AYP VERTICAL ACCELERATION AT PILOT'S LOCATION, FT/SEC² **AZP** VXP LONG. VEL AT PILOT'S LOCATION (ALONG BODY X-AXIS), FT/SEC VYP LATERAL VELOCITIE AT PILOT'S LOCATION (ALONG BODY Y-AXIS), FT/SEC **VZP** VERTICAL VELOCITY AT PILOT'S LOCATION (ALONG BODY Z-AXIS), FT/SEC YAW ACCELERATION AT TAIL ROTOR, SHAFT AXIS. RAD/SEC² RSTR.

AZIMUTH SELECTED FOR DAMAGE OCCURANCE. DEG

PSIDMG

DTO	AUMBED OF TALL DOTOD DIADEC
BTR	NUMBER OF TAIL ROTOR BLADES
MADD	AUXILIARY PITCHING MOMENT (BODY AXIS SYSTEM), FT-LB
XADD	AUXILIARY X-FORCE (BODY AXIS SYSTEM), LB
YADD	AUXILIARY Y-FORCE (BODY AXIS SYSTEM), LB
ZADD	AUXILIARY Z-FORCE (BODY AXIS SYSTEM), LB
NADD	AUXILIARY YAWING MOMENT (BODY AXIS SYSTEM), FT-LB
LADD	AUXILIARY ROLLING MOMENT (BODY AXIS SYSTEM), FT-LB

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t		9 6			273.0		9.0	36.888562	26.911857	56,424233	24.446562	1.3200811	0.0		8.8	0.0	0.0				0.0		9.0	9.0	0.9	9.0	0.0	2,5297503	6.91320406	-32,061928	67,389674	8.6	5.2942291	9.5	0.0	•							
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HUN 22.	8 1 3 1		•	-16.0	-18.0	234.0	32,500000	31,582801	-3.6	9.7478772	16.535061	5.6424233	9.0	4,3234276	-2.27.55399	-0.53871962	. 5987154	W.28154563	8.84644779E-1	0.69240182	1:0350151	9.0	6.64652613	-51,925581	30.011919	1161.9499	1767.5339	1.0	8.10844655E-1	0.27648868E-1	-0.26583888E-2	0.155014596-2	0:10375686E-3	W.12684834E-2	2.0	1691,9569	-397.44861	6670.4382	-12314,034	363	-1.9617445	6.9	2.2986331
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10.535312 20.535312 20.535312 20.535312 20.535312 30.535312 30.535312 30.535312 30.535312 30.535323 30.53532	Ħ	788.48078	BIT	8.04	SVT	32.586888	CSTR		
15.25312 18.525237 1	TSTR	-1.4311505	A18	-2.2507430	IHI	-6:0057127	XX	42:185259	
20.500.00 10.500.00	GSTK	15,55512	913	15.668603	15	9.5.	e x	1128617	
14.526.90 14.526.90 14.526.90 14.526.90 14.526.90 14.526.90 14.526.90 15.1670.22 16.306.23 16.30	COLSTR	20.450347	THETAB	20,456547	THISMR	18:376547	X	62.353425	
7.167024 15.167024 16.40024 16.16702	UAL	14,528298	THETTR	26.58415S	THISTR	13.000155	×	41.32/628	
253.26489 THETAB 2.346831 RSTR. 8.8 COSTR. 15.1000c254 FIRETAB 2.346836 RAIT - 0.5512567 RSTR. 15.1000c254 FILETAB 2.346836 RAIT - 0.5512367 RSTR. 15.1000c254 FILETAB 2.346836 RAIT - 0.5512367 RSTR. 15.1000c254 FILETAB 2.346836 RAIT - 0.5512367 RSTR. 15.1000c254 FILETAB 2.346836 RAIT - 0.55100c254 FILETAB 2.346836 RAIT - 0.55100c254 FILETAB 2.346836 RAIT - 0.55100c25 FILETAB 2.346836 FILETAB 2.	N.	4.2389239	MIGH	B:94158571	XCIN	6.2353425	XPIN	2.2316336	
15.10f0c2	ACTP	7.1876531	XBACTI	0.71676531	RSTH.	9.0	PSTR	9.0	
15.109622 16.109622 16.109622 16.109622 16.109622 16.109623 16.109623 16.109623 16.109623 16.109623 16.109623 16.109623 16.109623 16.109623 16.109623 16.109623 16.10963 16.10	\$0	253.28489	THETAB	2.3486844	AABF	3.2010174	DSTR		
16.506254 0.6 0.6 0.7 0.10	D	15.100662	PHIB	9.0	AA1F	-9,5523587	E STE	9	
0.6 491001 0.0 491001 0.0 491001 0.0 491001 0.0 491001 0.0 491000 0.0 49	20	. 38825	BETANF	3.3362936	BUIL	-6.46696413	-TITR -	0.0	
Colorary		9.0	GAMC	9.0	AABL	-7.5114746	HITE	9	
### ### ##############################		0.0	DHGRAT	1:0	AASE	0.34991318	317R	9.0	
## No.94064477		9.	PSIDOL	9.0	BUIL	0.79826238	RLITE	0.9	
70.591001 EFFORT 1.7502029 EFFORT 2.50710001 U.5.400000 V.5.400000 V.5.400000 V.5.400000 V.5.400000 V.5.400000 V.5.4000000 V.5.4000000 V.5.4000000 V.5.4000000 V.5.4000000 V.5.4000000 V.5.4000000 V.5.40000000 V.5.400000000 V.5.400000000000000000000000000000000000	d H d	N. 90004477	- ERTX	1.5097978	EKMPX	0.66251869	LHITR	0:0	
0.6	1199	. 39186	EKT2	•	EKMF 2	1,0116061	DHITE	0.0	
74.9100000 KGNT 6.87177979 KGVT 6.9205230 YITR 6.32900000 KGNT 6.39900000 KGNT 6.39900000 KGNT 6.399000000 KGNT 6.399000000 KGNT 6.39900000000000000000000000000000000000	2	5	EPSHT		SIGHT	6.56716665	XITR -	0:0	
0:34902707 0:34902707 0:34902707 0:34902707 0:34902707 0:34902707 0:34902707 0:34902707 0:34902708 0:44177 0:34902708 0:44177	<u>.</u>	74.916666	* OH	0.87177979	KOVI	0.83265236	YITE	9.0	
0.2462979E-1 CASLG -6.7779617L-2 0101 24.517443 LITA -0.12578497E-1 NZ 0.999487E-6 777 152057985 MITR -0.12578497E-2 VC 0.14385114E-4 X788LX 1.00 -0.06925649E-2 VC 0.14385114E-4 X788LX 1.00 -15174.656 VC 0.14385114E-4 X788LX 1.00 -15174.756 VC 0.		. 34962787	-61916		£101	3.9556616	#1#Z	0.0	
	2	58849979E-	91843	-8.17779817£-2	1010	24,517443	LITH		
	52	. 39754888E	DISER		TTR	1520,7905	HITE	0:0	
8.86825695E-2 VC	TOTE	u.	7 N	0.99938765	HPR	2010.4040	NITE	9.0	
2951.1387 H8AR -1084.7321 VXBUOI 8.4648699E-2 APP -1974.14301 VYBUUT -8.6060403E-1 AZP -1974.4501 TBAR -4717.4501 VYBUUT -8.4960401778-1 VXP -1564.4017.730 GBAR -4717.4619 GUOT 8.47584139E-2 VXP -1976.6650 X1 -498562.819 GUOT 8.47584139E-2 VXP -1945.58401 YT -455.83788 YTR 8.8 -1945.58401 YT -455.83788 YTR 8.8 -1945.8441 YT -455.83788 YTR 8.8 -1945.8441 YT -4563.83788 YTR 8.8 -1945.8441 YT -468.962.648 YTR -16116.931 YADD -64.395641 XYT -6.5384276 XADD -64.395641 XYT -6.5384276 XADD -64.395641 XYT -6.5384276 XADD	art o) A		KTRBLK	1.0	AXP	1.3170944	
-1917.14501	œ :	2651.1387	HOAR	-1604.7521	VXBUOI	8.46448699E-2	AYP	-8.12584278	
-10174.006 10AR 10242.616 V2800T -0.7459274E-2 VXP 15264.012 1045964717E-1. VXP 15264.012 1045964.012 1056	* :	-397 . 14581	- WAN	-397:14301	-vyBuut-	-8:00682683E-1	AZA	-32r159010	
-15664.612 LGARH	¥ :	-10174.000	TOAR	16262.610	VZBUOT	-0.170152746-2	a×>	253,28489	
-13004.012	¥ (-9645.4561	LBARH	-4719.4015	P.001	-0.59640717E-1-	dAA	-15.188622	
######################################	*	4.01	HARE	-25862.019	1000	47584139E-	474	18,366259	
-1916-6058 XI -91.448581 XTR 8.8 PSIONG -1946-50481 YT -505:03768 YTR 1429:1819 BTR -1840-50481 YT -505:03768 YTR -528:1889 MADD 1184-6017 LT -794:75285 LTR -528:1889 MADD -18758-8992 NT 18288-902 NTR -44288-819 ZADD -1840-1286 NT 18288-902 NTR -44288-819 ZADD -1840-39591 XYT -27.8468-902 ALFHT -6.5344276 NADD	œ	40117.738	DONE	41005.206	R001	61705999E	RSTR	9.0	
1429:50461	<u> </u>	-1916.6658	T X	-91,448581	XTR	9.9	PSIONG	9.6	
1346.5537 21 1342.0877 217 -526.18869 AADD 8 1164.8617 LT -796.75285 LTR 8736.4444 XADD 8 -6736.8492 NT 16266.962 NTR -42286.819 XADD 8 -674.44521289 NT 16266.962 NTR -42286.819 XADD 8 -674.445891 NT 16266.962 NTR -42286.819 XADD 8 -674.445891 NT 16266.962 NTR -42286.819 XADD 8		-699:58481	- V T	565,63700	-V1R-	1429,1619	-BTR		
1184.0017 LT -190.75285 LTR 0136.4444 - XADD 0 -0136.6492 LT 36462.648 MTR -16116.931 YADD 8 -0184.1286 NT 16286.962 NTR -42886.619 ZADD 8 -04.95691 XYT -27.846889 ALFITT -6.6344276 XADD 8	<u>.</u>	-346.58397	11	1342,6077	ZIR	-520,16869	MADO	9.0	
-6164-1206 NT 16268-962 NTR -16116-931 YADD 8 -6164-1206-019 ZADD 8 -64296-019 ZADD	<u>.</u> .	1164.6617	<u>ב</u>	-196,75285	LtR	0730.4444	XADO		
-6164.1286 NT 16268.962 NTR -44288.819 ZADD 8 -64.395691 XVT -27.846889 ALFHTT -6.6344276 NADD 8	•	-6758.6992	F	38462.648	£	-16116.931	YADD	9.9	
-64-345641 XY -27-846869 ALFITT -6-6344276 NADO 6	<u>.</u>	10	⊢	14266,962	α Τ	-44260,519	ZADO		1
SECTION AND SECTION AND SECTION ASSESSED.	<u>-</u>	-64.395641	- ×	-27.846889	ALFHTT	6.634427	NADO	9.0	
1361191361	_	- 4- 447 9 8 B B - 5 -							

AUN 36.
-
11.
1-77 8-3EP.
(0)
UTTAS(S)

×	6200.0	100	246.29999	DELS	8.5.	VXSTR.		
IT	41367.0	OH &	8.17569#88E-2	A SOUND.	1077.6	VVSTR		
17	50224.8	1111		DF1 SHE	-	918/2	•	
OMEGHE	20025/ 45	2000		SEL SEL			0 6	1
DMEGTR	137.67999	50002	6	THSTIR	9	>	E-1-0	
	15.0	PASCHT	1389.8	MC Pr	256.8	145	6.000	
FSHT	755.48554	LHE	45.8	A V T	626624	0570		
LATSTE			-1 .0002174	-	200000000000000000000000000000000000000		10000	
	24.644.					E 6		
200	991929192	910	969646991	2	B 10 4	D '	30.013498	
COLSTR	14.548158	THETAB	19.596150	11111	4,3181582	UX	55.730498	
PEUAL	20.001828	THETTR	34.765617	71271F	17,263617	×	25,967391	,
MAIN	4.8878954	MIGH	3.6815496	XCIN	3.5738498	HEAR	1.4632626	
XBACTP	45.576664	XBACTI	4.5570009	RSTR.		PSTR		
AXB	B.10537579E-1	THETAB	4.6026584	AABF	5.5818833	DSTR		-
844	0	PHIC	-2.9721521	AA1F	-1.7846564	RSTR		
87A	B.145539876-2	BETANE		8816	-111839717	TITE		
•	9	GAMC		AABL	-6.5223581	HITK	9 6	
		DHGRAT		ARIE	-88:15726788691			
Œ	9	PSIDOT		8616	8-12697491	MHITE		
ALFHF	-13.236188	EATX	-6.26184423	EKAFK	6.79995393E=5	LHIPP.		
CHITPP		EKTZ	0.24242269	EKNP Z	B.11899566E-3	31110		
ERTR	80.6	EPSHT	6666644-8	SIGNT		XITR	6	
110	0.274242656-6	KOHT	0.67177979	T AOX	0.84852613	YLTR		
HUXS -	- 6.c1179503E-4-	-67819-	6-696968988-1	+101	-28.771652	21.4		
BLVB	5	CHSIG	-8.31416568E-2	1010	26.627283	118	6	
8704	B-696196196	COMBIG	- 6 - 6 51 6 5 5 1 B E - 6	110		1170		į
LAMBER	-8-6265626E-1	72	E-99448526	TOI	0.00 E 0.00 E	A 1 1 2	9 5	
DESTIN	8-628569216-1	\ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \	6-12232427E-3	K TRBL K	7959999	4.0	3.747483C	
KIK	1624-1981	HOAR	-655,19994	VXBDOT	B. 94352139E-2	d A V	1-6699127	
- 344	-356.56612	JUAN-	354.58612	-VABUOT	- A : 7 : 8 5 : 2 5 6 F = 2		32 Bug 22	
ZHR	-16510.732	TOAR	16571.011	V.Buut	-4.25768788E-3	Q ×	9-16847474F-1	
LHK	-6169-6165	LBARH	-3865,0195	PD01	B-11469512E-3			
211	14079.140	HOPET	-5982-1947	1000	8.583594BBE-4	VZP		
212	41556.573	2400	42379.974	RUOT	-6.20749646E-3	8878-		
XINF	-8.58838852E-5	F	6-6138894	X T R		PSTUNG		
YEF		V T	-8.54902633E-6	TR	1340,8761	878		
342	W. 72173330E-5	12	6.4819962	ZTR	-488-83922	MADD		
LHF		-	-8.77658361E-6	4	6198.9987	XAD9		
	-8-14386782E-3	Ē		E	-15121-881	VADO		
412		2	0.973783476-5	2		2 A D D		-
KHM	6.01.50096	L A X	-8.79245321E-7	ALFMTT	-11-827455	MADD	3	
7117		***	-4-446004446-6-	-A-F V T T-		204		

REIGHT-	- 1	FSC6	368.29888		8.83	- 5.51182	6
×	6/68.6	937#	246,29999	DELS	3	2 - 5	
_	41567.8	O F	6.175cacsac-2	ASOUND	1277.8	-	•
71	3024.9	TIME	8.24883688E-1	DELSMR	•	-	6
OMEGHA	29.128888	2000		TEG TER	6.18.8		6
OMEGTR	137.67999	2002	2.0	THSITK		1 × 1	273.9
KFR	15.0	PASCHT		THIS	256.19	4.5	
FOH	788.46548	BHT		1 A S	020000		•
LATSTR		A18	-2.6482275	H			10000
LAGSTR	_	8.18	1527865) #	. :	******
COL STR	A 1 1 3 3 2 7	TABTA		1	; '	D (33.176306
1000	2011		1/2011-01	ALC LE		N I	47.689235
1	10110	112114		11771	18.535429	A	48,355132
MAN	•	MIGH	5.3796388	KCIK	4,7689235	MIGE	2021641.2
BACTP	5	XBACT1	4.8545247	SOIK.	9.3	PSTE	20.
9	67,351296	THETAB	4.8653151	AABP	3.4938685	GSTR	6
9	9.0	Prin		AAIP	-2.6327256	N S	6
97A	5.7332120	BETANF		8815	- S - S - S - S - S - S - S - S - S - S	1115	•
	3	CAMC			454545		
!		CHEST			• 1		• 1
			3		19204020	X	S
				1100	*********	*	89
	-	EN IN		EKMPX	962	LHITE	60
77.17.1		2173	2.0101556	2447	1.8554897	0H114	£.
X .	•	EPSHT	•	SICET	9.0	MITH	60
4 20	.943562	TO Y	4.93257628	KGVI	0.64652813	YITR	9.
FUES	8.54718986E=1	CT316	1:92688288E=1	1017	-38.773682	H112	8.9
BC48	9.	CHSIG	-8.33631868E-2	1010	29.904997	118	- 10
870H	8.275916286-2	COMSIG	0.63835853E=6	118	1143.4731	178	
AMBER	-6.39638381E-1	72		IPIR	1641.7487	178	
SECOND S	B.42397543E-1	AC.	8-15147328	KIRBLE			2.1711184
¥	695.7923	3401	-696.77522	VXBDDT	8.227142146-1		9 05845934
-	588.1858	JBAR		-			200000
2118	19662.4	TBAR	19141.000	V / Buot	22 41449546-2		700.00
*	121	LBARM	1790	P001			
Œ	674	TO VE	-24.78 . 1.284	1000	2 - 2 - 2 - 2 - 2 - 2 - 2 - 2 - 2 - 2 -		
***		200	7525 757	1000			301355150
XIII	158	L R	200000000000000000000000000000000000000	9 6	5-3019+C949+9-		
			95501510	A 1 A	9.0	101	
	•	- 1	•	414	1674.3931	BTR	2:5
447	365.61868	1.7	156.73501	2 1R	-391.12867	MADD	8.0
	1		-12,425187	L 1 R	-	XADD	8.6
A B E	•	⊢ I	4537,3834	Œ	-12116.222	4400	8.8
-		-	135.88347	K-K	-53294,477	ZADD	8.0
M H		XVT	-1,4002123	ALFHTI	-4.8552675	MADO	
- X		- A & A	24 482 14 10	The state of the s	6.6		
-						777	

8-3EP-77
1-51-17
UTTAS(376)

RUN 38.

×	6268.8	901=	246.29999	DELS	-5.0	CXBIR.	6
IY	41567.8	SHO	7	ASOUND	2772	TELENA	
	2 4774		100000000000000000000000000000000000000		•		•
77	900000000000000000000000000000000000000	341	•	UEL 278	9	WISTA	
200	0	2000		EL PRI	*18.E	T C L	8
DMEGTE	137.01999	2000	S. 8	THOIL	-16.6	HLVI	273.0
KFR		PASCRT	1595,0	RCMT	234;0	PSVT	695.8
FBHT	700.43000	LES	45.8	3 × 1	32.582888	DSTR	6
LATSTR		A13	-1-7875584	IHT	50.00 S	X	A 1. TOBBER
LNGSTX		818	4.4729568		5	3	41.04597
COLSTR	17.589717	THETAB	17.589717	717718	8111965.1	U X	A 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1
PEDA		11677	22 97294.5	0 - 1 - 1	0 01304 80	3	01 - 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 - 1 -
444	- 612666	2 - 3 - 3	-3-100000	K	- 3466		46,637331
9114	10000		No beautiful a		•		89564413
1 1							
	0000010	TAK TAB	9001001.5		2.656664		
	000	274			-1.7656763	1	9
874	•	BETANK	0.7106346	4199	-E: ++288478k-1	4114	9.0
	•	S M M	3.	AAGE	-4,1698185	HITE	6.0
0	9.6	DEGRAT	1.0	1111	B:10997988	4117	8:8
Œ	9.	PSIDOT	7.0	9016	8.55283028E-1	MHITE	6.
ALFNF	-6.5654667	ERTX	1.2607338	ERMFX		LHITE	
CHITPP	4	EKT2	1.9474187	EKHF 2	1.0146738	TITE	5
EXTR	9.0	EPSnT	0.53718633	S.547	1.3411167	KITE	6
9 11 10	13,453550	FUH	W.87177979	KOVI	0.73900551	YITE	
- HUXS	8:14714345	-C1816-	-0.91301501E-1	£10i	-7:9578454	2178	
FUTS	W. 25559488E-1	91810	-#.19846652E-2	1010	27,865971	LITH	
8204	8.44882732£-3	CORSIG	8.68458435E-6	718	944.43595	HITE	
LAMBAR	-8.298975456-1	24	W. 99857233	TATE	1518.5206	NI TR	8
DRUMA	0.29586572E-1	7,6	8.71525573E-5	KTRBLK	6.7	AXP	1.7663189
X	1396.9626	HUBAR	-418.88676	VXBDOT	-8.64754318E-3	AYP	0.66731189E-3
THE	-470:51056	- SABE	-476:51656	- VYBUOT -	-0-19651163E-2	424	1
842	-16621,216	TBAR	16676,888	VZBUDT	0.34937545E-3	4xP	181.23605
I	-	LOARH	-1566,3721	PU01	-8.63575558E-3	446	16.766643
a I	7.	TAVAL	-5996.4433	1000	2.67754798£-4	424	5.6324750
T	21202.375	- 4400	20101,629	ROOT	B.66985464E-4	RSTR	9.0
A.R.	-311.00462	×	-25.297428	N T R	8.0	PSIDMG	
- ARA	9	- 77	185,97236-	- Y # R -	867,56548	819	0.4
Z=F	155.19584	17	-67.867341	218	-323.84876	HAUD	
LIF	644.51248	ב	-225,29786	4 7	5421.1496	MADO	9.0
I		E	-1978,2643	ar I	-16866,679	VADO	•
L R	2	Z	2781.6654	G - Z	-27499,117	ZAUU	6
XHT	1.6589	XVT	6.5615758	ALFHTT	1.6288468	MADO	8
- 147	8060	- T V V	-102,11345	ALFVIT	12,554338	LADB	9.0

	•					-						1										!		1						-	E-2											1			
	,	- B	6	6.0	9.0		2			. 3	27.425896	42.292229	44.014336	2.3767125	9.9	6					9	9.6		6.9		0.0	0.0	0.0	8.6	1,9105933-	-0.16655710E	-32-117198	134,95721	16,863559	7.9703959	0:0	9.0	8.4	0.0	0.0		6.0		8.8	
		PSITES	VESTR		~	PSTR.	MLVT	PSVI	DSTH	XX	2	X	D.	MIGH	PSTR	E 100	RSTR	TITE	I I	JITR	MHITE	LHITR	DHITR	XITH	YITH	Z17R	LITH	HITH	7112	AXP	AYP	424	V×P	ALA	42A	RSTR.	PSIONG	BIR	MADO	XADU	YADU	ZADD	NADU	CADO	
RUN 31.		2.28	3.01	1077.8	9	-16.8	-16.0	234.0	32, 588608	0.8186632	-3.0	7,1667565	0.1/92552	4.2292229	0.0	3,3119296	-2.8742610			T.2266694B	0.16515222	0.69623934	1.8101640	1.0475258	8.7946285B	-1;439769a	25,389637	958;69496	1584.6521	1.0	0,12561543E-1	-0.52627583E=2	0.43332438E-3	8.10823287E-3	-8.59534868E-3	-8,51260118E=3	9.0	899.88664	-527.23591	5492:1253	-18138.797	-27856,862	-1.4361324	7.0398888	2.8742887
		•	DELS	VSCUND	DELSMK	THOUSE	TESTIR	- FENT	SVT	IMI	13	TH75MR		1	ROTE.	AABP	AA1F	9816	AABL	AASC	BUIL	ERMPX	ERMPA	SIGHT	KOVI	1017	Drot	118	IPIE	KIRDLK	V×BuuT	- toogta	VZBUOT	1004	avor	RUOT	XTR	YTR	Z1R	LTR	Z T	NTR	ALFHTT	ALPVIT	AABUIL
0-8EF-17		360.28688	246.29.99	6.17500000E-2	D. Zabkonobe-1	9.4	5.0	1007.0	****	-1.5624658	6.0738587	17,246756	21,679255	2.7425096	3.6599656	3.3796634		6.3493937	2.0	1.0	9.0	1.4156427	1.7961927	6.51349595	0.07117979	8.91656117E=1	-8.29859686E-2	0.59732579E-6	0.94819876	0.33762786E=9	-610.06174	426.95984	18954.996	-1792,7761		27848,879	5,1546576	-134-1084C	86.774994	-293,47235	2464.4688	3742.8797	0.67156657	132.39238	-0.15752187
-21-11		FSCG	#LC6	8 +0	TIME	NRSS	NSSR	PASCHT	DI1	A13	819	THETAB	THEITH	NIOX	XBACTI	THETAB	Phie	BETAMF	GAMC	OMERAT	PSIUOT	ERTX	EKT2	EPSAT	KUHT	-CTSIG	97810	COMBIC	7 N	V	HOFE	7044	TOAK	LBARH	HOFEE	DBAR	×	77	12		F	-	► (×	444	147
UTTAS(876) 1		199861	6264.8	41567.0	38264.0	1R 29.720006		15.0	766.48668	-0.86546915	6.5886475	17.246756	13,55914	4.46	30.699656	34.95	j	.9785	50.		9	3.92410	19,639941	9	6.36720	••	-		⇉	-	1666,4196	2	2.	-	4	93	~	3		25	-4356.7896	2915.94	2632	20	. 763
		WEIGHT	XI		71	ONFOIR	OMEGTR	2	FSHT	LATSTR	LNESTA	COLSTK		MIVE	XBACTP	VXB	AVB	97A	•	0	œ	ALFEF	CHITPP	FRTR	T. CO	NCX C	n L	820H	LAMBER	CHURN	X		247	E L	I	* I	×	I I	487	# I	ı. I	1 2	L I		147

3	•	•	_	D. D. ELD.	MLVT 273.8	PSYT	DSTR	10 XX 0	N S	** - DX -	dx		D. D. XIOL		2-	ì				9	YITE	21TR	LIT	1		AYP	424-5	- AXA	dia		A TOTAL					
9.991	-5.0	1077:0	9.0	-10.6	-18.0	234:8	32,500000	. B. 31 88824	-3.8	7.5520752	6.603466	4,4575476	2062011	100000C	- 8.06694683E	-4.5854114	0.27105975	0.32303940	0.91983938	8.91596287	0.81627659	1:0720475	24,755723	1867,6076	1017.1013	0.52266777E-2	0137055917E-2	8.67652918E-3	-0.18188469E-	8.12982566E-2	-0.69666396E-	- 1 963-7140	-364,96682	6125,4816	-11507,665	- 44 25 4
•	OELS	CHOOSA	UEL SHH	THOLES	71817R	MENT	SVT	INT	15	117588	THISTR	XCIN	X - 0 -	4144	9814	AABL	RAIL	9016	NX Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z	PERM	X 2 V T	-t101	0101	7 7 7	A TOPY	VXBUUT	-VYBDOT	VZBUOT	POOT	1000	1001	X - X	812	LTR	212	4
364;28666	246.29999	8.175canonE-2	6.284PC988E-1	D	5.8	1878:8	45.8		9.305000	17:614075	21,583468	2.0324412			5.2695393		1.0		1.5824716	B.54289539	8.67177979	-#:92432559E-1-	-8.49248732E-2	#.59658797E-6	8-71525471F-5-		421:45001	19695,399	-2281.6698	-14997.523	50365:973 A AMOSATA	0.00000	379.66284	-366,33176	16766.571	****
PSCG	9074	Pho .	TIME	NG DR	2002	PASCAT	PI1	A18	918	JHETAB	THETTR	MIGH	TURTER		BETANF	GAMC	DHERAT	PSIUOT	EKTX .	FPSET	KOHT	_C1918	CHSIG	918133	72	HEAR	-JUAR	TOAR	LBARH	MONE	* TO *		12	-	F	-
19960.8	568	1587.	-	4:128	137.87949	13.6		8	348	7.01	2.94	1862			7		20.00		-0.98771667	. 5	29.496401	211		6,36479315E-2			-421:45601		1.5	450101411					119.	11549 4411
THOI		A 1		CHE		X	FSHT	LATSTR	637K	CULSTA	PEUAL	MAIN		1	10			1	ALPHF	ERTR	ONF	HUXS -				×		*	Œ,	ac s	K W.	1	ZHE	LHF	•	4114

					6	5		4.571558	6.915961	197867	46.246954	9212165															781804	8.38676194E-1	12.136185	97.60	224001	10.703436							
8.8			9	6	27.5	*04			16.	49	46.	2.49	6	8.8	9			8.8					8:8		8.8		1.70		2	9			9 6		6				
PSITRE	=	STR	SIE	PSTR	HE VT	LASA	ESTA.	4×	90 X	U	Ø.	MINI	PSTR	DSTR	HO TK	14	#LTH	JITR	44118	CHITA	1111	YITH	2178	LIT*	HITH	Z LIZ	AX.	44	474	¥ ;		474	PSIONG	***	MADO	XADD	YADU	ZADD	
123.8	-5.6	1877.0	6.0	-19.6	-18.8	234.8	32.384888	-1.2885797	-3.6	8,2715338	8.6996388	4,9197887	9.0	3,2838338	-4.9227747	-0.2323239BE-1	-5,3454681	8,29737382	8,46182949	0.92132236	1.0003340	6.52658866	2,3851923	24,582415	1236,3213	1922,7186	8.1	-8.1597895BE-1	U. C1027466E=1	-0.34coc146t-c	1 39000000000000000000000000000000000000	- 0. 330c4/04ce		1188:8318	-429.78841	7215.3471	-13516.277	-36386.168	
A	DELS	VSDUND	DELSMR	THOUT	THSITE	W. MT	SVT	IHI	Is	THISHR	THISTH	MCIN	RSTH.	AABP	4714	9617	AABL	MAIL	901L		74843	7	1017	1010	118	I	KINDLA	A x a no	10001	00074				418	218	17.	HTE	RTR	
368,29388	.29999	.175geggee	B. 25000000E-1		6.0	923.2	9.00	-1,6793216	9,5956483	18,451533	22,199638	1,6913961		3.8486592		4.6120618	0.0	9.1	9.		1 - 00 1 00 - 1	6.67177979	8.92687675E+1	-0.46761326E-2	3.59245683E=6	6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6 6	0.71323373E-3	-466.02443	100-100	17151.376	2001 1000	1010110101	-6.2662697	-224 . 79643	585.49636	-491,38673	14345,381	6275.0481	THE PERSON AS
FSCG		010	71ME	SOUS	NSSS	5		81 V		THETAB	THEITR	NIGX	XBACTI	THETAB	DILO				PSICOT	F K T X	7 1 2 2 3 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	-	TS16	9761	DISHD	~ :	: :					K 4	. .	-41	7.7	L.1	Ŧ	-	• • • • • • • • • • • • • • • • • • • •
2.00661	6266.8	41567.8	36224.8	29,123888	\$1.0199			-8,66835581	9.5635467	18,351533	V	4,4571538	17.145520	202.49728	1.25400	10,785450	•	•	2	19649491			0.25429899	8.21658262E-1	21600909E-3	0 1	8.15500674E-1	1950,6768	2/00/0000		2000	18827. 888	-951.00761	-466,65297	-65,416314	172.35185	-4668.4655	95	SAME OF THE
HE IGHT		•	7	DHEGHR	OMEGTR	KFR	FSHT	ATSTR	LNGSTR	COLSTK	EUAL	MIN	XBACTP	A X B	VVB	92A		1	- 1	4 A A A A A A A A A A A A A A A A A A A	F F F F F F F F F F F F F F F F F F F	-	HUXS -	SADE	87DH	T T T T T T T T T T T T T T T T T T T	Ē	# 1 E 3	K 1	K 0	1		A E	AND	J#2	LHF	TH.		

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Color Colo									
12.00 12.0	THE	19968	FSCG	368,28888	-	148.6	PSITRE	89	
A		6266.6	*LC*	246.29999	UEL3	9.6	VXSIR.		
10.00 10.0		41587.8	RHO	8.17508880E-2	VSDUND	1877.8	VYSTR		
N		36224.8	TIME	8. Zabbebest-1	DELSMH	9.0	VZSTR.	6	
137.47949 N.855 5.4	KI	29.726998	2000	•••	TRSITE	-16.0	PSTR.		
13.4 december 14.4 14.5	18	137.67949	9992	5.6	THSITE	-18.8	17.1	273.0	
1		15.4	PASCNT		MCHT	234.0	FSVT	645.6	
		700.40000	BHT	45.6	SVT	32.584008	OSTR.		
1. 1. 1. 1. 1. 1. 1. 1.	T	-1.1377135	A13	-2.0796288	IHI	-2.6254759	××	42.689298	
	TX	16.96465	618	11.377025	18		×B	11.468325	
15.200700	X	17.024686	THETAB	19.024606	THISHR	9.5448882	×C	57.155052	0
9.000200	4	15.200788	THETTH	24.182558	TH75TH	10.664558	A .	7-67-50	
P. 0.500011		4.2569207	KBIN	1:1468329	MCIN	5.7155851	XPIN	2:4291881	
2.0.47802	41	9.0598-11	XBACTI	W.98398411	RSTH.	0	PSTR		
0		230.47862	THETAB	2.4193998	484	3.2894344	2378		id id
## ## ## ## ## ## ## ## ## ## ## ## ##			PHIB		4417	-5.6721763	RSTR		
Colorard		BETANF	4.6197617	9016	-6.21711430	TITE	6.6	-	
## B. 20 0 1 1 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2			SAMC	9.0	AABL	-6.6822689	HITE		
Date	1	9.0	OMGRAT	1:1	AASL	0.34387986	JITH	9:0	
## - # - 1003100		•	PSIDOT	9.0	8811	8.52936769	MHITE		
## ## ## ## ## ## ## ## ## ## ## ## ##		3069190	ERTX	1.4793429	EKMPX	0.9899941	LHITE	0.0	
### ##################################	4	1199/	ERT4	1.6882424	ERMF 2	1.8892338	DHITE	9.0	
## CASSISTANT C. C. C. C. C. C. C. C	_		L	0.49615/01	4	0,61257760	XIIH	9.0	-
### ##################################		53,379471		8.87177979		0.62646275	TITE	9.0	
## \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$ \$		F. 29667343		#: 93388898E-1	1	B:1043983	*117	9:0	
### ##################################		0.249259126-1		-8.45449846E-2		24,583156	LITR	9.0	
131979455 1	1	-8.58983863E-2	À.	4.593558398-6		1686,5453	MITH	0.0	-
1316-294	Œ	6,18478251E-1	72	8.99881479	I P I K	2444.7146	MITH	9.0	
	T	0.13379485E-1	2	0,14385114E-4	KTRULK	•:-	AXP	1,1165268	
		946.2441	HRAH	-936,91988	VXBUUT	-8.44989556t-2	AYP	-8.5474688E-1	
TOAR 19274,787 V2800T 0,28539489E-1 VXP LUARH -18730,686 GD0T -6,16879582E-1 VYP HUARH -18730,686 RD0T -9,284595E-2 VZP CDAR 45542,826 RD0T -9,28455189E-2 RSTR, XT -18,4566 XTR 8,0 ZT 784,87852 ZTR -549,51279 MAOU KT 22267,272 ZTR -549,51279 MAOU KT -18,24285 ALFHTT -5,6272312 NAOU XVT -18,24385 ALFHTT -5,6272312 NAOU		-574:18414	SPAR	579:10414	-44800t-	-0: E9383889t-1	424	-32,127057	
CONTROL		-19197.137	2	19274.787	A SBUGT	0.28539489E-1	d× A	236.47862	
NBARH	ļ	-9691,7562	LUARH	-3073.2451	POOT	-0.166795#2t-1	4.4	14:675613	
CDAR		1334, 5227	HUARH	-16736,606	1000	8.65798585E-2	47A	6.3222676	
Total		44617,667	DBAR	45242,026	RUO	-9,20452169E-2	RSTR.	9.0	
1		-1261,1144	LX	-16.454666	N T	0.0	PSIUME		
### 21 704.87952 ZTR =549.51279 MADD 8			-	-302,53150	-VIR-	1569,7710	- BIR		
			17	764,87052	218	-549.51279	4400	8	
### 22207.272 MTR -17025.737 YADU 8 MI 8442.4288 NTR -46777.739 ZADU 8 X XY -18.248345 ALFMIT -5.6272312 NADU 8 4		1626.8193	<u>.</u>	-668,15827	LTR	9222,7395	XACU	0.0	
3 NT 8442,4288 NTR -46777,739 ZADU 8 1 XYT -18,248345 ALFHTT -5,6272312 NADU 8 4		-6510.5540	Ī	22267.272	MIR	-17825.737	YADU	9.6	
1 XYT =10.Z48345 ALFHTT =5.6272312 NADU 0		-5762.4643	-	9442.4208	R -	-46777.739	ZADU	9.9	
19946194YYTY91		-6.1945411	X > 1	-10.248345	ALFHTT	-5.6272312	NADU		
	1	3,9946194-	TVY	-298,53696	ALF VTI	4.6587647	-6400	0.0	

RUN 35.

	manufacture of the last of the																			10 1 10 10 10 10 10 10 10 10 10 10 10 10											10										depth-dry shift than make may be seen				0-0
6	5	•	•	9.	9	273.0	695.0	8.8	41.461003	A. 2197262	42.241535	44.127929	2 1827974	•			•	• •	9 6		9 6	9.	8.	9.	9.0	9.0	8.0	8.8		8.83776898	8-55100193E	-32,215273	253,39864	21,326423	6.6432341	9.	0.0		6	6		•	0.0	9	
PSITE	VXSTR		* TO TO	VZSTR.	PSTR.	WLVT	FSVT	DSTR	V X	- 6 ×	<u>u</u>	dx	2107		21.00		TITE	1111	2		2 0	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	SHITE	XIII	VITE	ZITR	LITE	HITE T	KITK	AXP	AYP	AZP	VXP	VYP	42A	RSTR.	PSIDMG	818	MADD	XADD	YADU	ZADU	NADD	LADU	
150.0		5	9.1191	9.	-10.6	0.61=	234.0	32.500000	-2.6601108		10.360245	11.970792	7.2251515		3.3854247	56.1142358	-0.35353374	F7.7638701	2 47419744	100000000000000000000000000000000000000		101010	000000000000000000000000000000000000000	0.61766713	0.82653728	1.9519770	24.509166	1834.5744		1.0	-0.22431863E=1	0.305037116-1	-8:35945342E-1	0.15224445E-1	-0.98799490E=2-	0.198458715-2	0.0	1724,0642	-627.50926	10551.791	-19442,328	-53417.255	-5.8890562	4.8342614	6.1244661
>	DELS	2000		UCLORK	TERLER	THSTIR	エニュ	SVT	IHI	IS	TH75MR	THISTR	XCIN	RSTR	AABL	AA1F	8815	AKBE	741		7 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1		EKMP Z	12970	KOVI	LTOT	DTOT	R	K	KIRBLK	VXBDD1	VYBDOT	10097A	PUOT	-1000	RUOT	XTR	YTR	ZIR	7	HTR	Z _ Z	ALPHTT-	ALFVTT	AABBIF
369,26088	246,29999	2 - 15 - 2 - 2 - 2	NI John Maria Co	T-Laggagaav.	9.	2.6	986.6	45.8	-2.5889447	12.446415	26.448245	25.478792	8-82197261	0.56647396	1.5616719	200	4.6456356	6		:		20330100	2 199094		0.87177979	W. 95621943E-1	-6. 4400430NE-2		Į.	0.35762786E-5	-925.66031	20.1	19362,428	-4466.4799	*28179.185	51644.092	-18:558516	-547,39607	951.25112	-757.65863	26419,285	9694.4696	-11.724B46	-342.68664	b.61629634
FSCG	MLCG	CI	2	367	NON		PASCRI	SHT	A13	818	4	1		1	THETAB	PHIB	BETAMF	GAMC		Paront			7147	FEBRE	KONT	CTSIG		SISHOU	72	ن د	HOAR	JUAR	TOAR	LUARH	MUNKE	DUAR	X	14	17	_	Ē	-Z	XVT	4 > 1	ZVT
19960.6	6266.0	91.87.6	5 5000	3055	20027	137.07999	3.61	700.40000	-1.5658594	11.625617	20.440245	13.518497	4.1481665	5.6647.597	255.59864	21,326423	6.6432341	6.6	3		441505155	•	99.1.39.56	9	60.667659	6.517/6164	0.20745566E-1	-6-65119494E-2	8,28867926	0.125559776-1	1958,6271		-14305,919	-10000.004	118.07983	50311.795	-1455,1557	-7.50,43218	-99,495226	1162.6711	-7505.8338	-6547.7365	*6.6344787	-4.7894289	930.61483
WE I GHT	XI	<u> </u>		31		OMEGTA	*	FSHT	LATSTR	LNGSTR	COLSTR	PEDAL	XAIN	XBACTP	2 × >	AYB	97.4	-	. 0	e e	4	- T. T. T. D.	1 1 1 1 1	F . F	410	S I	STOR STORE	SZOH	LAMBER	CENTRA	X X	Y I	ZMK	¥ 1	I	Œ Z	MEN	¥ ¥ ¥	JM2	T.	I	T.Z.	THX	¥ H →	ZH1

					70110	STIE	2.0	
	0,000	477	540.27999	DELS	∂.c.	VXSTR.	9.	
	41381.0	DEX	8.23700rube-2	VSCUND	1111.0	VYSTR	6.0	-
	Seces.	7186	B. Conordore-1	UEL SME	2.	VZS1R.	8	
OMEGHE	21.011744	2222	5.4	TRUINE	16.5	PETE	8	ŀ
UNEGIF	164.764.00	2885	5.2	TASITE	-16.0	41 × 1	273.0	
	13.6	PASCNT	1354.3	MC HT	254:0	+ SVT	9.549	
FSHI	750.4.0000	SIL	47.6	Svl	32.500000	USTR.	8	
LATSTK	-8.5157 4518	818	-1.1812831	INI	28.464350	XA	46.038792	
LNGSTA	3.1372044	215	d. 1127131	ST	-5.6	×B	38,553635	
COLSTR	18,401455	THETAG	14.901435	THISTIR	6.421425	×C	52.634168	n
PEUAL	10.55:057	TERITA	28.455784	1H75TR	14.935784	7	30.788319	
MAIR	4.8035742	XBIN	3.8753835	xCIN	5.2634109	MEAK	1.6625239	
KBALIF	45.475624	XOACTI	4.5496269	FULT.	20.00	PSTR	6	
	W.106255266-1	THETAB	5. 35658884	AAFF	4.1366556	CSTR	6.0	1
	2.5	1110	-K./139788	VAA1P	-2.2111519	E COL	0	
	8.1374418tt-Z	BETAMF	8.0	8611	-1; 31#7922	111x	0.0	
	\$ \$	GANL	2.	AAGL	-1.6950609	HITH	6.0	
!	- L. B.	DHERRT	1.0	BAIL	-4:13551176E=1	J178	9:0	
	0.0	PSIUUT	. e.	1144	0.10354457	HHITE	9.9	
ALFXF	-16.533461	ERTX	-0.25661120	ERRTX	0.199953936-5	LHITH	8.8	
CHIPP	-4-1454949	FAT2	0.20475116	LKMT Z	0.1109955BE-S	CHITE	8.8	
F	2	LP5#1	******	SIGNT	0.0	X I TR		1
	A. 554:11456-6	L I J	N. 87177979	K C < T	6.64852013	*11*	8	
コー・シェンモ	で、よいでもよりなどれしま	-61819-	- Dantololyrot-1	Liai	13:656564		0.0	1
MU45	\$.	518HD	-0.3511/0756-2	0101	25.232740	LITE	2.0	
670W	で。 ケションテンタがヒーケ	CUMBIC	0.667/1055E-6	<u>a</u>	1317.4224	HITE	9.0	
LAMBER	-6.306/6571E-1	711	6.99437635	TAI	1926.6861	Z TZ	9.0	
たれていまつ	0.504-1555t-1) A	0.11212716E-3	RIRBLA	6.19544499	AXP	3.0190747	!
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	150.555.01-	TA D	18064.685	V28001	C.16658214E-3	VAP	0.16825686E-1	
	-1317.7115	LEAKI	-5565.5420	1001	-C.46695517E-5	446	6.9	
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		1403	598c2,352	RUUT	4.196164dBE-3	RUTE	. 6.6	-
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			-0-41446466E-6-	-41K-	1234,9456	-8TH	4.6	
	8-63/02:00E-3	17	5.9459734	218	-451.50347	MADU	9.0	
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	-1, 15/56467E-5	7	172.93656	# T	-13182.685	YADO	6.9	
	2.	ī	W.15256122E-4	œ	-38417.588	2400	8	
	4.7512515	I ^ X	-c.1105158Kt-6	ALPHIF	-10.699429	MADD	8	
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£0.43	c. [+645] c. c 1	-	AJ.	1010	24.575118	L114	8
5704	E. * 11/010512-4	PINTO IG	•	œ. L	1416,5155	1	n.
3	2-16:256 16 500 5-	7.	6.99947796	1 1	1935,1232	****	0
Serve	E.110134951-1	, *	E.71565573E-5	A TESTA	7	die	1.9865274
1 1	すらちゃ ひとしゃ	A D	-1843,2557	TOOSTA	-6,282451165-1	44	2.16698
1 1	7	49.62	399.22497	****	-4:45485838E-2	474	3417
1	٠	1040	19236,412	-	-2.41929991E-1	4	236.19692
1 1		L3447	-4456.54.7	5	-2.4415d5b9t-2	1 >	
1 1.	-/411.1-12	4	-<1729.535		-0.9001d562t-2	474	14,124114
1 1 1 7	11100.101	-4:0	30911.969	•	E.24953125E-2	ことから	0
a a	47	· ·	1.6692.	# H	3	PSIONS	3.6
A R	まかいまってもいいま	-	100	**	1331 = 1941	414	4.4
202	-53 25/54	17	456.64539	Z 1 2	1464.11415	400	60
, E	4111111	۳,	-591.16122	را اند اند	5131,5557	204	6.
4	#21.14.61.F	<u>-</u>	21164.614	1 1	-15011.863	TADO	N. N.
	-1.500	- -	1930-1660	X F	-41644.575	0047	20
<u>.</u>	4) P	-25.247541	ALFHII	-5.6753495		n.,
	921114991				3,3748514	E # 92	0.00
-11							

WE 1GHT	3.306E	FSCG	360.27630	A-	150.0	PSITRE	6.6	
ı x	1 00 20	3. C.	246.29499	CELS	,	VESTR.	•	
1	5.725.12	RHO	32	VSOUND	11117.0	VYSTR	•	
71	Seces.	11116	F. CUNCOONDE-1	UELSMR	9	VZSIR	0.00	
ONEGER	606610-12	SSAL	5	TELOIT	9.0	PSTR		
UMEGIK	164.066.5	2888	2.2	TESTTE	-16.0	1	273.0	
KFR	13.81	PASCNT	545.8	MUHT	234.0	FSVT	-56	
FSHI	low art ou	SHI	45.E	5 v T	34.500000	USTR		
LATSTK	-1,7014538	A15	-2.7688096	IHI	-5.6884866	× ×	39,565913	
LNGSIR	13.0000	818	15.684911	S	-3.6	×	1.2172113	
COLSTA	26.00257B	THETAW	20.8425.18	TH75HF	10.762678) X	64.766739	
PEUAL	A548612.0	THEITA	<1.854542	THISTR	8.3242420	ď	52.498884	
MALK	3,9305913	MEIN	D.12172113	XCIN	6.4766739	MIGH	2.9954298	
XBACIP	10hussu2	XBACTI	-4.686w#382E-1	RSTR.	2	PSTR	5.5	
VXB	255,04000	THETAB	3.1818554	AAGP	4.0124304	DSTR	6.6	
V V	17.669755	Palo	9.0	AA1F	-8.8658286	RSTR	6	
47 A	14,670379	BELLAND	3.3652548	RBIF	÷0,44662349	TITE	6.6	
a .	2.5	5 Å 4 C	3.	AAGIL	-8.6268362	HITE	8.0	
o	2.6	DMGRAT	1.2	AATC	0.44952035	-JITH	0.0	
r	3.	PSIDOT	\$.	401L	6.84590151	MHITE	9	
A L T :: P	1.4567.464	ERTX	1.5412553	FREFX	3.84614260	LHITR	8	
77.10	14155	EKT2	1.678/666	LKHF 2	1.0103506	CHITE	9	
۲ . د کا	S. S.	EPSKT	Ø.4F565254	SIGHT	8.59228487	XIIX	6	1 1
¥ .	80. 536011	FEDY	2.81117979	KGVT	0.85221584	YITR	9	
MUXS	E. Save Cray	51813	# 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	1017	4.9574282	ZITR	8.0	
3 L L	こ・11045045111	CHSIG	-v.89655100E-2	DIOI	24.688271	LITE	8	
SZUM	V.11849596E-2	DISHDO	0.64451969E=6	418	1615.1310	MITH	6.0	
LAMBMR	44451515E-2	2.1	0.99843649	TETE	2171.7298	RITA	8.5	
DESHIRE	0.162437161-1	٧ć	M.715255731-5	RIRBLE	1.0	AXP	1,7512953	
X:X			-2006.3996	VXBUDT	-0.25689834E-1	AYP	-0.10429558	
X X	-486.54584	ר	465.35.394	- TOOSTA	=0:49632113E=1	424	=32,140161	
HW2	-1465/.641	TUAK	19367.000	VZBUGY	-0.17732964E-1	VXP	253,04066	
r r	-5110.2813	LBARN	-4679.8183	POOT	-0.28621096E-1	VYP	15.209756	
Y :	-12456.103	REALE	-23058.565	1000	P.46496791E-2	471	14.070379	
X Z	45127,215	DUAR	44286.196	*not	0.11705986E=2	RSTR.	- B • B	
¥ X	11454.4000	Į.	-94.634451	XTR	0.0	PSIUMG	6.0	
× ×	-711.37796		** S68.90397	TTR	1517.8399	BTR	4.0	
447	-440.01160	21	1184,2084	218	-552.44948	MADU	0.0	
د د	1165.7017	֓֞֞֞֞֞֜֞֞֜֞֞֞֜֞֡֞	1350.43485	7	9272,0273	XAUD		
£ :	12144.5504	=	55502.126	Y	-17116,726	YADU	8.8	
	16684.3591	2	12274.347	* - 2	-47027,728	ZADU	9.0	
I I	-64.24330	- (-45.786145	ALFHTT		NADO	8.0	
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- L J	1104.1540	1 ^ 2	U. 74287865E-1	AAGRIP	8.8762639			

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•	71mE	. Zuadobeez.	DEL 3MR		VZSTR.	0
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.62888	9992	N. 8	TROITE		-	273.0
	PASCHT	1112.8	BERT	1.0	PSV	8.50
. 486.68	LIO	45.6	F > 8	8222	DSTR	
.25555010	. 418	-1.3975883	THI		× ×	
3516	913	1.5697983	1.5	6	. €	40.545.05
179518	THETAR	21.179518	TEACET.	1 . 9994.		
21.482943	THETTE	34.242365	THISTR	28.742345) Q.	40.000.00
422686	KBIN	3.9564128	KEIN	A 5871989		されたのかのかり
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2.0		-8.39785213E-2	ptot	26.685217	LITR	0 N
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3283924		E-99461136	I	0000) d
16284783E-1	u	2.12781282E+3		70,0000		2 7419111
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13.23119	JOAR	485.23115		C. 249646664	178	
-18476.881	TOAR	5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5	VZBDDT	-6-1885/4886-2		9-14-44-44-1
674.4494	LBARH	-3921.2377		8.992883735+6	. >	4
-	HOARK	-4622.8445		8.18912559E-3	474	8.144849775-2
108 622	2888	46965.881	•	F-4555555		
	X	4.5326712				
		-8-349835178-8		1819-9878		
717842456-5	12		Œ		00	9 69 9 7 69
	-	-E.77668327E-6	11	8874.2758	MADO	
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•	-	2.973688138-5	# Z		7.400	
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36-406-17
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UTTAS(876)

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₩.	2	OI C	. 17 Secree	A SOUND	1077.0		8.0
71	36224,0	TIME	8.2000000E-1	OLL 3MR	8.6	VZSTR.	8
OMEGER	27,011999	2000	4.0	TELOIE .	10	PSTR.	0.
GTR	124,62888	0002	5.0	THBITE	-16.8	17	273.0
	19:8	PABCHT	106.0	MENT	234.8	F3V1	0 4 W . E
F CONT	706.48686	PH.	45.6	S V T	32,300000	DSTR.	
¥	-2,1118964	A 1.8	-3.6744286	111	31,566110	W X	36.660647
LYGGAX	4,2685735	818	4.4329725		-2.0	×	6463
COLSTR	19,024095	THETAB	19.024095	HVSH	9.7448953	U X	
PEDAL	14,592477	-	25,606352	5	2.38		1.
KAKK	3.6600647	XBIN.	3.4846829		3.648539¢	MAN	2.222863
XBACTP	-	XBACTI	4.1459425	RSTR	9.0	A LOG	8
VXB	67,387914	THETAB	.588936	104	4,3226695	E 100	
0 × ×	50	9110	-1.6793969	AASF	-2,2761987	RSTR	6
97A	9,5143365	BETANF	D. 0	9017	-B. 34838161	4114	62
۵.	9.0	GAMC	9.0	AABL	-6.2976218	HIIH	8
0		DMGRAT	1:0	4	E.20227894	JITE	2.2
Ox.		P81001	9	9611	8.65421685E-1	Z Z Z Z	9.0
ALFAF	-10.661514	FRTX	0.97818378	MARK	0.69254940	LHITR	0
CHILD	•	EKTZ	ø	EKEF2	1.0349923	A - 1 - 0	0.0
EX-R		EPSH	666697.	916#1		XITH	0.
	7.9519971	X OX	6.93491845	× ~	0.64652813	YITR	9
	B. + 3284727E-1-	C1816	.11159621	LT0T	85	211A	6.6
	9.0	43	8.36558843E	ptor	30,001272	LITR	8.6
	N	COHOIG	8.94352743E-6	118	1162.6247	HITE	e. e
LAIBIR	-0.4396262E-1	7	•	X I d I	ŭ	-	
CINIO	B.46357315E-1)	0.14213562	KTRBLK		AXP	2,5326545
X IX	1620,6327	TOAR	-624,26542	VXBDOT	8,29748692E-2	AVA	8.91491764
I >	545,89594	JUAR	545.89594	**B001	0,26248981E-1	424	
Z 1 2	-18995,496	TOAR	19855.761	YZBUDT	8.526737996-3	4 × >	67,367914
T T	-6650.2716	LBARK	-1589,1654	1004	E. 79995752E-3	4 >	
T T	15605.066	EDARI	-6676.9498		-8,287575595-3	VZP	5.3143565
T Z		M 4 0 0	34737,781		B,18929961E=3	ROTE.	
L.	-158,67815	×	32,488864		8.	PSIDNG	6.6
E P	-	-	-5,5032776	æ	1892,1798	918	614
2 ii §	386,28732	11	65,735587	21R	-397.73983	MADD	60
- H		ב	-12,422793	•	6675,4682	9	0.0
X	•	ī	1988.9881	ď	-12323.305	YADD	0.0
3		r Z	155,77345	Œ FZ	-33857,937	9	6.0
H I	33,684168	X × T	-1.3962956	ALFMTT	-1.9713439	MAUD	9.0
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		2.2	2.2	2	-	M	8	43.2	2		-		-		2	2.2	6	6	2			2	6	2.5	2.0	8.0			2.11	- 32,	•	•	5.18	3.3	69					2.0			
	9 0 1 1 8 2 × 2 1 8 2	STR	2378	0	>	•	SSTR.	-	0	UK	4	-	5	8 T 8 C			T T			T I		MITH	MATA	4112	LITA	N T H	3 L I Z	AXA	4 4	424	0 × >	4 1	424		PSIDMG	818	9	0	a	COVZ	9		
en zo k	N	12		-:0.8	-16.8	234,8	26538	2575	, N	19452	12.476342	-		4.1689293	-1.4526829	·2.24434441E-1	924	2.23417631	-2.12539813E-1	,	1.2143679	1.2282688	8,76237277	61286218 ··	27,337912	939,73678	1378,6902		-2.446B4523E-2	2-81945188E=2	E.11826538E-2	2.15821422E-2	-E.48167169E-4	E.43749456E=3	8.8	\$82: 3168	-322,43726	J.	927.284	727	5495	16:033636	452626
	() ≪ () () ≪ ()	SOUN	DELSHR	1	#91T	FLET	3 v T	H	50	I	TITSTR						7244	AIL	اا	X M M		1527	_	101	•		re-	KTRBLK	10001	10001	18001	100		100	OX.			114	Q.	# L 7	ALFHTT	ALFVTT	AABUIF
36-800-77	368.23888	.1752228	- 322883335-	P. 4	•	1553.8	45.6	-1.9965146	5.8766767	18,999452	23,976342	3,2553584	.711421		6	7.7441239	63			249727	955	.52	071	.11811818	8.411	- 98996988E-	99857626	8,35762786E-S	-360,68732	500.4450	16863.377	261.944	-3448.9689	•	-58.815627	5	•	52	-4207.6632	63.631		3.3114	-1.5376659
-21-17	# # 0.00	ar CO	TIME	めのむと	33	PASCAT		200	==	HETAB	-	210	BACT	THETAB	0110	BETAMP	SAMC	DEGRAT	PSIDOT	ERTX	EKTZ	FEGE	TO Y	CTSIG	CIBIC	COMBIN	24	U A	4	1000	BAR	LBARI	440	5	×	- A A	17	-	¥	-	- ×		1 ^ 2
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	158	OIL		24008A	1877.2	VYSTR.	
نے		1111		DELSPR		VZSTR.	
WEST P	27.619999	5000	1.1	TAG 1 ER	.10.8	A 18	
FESTR	124.62868	8002	3.6	-	-18.2	MLVT	
*	15.6	PASCAT	1190.0	H L H L	234:2	FRA	
F 11 2	728.42868	F E	45.6	3×1	60	2100	~
LATSTK	~	P 1 8	-1.9874256	FIN	11.231911	•	43.543657
4587R	6.2627498	913	.919368	23		Q ×	27.676164
SLSTK	•	THETAB	6.96942	1	-	×C	53.858928
EDAL	-	-	22.23564	THISTR	.5556	• 1	49.721562
11.7	. 354365	2101	.787814	XC14	146528:	MIGH	2.6298159
BACTP	-	X SACTI	3.1666271	B B T K		4184	8
•	34.9761	THETAB	.226989	4244	4.1237845	X 1000	
20	4.849	PILE	8.0	A 4 1 4	-2.4435116	3 L 60 C	2.0
9.	.612374	BETANF	9.2935397	0916	E.93327841E-1	1111	
•	8.0	SAMC	9.0	1244	4.7915974	I H	
				AAIC	66263532.3	- 31TR	1 •
		31		5616	E. 66913994E-1	-	
A R d	4.23627	ER TR	1,3921697	EKMPX		11110	
TIPP.	2,141	EKTZ	1.8647238	Exat 2	1.2296383	NI NI CO	
46		EPS#7	E.56293339	B1611	£.91637412	M I T B	8.0
	2.38662	TOT'S	B. 87177979	K G × T	2.61519296	* 1 1 4	9.0
8 1	.10646744	C1816		1101	-1.7678262	2179	
0 4 S	.19388538E-	49	-e,34e19237E-2	5701	24,756023	L178	
	.756986386-3	918100	8.90011595E-6	4	989,93346	4111	
1	0,24252235	24	8.99846366	Y	1329,3964	# L I Y	6
GERM	.24991184E-	C)		* TBBLK		414	1.0121601
•	566.186	ILLA	-568. 95834	VEBUGT	25.	4 4	-E, 64856562E-2
*	59.4217	JBAR	459,42177	- VYBUST	4	474	-32,126245
•	3.6	TOAR	10841,296	VZBUD1		A H A	134.47611
×	479.796	440	-1349,1199	1004	-2.45626338E-2	4 >	14.849855
•	9925	4	-6591.6728	5005	2.24476364E-	42 A	7.6185747
œ I	25603,032	9848	26551,249	P 001	5766748E-	PBTR.	
-	14.9770	H	-17.677112	a L H	2.2	PSIDME	0.00
	-201,03292	- 41	-117,37152	414	855,12321	978	
1	49,922106	17	-53.743896	218	-311.24888	MADO	e. 0
•	49696,064	-1	-257.01745	4 1	5223,6917	MADO	
_	1,226	¥	-969.32766	2	-9643,2525	VADD	. •
	-2491,3548	r 2	2	a ÷	-26494,567	2400	
F	241	X V T	-	ALFMTT	2.61877956	100 A	
	2590	***	-116.54562	ALFVIT		£409	
	A 88026	7 . 7	-	AABRIE	445293		

	UTTAB(878)	1-21-17	38-406-77					
ME IGHT	19988.8	9284	368,28222	•	2.221	241184	Anne de des - séphiles qu'empiles séminares :	į
N P	6266,8	937=	246.29999	2730		VESTR.		
11	41547.8	OF	-3382886	4 800%0		VISTR	. N	
71	30220.8	TIME	B. Zuerenene.	DEL 344				
DACEME	-	1988						
DMESTR	124,6222	2000	3.6	7=8178		-	272	
***	15.8	PASCHT	1178.6		2 2 5 C	244		-
	725.48202	- 10		1 2		4		
LATSTR	-1-1181388	A 1 A	-2 - ETABT : T		5745V-7 6		24 00 00 00	
NESTA	A - 444 A A		442474		C. 6061100		20010200	
COL STR	10.10.10.	77674		787988	10101	D (77.50.00	
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_	001000112		2.1004160			1 W	w. w	
2 1	166,54667	THETAB	4.8871615	4217	6.1176163	K 100	2.2	
0 L A	13.676693	DI I		4410	2	2130		
87A	11,011597	SETAN	6.2915118	4166	8.11192172			
	9.0	SAMC		7277	-5.2224270	-		
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1	-465,62426	2000	405.02426	Trougra	-E:31936368F#	424		1
	-10065.179	1967	10001.001	100624	B.17376592E-1	4 × 4	168.59867	
	-5958.8469	LOARH	-1672,4769		2.512961692-3	4 >		
	7926,4675	INVE	-11854,585		2.96179172E-3	474	1150	
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	11149(876)	1-12-1	36-400-77		* N. P. T. D. E.			
MEIGHT	19989.8	P 8 C G	308,2888	1	122.8	PSITRE		
1 x	6266.8	937*	06667.4	DELS	8.8.	VASTR.	•	
1	1.	P	8.175eedesE-2	ASDOND	3.7751	VYSTR.	•	
11	2	I	Secreta	17		. BISZA		
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03	18.901518	Ī	22.977398	THISTR	•	•	51,215595	
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•	15.667398	•	1.5687369	B G T E	2.5	2 00 0	~ ~	
-	265,54562	THETAB	2.7683863	4244	125 4.2823371	2010	٠. •	
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67 A		-			E. 93369974E-1	7178		
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æ		-	2.5	100	2.22399142	A T I		
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2 4 8	-16968.866	4	18975.688	VZBUOT	.21726296	4 11 >	•	
* 1.	-7317,e450	1	-2866.6813	1004	2.63103010E-3	4	14.144726	
T .	8621,1462	3	-18961.284	1000	.25372441E	471		
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4 11 1	396,5922	-	-193,34895	- 4	1151,5257	910	The same of the sa	-
4 2	588	17	329.34968	412	-422.67682	a		
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Œ.	15.0	•	1385.8	HIMI	234.3	PSVT	2.564
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•	57.3061	4	-661, 56612	VXBUOT	0.16449117E-2	ATA	
·	55,098	ĭ	615.84614	VYBDDT	E. 30837896E=1	424	•
ZHE	16976,01	TBAR	19840.227	vzBuot	0.28857812E-1	0 × >	236,52902
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œ	11302,533	A	-10132.747	1000	-0.12125586E-2	47 4	4.6517224
Œ	9156.03	ž	50760,612	P00+	0.177830551E=2	BOTA.	0.0
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_	-532,64871	11	-261.66813	Y * R	1660.7147	BTR	
•	2	17	535.24574	218	-627,35567	MADO	
L	7.33	<u></u>	-973.84739	118	10193,678	MADO	
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×	4729,842	← z	-	* L Z	-51702,544	0	•
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<u>-</u>	.501617		-257,89831	ALFVTT	3.9768942	Ω	
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-1096-8587 JOAR 1996-8567 VY0001 0.35627386-1 AZP -32.256259 -12966-237 TOAR 19156-613 VZ007 0.3662731166-1 VXT 2533.49274 -12966-237 TOAR 19156-613 VZ00 0.00 0.3662731166-1 VXT 2533.49274 -12966-237 TOAR 1916-904.25 POOT 0.362740526-2 VXT 10.357200 -1437-775 FEBAR 63334 XTR 0.3662136-2 VXT 10.36274 -1437-775 FEBAR 63334 XTR 0.3662136-2 VXT 10.3623434 -1437-7817 XT 10.363343 XTR 0.366213 FADD 0.368 -1437-7817 XT 10.3633437 XTR 10.3542-3662 VADD 0.368 -1537-7817 XT 10.3633437 XTR 10.3542-3662 VADD 0.368 -1537-7817 XXT 10.363347 XTR 10.3525033 YADD 0.368 -15362-4663 XTR 10.3633437 XTR 10.3525033 YADD 0.368 -156711310 YXT 10.3623671 ALFRIT 14.3623633 YADD 0.368 -156711310 XXT 10.3623671 ALFRIT 14.3623633 YADD 0.368	~	1592,4684	HOLE	-396.4843	VXBUOT	-B.21247699E-1	AVA	1956719E	
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- ZIVX	4,3777156	NICK	1.2741350	XCIR	4.5162915	META	2. RZESTTE	
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YEZ	21e4e,164	COAR	60340.636	FUOT	P.10053456E-3	KSTR.	0.0	
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1-21-77 S0-AUG-77

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82.4	45662	EKTZ	1.6779347	アイエイ 7	1.0095531	DHITR	0.0
9		EPSHI	6.403b9476	LMSTS	.5824678	XITR	0.0
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6	23	DISHBO	0.65847518E-6	TTR	1415.4938	HITE	0:3
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341 . 7619	6194	JUAR	397,76794	900	0 551572367E-5	- 1	-521172627
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-16+14.113	TOAK	19251.568	VZBUOT	-8.13471720E-2	4 × >	67.391243	
-5554.rold	LUBRA	-1463,6676	Puci	0.695/8148E-3	1 →	9.9	
14501,405	TOPET	-6957,1765	1005	-2.<0397178E-3	47 A	5.2076417	
21170.756	C544	26661:696	1004	-e.75261841£-3	RSTR.	0.0	to a distance of the control of the
-144.41154	-	6.3525415	ж 7	2	PSIUME		
1	1 4	-1.3846545	- Y T H	964,17514	- trra	- 10	
104.51411	12	34.193866	717	-524.31185	MADU		
3.6	,	-14.6enr62	1 10	5557.5556	NA VO	0	
-515. 4651	± X	945.645.849	7	-12463.178	VAUD	0.0	
2	-	211.66716	414	-<8632.949	10 V Z	9	
14.11160	X e T	-1.9951588	ALF n 7 1	. 56	NAUU		
¥ .	IAA	7 25556345	ALFVIT	-6.8-	4047	9.8	
51.50000	Z + T	1.6949544	AADBIF	2.6543918			

WE IGHT	13962.2	P 500	350-20302	A	15.0	PSTIRE	5 6	
, ,	•		00001 400	9		• •		
- 1 - 1	7 - 2 - 2 - 2 - 2	۵ ۱ ر	******	2		A X S I K		
.	2.19619	10	5 - 23 1 6 2 5 5 FE - 2	VSDUND	1111.2	A LOLA	8	
71	30cc0.	7 F F F	P. Crevaroue-1	ULL SAK	9.	3	2	
ひとないないの	21.11444	2000	0.0	TECHE	-18.0	PSTR.	8	•
C 2 2 1 4		2000	5.6	TASITA	-14.6	#LV1	273.10	
244	13.2	PASCHT	1277.5	ML MT	234.9	FSVT	645.0	
E n		SnT	2.0	L ∧ Ω	32. SUCCED	QSTR.	5	
LATSTE	٠	A1S	-1.6552692	1	14.517445		45.661458	
LANSTA	5.201.5/0	010	5.1234212	15	5.5	מ	95450	
CUL 31x	10.113972	THETAN	15,715972	27472	5.534722	×	35.952327	
PEUAL	13.15/115	1nc 11x	23.447886	177514	9.7475564	Kr	31,693341	
TAX	4.5°51458	TEIX .	Z.en24961	TUTE	5.8762327	XFIR	2191902:2	
XOACTU	\$4.	ADACTI	3.2434756	3101	**************************************	FSTR	2.	
CXA	101.1.523	THE TAB	3.7211185	4044	3.9477387	SSTR	\$2 \$	
-	16.421407	0110	6.0	AA1F	-2.0161555	* R B T *	20.00	
27 A	6.3543963	BETANT	6.355116	4100	E. 44100485E-1	TITE	8.0	
a.	2	SAME	2	AABL	-4.5849524	HHE	8.8	
5	200	UNGKAT	2:1	BA 11	e.equatost	J119R	0.0	
¥		PSIDOI	9.	Boll	8.11604274	E I	8.0	
4447	•	EATX	1.3653168	FAHTE	0.98E53E79	Lulia	9:9	
72.17	_	EKTL	1.8474976	ERMP Z	1.011/861	Z 1 1 3	8.0	
4 4 4	J. Q	EFSHT	#.513v5517	Stent	1.6416641	KITH	9.6	
ر ۲ ۲	10.054414	7 E 3 T	0.81117414	A C < 1	E. 79561581	ALTA		
EXG	- 2:13437545 ·	-C1518-	0:615e3516E-1-	1017	5:554921e	-217K	6:0	
2010	8-1/0243476-1	97843	-0.c8365125E-2	1017	25,515281	LITE	8	
4704		CLMSIG	C. 63412190E-6	-	614.71355	¥	9.0	
11011	-0.6733446	7	6.94167139	17 27 27 27	1175.6559	X L T Z	9.	
111010		٠ • د	9.33162186E-5	RIRBIR	1.9	D X D	2.8988881	
r I M	1245.455.	HOFE	_	vxeuot	-0.13364567E-1	ATP	-2.59639441E-3	
71.	- 204: CDG19	140P	-504,26879	- 448001 -	-0:2246442#E-2	424		
Y 4 7	-14061-414	TOAn	10803.700	100974	.62022825E-	V X P	101,16342	
7	ナルジナ・シロアサー	COATI	-1417.6652	PU0.1	-8.25578157E-3	444	12,954467	
Y S, E	14155.150	IT	-7115.1114	500.1	d.27489685E-3	d7 A	6.6345625	
r E Z	63101.630	400	23958.414	1008		RSTR	6.9	-
A R	-521.454.54	×	-17.240102	X X	6.3	FSIUMG	3.	
444			-145.54621	-V T.R-	705.63760	*14	2,0	
107	90.144.00	17	-39.55541	714	-278.66962	4400		
T,	516.15167	_	- 434.47546	ر ب	4677.6472	XA06	9	
ĭ		ž	-1142,1912	r r	-0654,1136	YADU	8.0	
4 E	-6554.1516	<u>,</u>	2959.5501	1 - 2	-23721.986	7400	0.0	
i H	-10.035410	X < 1	4.94581ceb	ALFATT	2.8275223	NADU	8.8	
1	-1-11-11-11-1	1 A A		-AEF > 11	-7 : 6456283	-E#00	0.0	
147	-34-5105-1	1 ^ 7	-8.43686561	AABUIL	2,5164951			

33-AUG-77	
1-51-11 5	
U1 [A . [S]E]	

RUN 29.

			Į	-		
יי ני	56652° 48	חזח		A H S H A	8	
ב ור	-23782587E-	VSGUND		VTSTR.	89.	
TIFF	6.2368drude-1	UFLSMK	5.	V 251K.	20	
2000	4.0	RELOTE	-16.3	PSTR	8	
2050	5.0	TESTIN	-18.3		273.0	
PASCAT	1322.0	#LMT	234.0	PSVT	695.8	
SHT	10.6	5 × 7	54.580B5B	CSTR.	, s	
A15	-1.3459529	InI	2,56252789	XX	141	
818	8.1439654	12	3.6		16.577201	
THETAB	15.812149	THISTR	6.7321494	O K	39.575934	
THEITH	24.481246	THINTE	6.5412266	7	41.454181	
MBIN	1.8577281	NCIN	3.9575934	PICK	2.2384575	
TANKE	2.2011980	KO LX	9.5	T	3	
THETAB	5.3592774	ABAR	3.9492476	GSTR	0	
PHIC	5.0	A 4 1 P	-5.1259821	#51#		
BETAMF	4,6932983	5015	82351676E-	1114		
6 A M €	2	AABL	6346462	r I		
-D-IGRAT	1.2	AAIL	BES48862.2	JITE		
PS1001	2.0	9011	0.35862232	3 - 1 - 1		
FRTX	1.4071622	MARKE	5	LHITE		
EKTZ	1.6799861	EKAP 2	1.0655500	GITTS		
FFSH	8.47693291	316#7	0.64682838	XIIX		
F 1 2 4	0.97117979	>3	0.82610750	Y11H		
CTSIG		LTOT	2.5434116	Z17R		
CHSIG	-6.19344284E-2	בונו	4.51865	*111		
COASIG	8.55433846E+6	1 T R	624.67256	#11#	1	1
71.	7.97553448	X E A		1 1 2	8	
۲ د	A. 71525573E-3	RIRBLA	8	AXP	2.9866327	
-	-1568.5877	vxbudl	-6.51768114E-2	AYP	-2.18546389E-1	
JUNA	325.18889	-TODDAY	-c.182536142=1	424	-32.629499	
TOAH	19066.195	VZBUUT	6.21246476E-2	3 ×	154,68519	
Lakal	*1611.519E	Pugi	-5.494978tet-2	d A A	12.276953	
TYAGE	-15455.217	5001	8.11655551E-2	471	12.588529	
E CO P K	25666.498	F004	*8.24742914E=3		8	
×	1.3415611	ATE A	8.3	PSIUNG	0	
A	-142-11436	TIME	779.88313	BTR		
17	304.95179	2 18	-265.65479	400		
<u></u>	-515.23249	a L 1	area.0725	MADO		
Ē	8059.3858	dr	-8794.7577	YAUG	•	
2	5455.6176	a - 2	-24165, 578	ZADU	•	
¥ ← Ţ	-3.9785139	ALFMTT	-4.6366154	NADE		
TAL	54912.281*	-BEPTF		CORT	1 .	
7 . 7	200000000000000000000000000000000000000	3		1		
	THE STEMPT OF THE STANDS OF TH		2	2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.	2.2.3.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2.2	2.2762222222 2.27622222222 2.2762222222222222222222222222222222222

FEE TON	J. 12661	F 3C6	22362°248	A	8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8 8	PSITE	2.2	
	6400	* LC 10	4	LELS	2	C X C TR	5	
	41547.8	DER	8.2379828PE-2	VSCURD	11117.8	YYSTR.	e .	
	30cc4.n	114	oucooct.	UEL SPR	\$. 9	VESTH.	20	
222410	21.019339	ののよこ	6.0	TROPER	*10.8	FOTH.	60.	
UPELTE	100.0000	.555	2.0	TISITE	-18.8	1 / 1	m	
	15.3	PASCRT	839.8	FLMT	234.0	PSYT	595.2	
FEST	700.4446	P#S	7°C7	5 , 1	52. Sedebe	SOTE.	(V)	
LATSTE	-0.45152011	A13	-1.3596761	F-1	-1.7247120	4	.61464	
LINDSTA	3775004.6	918	9.5019332	15	2.5	D	_	
COLSTR	11.333540	THETAS	17.355626	TI VIER	7.2739239	×	96616	
PEUAL	13.440135	THETTH	24. 234606	T1757R	6.7544851	d	42.038638	
MAIN	a. a. 1 a. a. a	FRIN	1.3026197	xcin	4:2968162	- XP FR	7. 3127938	
XBALIF	1.55.15	KBALTI		FS1K.	20	B	1	
	150.47500	THETAB	4.5077631	734	3.9614248	N SC		
	11.49de15	1 0	2	414	-5-6326548	FSTR		
	13.200449	SETANT		5011	K.12111235	7114		
	5	GAML	9	1044	-4-6913126	1	•	
!	2.3	DMGRAT	1 . 5	BAIL	T. 32184F3F	ALT.		
	٠.	PSIDOT	8.0	6016	E.42414765	FHITE		
	6.73.72160	ERTR	~	FRHTH	E.92859487	LHIIR		
CHITTP	54.10:11.0	7143	1.6435592	FKnF 4	1.2070150	G1110	•	
FATH	7.	FPS#1	2.400000	SIGHT	B.64749357	XIIX	9.0	
	Seernrace	761	U.B/117979	1 × 5 ×	6.93168449	4 T L K	9.	
	F. F 55.1199 -	1	+: e2166557E-1-	1013	10	Z11×	6.6	
B C ₹ S	0.1305E364E-1	CHSIG	-8.51824551E-2	010	24.555673	LITE	0.0	
•	E. 013347666-2		r.6499rerae-6	116	963.66141	****	e. e	
LAMONA	ت	7 2	c.99661716	1 1 1	1667.1661	at I z	0.5	
CIOITI	たっしょうしゅうしゅしょ		6.11525573E-5	KIRDLK	2.0	D H D	2,5184483	
	2450.1510	T T	-1454,5365	VXEUGT	-F. 65919388E-2	4 4 5	-8.27510751E-1	
-	-360:11096	140	3c8:17692	- * YEUGF	-+: 13286364E-1	474	-32:263499	
	-16751.516	4 4 0	15063.999	VERDOT	E.19549162E-2	4 × >	168.47868	
	-5551.2534	しちをひれ	-<< 11.5081	1001	-2.07341656t-c	4	11.490013	
	15/6.6460	LIBER	-15666.447	1003	6.21379576E-2	471	15.262449	
	43124.51¢	ED AR	J.D.B.D.B. 447	1001	-8.12123437E-5	ROJE.	0	
	-910.chers	_	-< 5.040625	¥ - 4	3.	PALLERG	8	
			-114.56263	***	657,65514	- t+t	2.6	
	-151.41956	1.2	449.67685	712	315.169	2004	0.0	
	510.00.019		-543.23299	7	5,94,6974	MAGU	0	
4 4 1	-33ct.ahs1	ĭ	11564.854	ĭ	-4751.7686	TAUD		
	-3657.4541	- 2	55.00.5000	a Z	- Abr 75.8/8	2400	•	
	-16.40556	H > H	-10.555973	ALFETI	4.67425	200	2.2	
-	サックシャ		-+11106114-	At 6 4 7 3	3.972105+	£400	9.6	

RUN 31.

						-																												A THE R PERSON NAMED IN									
6.6	0	2.0		8	~		53	45.827828	8-9514927	49.536758	· V	2.3118853	5	1 N	8									1 1	0			2,1729514	-0.53915172t-1	2.08998	202.34284	12.147740	15,725623	6.0	0.0	2.2	.50	80	2	5	€ €	3.0	
PSITRE	VXS1R.	VYSIR	V 251H.	PSTR	1 > 1 3	FSVT -	USTK.	XA		X	1	MIGE	T	× 00	T LOG	TITE	r_TI	JITE	X I I I	LHITR	CHILD	XIIX	4114	ZITH	LITE	RITE	X L HZ	AMP	AVP	42×	d×>	44.	474	- としのよ	PSIUME	818	MADO	X A D U	YADU	ZADU	NACO	LAUG	
120.6	9.5.	11117.0	8	1.00	-16.8	2.54.5	34.378068	-5.7652658	5.6	8.2436818	9.9448616	4.9536757	0	3,9343464	-6.6272144	2.41697418£=1		90268955-2	34698838	0.92182146	1.9683214	8.51943242	C. 63285611	a.5227326	24.666468	1115.8902	1565.8917	2.	P. 35966427E-2	-0:10252371E-1	P.16914587E-1	-2.1178932BE-1	6.565956926-2	-8.42784791E-4	2.8	1845.1923	-561.00197	6344.5400	-11804.711	-52455,114	-4.6716617	3.4184942	6.6273436
	() FT S	VSOURE	UELSMR	TESTER	7 F C 1 7 K	LHI	Svl	141	12	TINGER	TH757H	XCIR	STR.		A A 1 P		ADDL	ARIC		×				LTOT	עזמו	718	I I	RIPDLR	VKBUOT	Trangla	120021	. 1004	BOOL	Regt	×1×	4 T R	218	1.78	æ ₹	2 - 2	ALFHTT	ACP PT	AAbul?
367.26582	246.29999	W. 23783808E-2	SOUNEWROE.	5.4	5.6	658.8	45.5	-1.8006184	11.50/668	10,575861	<3.464861	W.89514927	8.92021140	5.8803558	2.2	5.2950735	7.9	2.1	5	1.4641259	1.6446784	W. 48645k75	0.67177479	E : 825347 35E=1	-4.64692014E-2	8.632F.8943E+6	6.97746653	0.715255738-5	-1617.6586	365.64255	19145.861	-<901.3836	-17865.492	50932.667	-41.617626	- 7345 SEVET	641,1/845	-515.73705	16164.1/8	9561.5356	-17.666576	-285:4138E	E. 21166242
FSCG	۵ ۲۰	DHa.	TIME	1.585	1.255	PASCNI	L L	A15	915	ThetAU	THEITH	X SI S	KGALTI	THETAB	7110	BETANF	545.6	DEGKAT	PSIUDI	ERTX	EKTZ	t PORT	FERT	-C1810-	CHSIG	CCHSIG				1	TOAR	LBARM	MIATI	DUAK	K.I	¥1	12	11	Ę	2	TVA	TV4	1 ^ 2
2*.45.661	5,0000	41557.5	30,000	21.013549	164.00000	15.0	760.4950	ひしいコワノスア・ロー	11.010/61	10.5/3001	15.996464	0.357782E	9.3651141	272.542A4	16.14//40	15.103063	7.	7.7	S- 9	1.00000	01.01(5)(3)	3.8	55.014547	B. 21 970 ET3	A.19/58/58E-1	8-46305050	-1.1755363c-c	P.1c5+5+59E-1	2615.2505	-	-1 4. 51.3FG	-6571.9436	• < < 1 1 . 1 . 7 . 5	24030.504	-16/4.1241	PRINCIPARS -	- = = 1 - 1 + = 1 >	163.41715		-4V3V. 1915	-51.63/634	#2: asserat	D46.4.740
ME I GM I	1.	1	71	DMF GF'F	OPECIT	A T	-tn	LAISTK	Lhbola	COLSIR	PEUAL	MAIN	KOACTF	VAU	C 1 >	47A			æ		<u>.</u>	1 1 1	u. 6	SKOK	BC.4.5	27nw	LAMBRA	DASHER	I I	472	× - 2	ار لا	χź	Z Z	F F W		1 = 2	1 × 1	4 2 5	7.5	r r	444	ZH1

HE1CH!	1. A.	FSCE	36r. Prest		140.0	PSITHE	2.2	
11	6460.5	3 L C 6	446.24945	UELS	5.5	YESTE.	80	
14	91707.8	DHA	2.237693664-2	ASOCINE	1117.6	VYSTK.	B. 60	
7.1	30460.4	1144	5 . A O S Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z Z	UEL SAK	9	. 4187V	6	
してたいいれ	**************************************	2000	7.	THOUSE	112.2	PSTR.	5	
9	104.101	1.553	5.4	TABLIR	-14.F	#L V T	275.0	
X W X	13.61	PASCAT	- e71.e	FURT	2.852	PSYT	2.249	
FOIL	162.4.5.10	Snl	45.6	SVI	34. 384666	CSTR.	0	
LATSTA	-1.54.0556	814	-2.5440389	FE	13.2418567	KX	41.570883	
LNGSIA	15.4/1514	F15	14.141699	12	-5.8	Z ×	6.65868329	
CULBIR	19.557514	THETAR	17.009514	TINET	9.6095188	N C	56.687465	
PEUAL	14.47.74	THELTE	25.117401	THISTR	12.217481	2.	41.611844	
- MISK	4.1575862	- MISK	F. 63F68929E#1	RCIN	1. 555 V 653	MENT	2.2467379	
ADACIF	-4-45654664	XMACTI	-E. 45c34009E-1	BSTH.	5	4181		
1 X X	256-1-713	THEIAS	3.4143666	1211	3.4144319	2007	9.9	
24.7	144: 70551	0110	2.	4416	-6.2611698	ましいま	0	
420	14.175138	BETANT	3.3123762	1100	-0.45826872	TITR	2.2	
a	2° 0	GAMC	2.0	AAKL	-1.0194735	1111	8.	
٠	4.7	CHESTER	1.8	1144	12696618:2	327K		
O x	S. C.	PSIUUT	5.0	EBIL	W.75561156	RITE	6.9	
ALFXF	1.461436/	CATX	1.5167144	FRETX	0.90576748	LHITE	2.0	
CHILF	56.443501	EKIL	1.6/801/2	Er. mr 2	1.0095561	GFITE	6.0	
1 1 1		EPSHT	d. #0512571	14516	0.50297751	X11x	0:0	
, ,	77.500.020	I ?	6.01177979	X C V 1	e. 65276788	YIIK	6.0	
BENDE	おこひとすかないかな	C1516	-c= 05113c05t-1-	£161	4:9195789	-2114	8.8	
AUTO	8 - 1 + 2 + 4 + 1 + 4 + 4 + 4 + 4 + 4 + 4 + 4 + 4	ETSHO	-n. 31531914E-2	010	24.674911	LITH	8	
5704	ロッドショイナル 14にっと	918473	M.65961429E-6	¥	1416.4879	3171	9.0	
LABEAT	-0.05 SCH5/t-c	7 N	E. 99645077	I	1988.1615	* - 1 2	9.0	
これのまな	8.1cc/5755t-1	ں د	7.14385114E-4	X I B B L K	. u	AXP	1.8798546	
ĭ	2645,7115	T C P	-1861.1726	YAPUUT	-C. Sed41757E-1	AVP	-0.55916692E-1	
* X.	Diaz	1984	404.557.09	100444	P.12122362E-1	474	-52,143636	
747	-1715-313	1985	17655.654	vzeuot	-2. <1 2440 E 2 E - 1	0 × >	256,16713	
, 1	-8467.12E1		-4466.6966	1004	-F. 43392359E-2	444	14,698651	
Ţ	-1350-051-	LOBER	-c1/e1.684	ancı	-0.13543825E-2	474	14.186158	
7 7	37275.203	***	50941.185	1001	8.2/648984E-2	またのは	4.3	
×	-11/1.0354	K.	-17.554164	KIR	50.5	PSIDMG	9.0	
484	-614.03767			#18	1551,1622	244	516	
1-1	-340,11561	17	967.91142	×17	-484.28438	MAUD	6.	
1.46	217.2.212		-697.71575	ا د ا	6131,6717	X A D U	20.00	
***	-5115.3761	Ē	21457.257	I I	-15211.568	YAUU	0.0	
* * *	1247.1745-	-	8195.85.8	1	-41645.844	2460	8.0	
I	-54.7: (351	XYT	-<5.546633	ALFHIT	-5.5385216	NAUG	0.0	
I >	-5.50.0040	- *	-515.19184	ALF VTI	5.5946457	FAUT	9.9	
147	267.12726	1.7	0.1c16349t	4 E B B 1 F	6.2/16698			

MUN 53.

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3.13619	012	2.23789688E-2	A SCONC	200	* 15	. N.
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41.01444	2000	2.0	THRIBER	6.00	*	
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13.8	PASCRT	2988.6	FLFT	234.8	PSVT	-
333.8° 30.	FES	3	SVT	2.5	USTR	,
-1.5651156	A15	-4.7524689	L	-5.5426965		100100
242421.11	618	14,565055	1.5	25.60	E M	
27.927618	THETAR	21,622616	TIVIER	12.742576	×L	54.641738
14.14105	THEITH	27.616347	TH75TH	15,716597	¥	46.576655
3.9267567	X18x	2.2	TUIN	5.4541732	MIGI	2.1912829
-2.55.3545	XUAL 11	-0.47503090	RSCH.	60	1 01	8
253.37319	THETAB	2,2652875	4224	4.8975846	GSTR	
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12,22,155,2	SETANP	3.2223534	2120	-8. S4473884	TITH	
	C.AML	2	7877	76888436	1	1 N
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7 3	PSICOL	2.0	8816	6,71456545	AT I P	· · · · · · · · · · · · · · · · · · ·
3,52996536	ERTX	1.5496155	PRHPH	2,84983681	LHITE	8
14. 154561	EAT2	1.6903649	Exnf Z	1.6163155	2 2 2	60
5.8	EPSKT	7.46228533	SIGNI	0.55616578	XITE	
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2.30762713		W. 85288835E=1	1017	3.2455288	MLIZ-	
K.cecobodak-1	CHSIG	-C.14659716E-2	1010	24.384662	LITH	9
-2.44746511E-e	CEMBIC	6.9381931RE-6	7.	1073,4868	HITH	63
	7~	0.94945174	I	2154.7250	#LTW	5
2.104r.51e3t-1	٨C	8.715255756-5	KTRBLK	2.0	AXP	8,73786393
6156.2564	1001	-1717,9692	VABUOT	-8.25181445E-1	AVP	-8.83964128E-1
+845,75551	JBAR	15551.349	-YPUNTY-		474	#31. #9#626
-19061.1r	TOAR	19779.502	100874	-E. 58879724E-2	4 × P	255,36319
-9607.1752	LBARH	-9840.1771	P001	-0.2360E551E-1	444	14.084941
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46735.469	COPK	362.20454	RUGI	-Z. 281951792-3	BUTH.	60
-1959.0100	-	-61.67.0546	AIF	2.3	PSIUME	8
-582.7# 16F		* 559.2845	YTR	1328,8988	BTR	2:2
-217.15166	1.7	1411.9615	512	- ひなむ。 なわなか	HADU	6
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-55.441364	1 1 1	-67.5196,38	ALFH11	-t.6334578	DOWN	. S
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2.5	PASCRT	1001.0	HLM7	250.7	PST		
13	ST	47.6	541	34.500003	4514		
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-15514.144	TOAR	15557.356	100074	-8.41171881E-2	7 × 2	67,458235	
-4625.1654	-	-11117.6111	1004	-6.326186835-3	3 * *		
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	1405	014.0.010	1004	* - 12/8/82/2/	31.53		
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KFR	15.7	PASCHT	1196.0	KLMT	234.0	FSVT	5	1
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LATSTA	-1.11.03609	A 1 S	-1.3046502	IHI	13,165515	4 ×	45.132297	
LNGSIA	5.7671454	312	4.2617621	15	-5.0	E CE	36.476/93	
CULSTA	15.255565	THE TAG	15.565426	TH7SHE	5.5753262	XC	31.986414	
PEUAL	12. 57.555	INCITH		TITUTE	7.9846870	P ×	39.025682	
XAI	4.5152296	XBIN	3.3973793	MCIN	219261.5	MIGK	2.1073322	
XFACIF	\$5.441030	ASACTI	3.5041890	E . T .	8.8	TOOL	9.6	
V K G	141.75644	THEIAB	2,9682963	4244	3.0666189	ESTR	8.8	
v ₹ 13	11.750/61	PHIU	9.0	A A 1 F	-4.1670881	T C T	3	
9 7 A	3.43,60,41	RETAMP	5.9465193	1100	8.56843584E-1	TIIR	60	
a.		ISAML	S	AABL	-3.4866591	1111	8	
	20 · E	DEGRAT	2.1	ERIC	6.16364891	JITH -	9.6	
¥	6.0	Palvol	5.0	8016	8.9435842NE-1	MILIE		
ALTA	-4.718716	FATE	1.3796324	ERNTX	_	LHITR	8	
CHIPP	7	CKT4	1.8685069	EKAF Z	1.0104294	STIFE	2	
£ K 7 h	5 .	EPSKT	U. 30940316	SIGHT	8.94662441	XITR	a	į
4.3	12.469410	KGHI	0.87117474	->3	2,84362914	Y 1 7 K	0.0	
S C X C X	2,13985765	_ 01810	J. 56511882E=1	LTOT	-2:3464282	ZITH		
MUTS	1-301700011.5	SISHO	-c.16/81210E-2	1010	25,144511	LITE	5	
477H	-0.16728856E-4	CUMSIG	P. 44259199E-6	7 7 10	671,61543	MITR	8.8	
LAPENA	- K . C . C 4000 5t - 1	72	.94855558	I	959.21262	MITH		
UNSHAR	2.20199750E-1	<u>د</u> د	N.71525573E-5	KIRBLE	9.1	AXP	1.6534888	-
T T	1140.4419	HOAR	3.8	VXBDOT	-4.19001090E-2	AYP	0.55388183E-3	
or E.►	-245.35483	2004	285.83483	TODBYA	P. 1 P. SI D. 9 G. E. Z	424	- 52; 126442	
ZMR	-13340.250	TOAK	15452.983	VZBUUT	4.20245234t-2	d×>	161.25684	
ار د د	-3910.850	LSARM	-1358,2255	Poot	-E. SIYZA682E-3	A A P	11,936227	
ĭ	16765.044	HAVE	-2945.Seg9	500	2.98061257E-4	474	5.2365291	
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1-2	71.55255	7.7	-66.324126	218	-224,12285	MADU		
ر ۲	546.21154	5	-176,52161	L1#	\$605.0870	XAUU	e. e	
Į Į	-5451.5045	Σ	-1914.6530	ĭ	-1117.5197	VADU		
Z	-2146.5671	-	2751,2659	œ <u>-</u> ≥	-19555.351	ZAUD	6.0	
_ r	-dc.*115ce	⊢ ×	-6.42co53c4c-1	ALFHTT	1.4761458	NACC		
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7	31000.2	TIME	0.207362886-1	DELSAR	6	3 7	, N	
445340	21.019444	2002	4.4	TRUINE	-13.6	PSTR.	(N)	
DEFECTA	124,42400	4555	2.6	TASTIE	2 616	ME WT		
AFE	13.6	PASCAT	2000	101	234.0	F 2 × 1	: 0	
PSHT	166.45753	LED	87.K	S V	32.32220	Sul Mar		
LAISTA	06100110.70	A 1.5	-1.4221863	I'	-4.3695575	•	45.065458	
LTBSTR	11.220700	510	11.196584	61		60 10		
CCLSTA	17.105335	1 mg 1 4 6	17.106506	TINE T	7.6665852			
PEDSE	529661-81	THETTE	22:871789	Wander a-	4.41.1299			
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7 2 2	27311.036	2 A D D	26405.258	1028	6.264131995-2	2574	0.00	
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u.	-456.51333	L 1	-431.85546	7 1 4	976,12239	er 100	- N	
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1	-6.570256	144	-654.58517	ALFVII	3.45456	- AD:		

HE IGHT	16050.8	FSCG	362.25.868	A	2.291	PSITES	8.8	
*	5667.0	# L. C. G.	251.10000	Otls	2.5	VXS18.	6	
14	39335.6	ar Or	8.25788886-2	VSCUND	2.1111	VYSTR.		1
71	31620.0	1111	. enovouse	DELSER	3	V. 2518.		
DHEGER	666612.15	11555	9	TASTAR	-19.8	PSTR.		
OMEGIK	164.563.30	25.05	5.6	THBITE	90.00	46.41	273.6	
KFR	15.2	PASCNT	524.0	- THT	234.6	PSVT	645.8	
FUHL	100.4000	SnT	9.04		32.500000	DSTR.	5	
LATSTK	20011946	A 1.5	11.959964		-4 6 64 20.84		26.81.49.20	
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,) e	19		AAGL	-6.5747669	111	9.0	
	N	DISTRIC	2:1	RAIL	6.32362818	TIL	9.0	
		PSICOL	8.5	9011	6.64611572	HHILE	2.0	
	20 TO CO CO CO	F	1.5556353	EFRFX	2.8428888	LHITH	60	
111111	14.555555	FAT	1.7461666	Frat.	1.6166466	41117	9.0	
x - x	2.0	L ROLL	8.48333487	314#7	E. 53664379	x17#	6.0	
4 50	10.035615		0.01117979	-	6.53266565	4114	8.3	
HUXS	C.32551803	C1815	2.69272393E-1	1017	1182812.	M117	2.8	
FUTS	3.1 +45 557 1E-1	CHSIG	-6.71386429E-2	1010	24.535637	LITE	8.0	
SZOW	-2.21395531E-2	COMSIG	E.51815596E-6	118	1555,9474	MITH	6.0	1
LAGBER	-6.1153c7c3t-1	71.	6.94621726	I	1173.4758	Z I T E	5.0	
RETO	6.915697751-2		8.145851146.4	KTRSLK	1.6	AXA	1.5063642	
T	2442.6931	101	-1656.2649	VEGUUT	8.49614984E-1	ATP	8.19577234E-1	
br Sc.	-319,58345	JBAR	319,58546	100811	8.81823265E+2	424	*32.894723	
7117	-13755.255	1044	15073.514	VZBUGI	3.22139913E-1	d X >	236,34101	
1 2	-7775.4561	LBARH	-4123.3264	1004	8.53233768£-2	444	10.124484	İ
Y	-1/2/5.446	13401	-<3407.059	2001	2.76495673E-2	424	12.769348	
117	\$3507.413	400	35142,292	*ppt	-0:14891885E-2	RSTR.		
A S M	-155.4505	, X	-11.312252	475		PSIDNG		
A. A.	-012:53063	F	18212:418.	***	1293.4733	*16	8:3	
Zmr		12	1124.1262	218	-455.45584	MADU		
144	1214.5827	5	-364.67786	L.7.	7167.1211	TADO		-
L	-5405.6364	ï	51654.937	a L	-14155.213	4400	0.	
LTZ	-3214.94F1		6767.5148	* 1.2	-35593.707	2400		1
T L	-53.14746	# · 1	-65.514369	ALFHTT	-6.2965649	2004		
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WEIGHT	16453.8	FSCG	363,23338	A	153.0	PSITRE	8.8
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ロアドロエル	.31399	2000	9.4	TERESTER	-18.6	PSTR.	9.
OMEGIA	124.52700	5888	N.5	TASITE	-16.4	R V 1	273.0
AFR.	15.3	PASCAT	3324.3	MEMT	234.0	FSAT	695.8
FSHI		SHT	20.00	Svf	34.300300	· LLDJ	
LATSTA	-1.4423172	A15	-2.4751981	IHI	E - 2	*	560086.00
LNBSTA		513	15.194999	ST	5.4	2	
COLSTR		THETAS	26.336920	THISAR	13.425923	. X	52.568254
PEUAL	7.1451510	THEITH	19.779341	77/77	0.2/30415	7	1
PIVE	2692962.4	YEIG		XCIN	6-2568254	4	3.7468111
XBACTP	-2.1.16010	XBACTI	-1.4101416	ESTX.	5	1150	5
VXB		THETAB	2.6743756	AASP	3,3631961	STR	
VYB	9	D I L		4415	8815851-84	3	•
97	11.526512	BETAME	-	8616	-Z-25667853	1114	•
		GAMC	50	AABL	-1.5713495	Y	6
	2.0	DISCHAL	3.1	ABIL	3615854	1112	8
œ	5	PSIDUT	2	H01L	3.85784505	1111	2 6
ALPER	1-1653978	ERTX	1.5121666	EKAPK	. 8633333	CHITR	6
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ERIK	5.0	EPSHT		SIGHT		XIIX	0:2
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3040		518HJ	-e. 61625965E-2	1010	23,435769	LITR	9.0
870H	E	COMBIG		-114	262	HILL	8:0
LAIGE	1000	72	9.99615561	YIL	2041.5954	MITH	60
DESTIN	0.67956198E-2		1-4612317	ストなびした	1.9	AXP	1.1687983
I I	•	I D A X	18	VXBUOT	U.15291701E-1	ATP	1.9482344
*	-545,271a1	- June	-343.27141	TOORIA	-0:14636425E-1	- d24	-31-471226
Œ	-16443.548	TOAR	16571.458	10087	0.20443894E-1	AXP	253,20654
¥	v	LUARH	-5959,0861	P004	-8.57497759K-1	ALA	0:0
III	-14637.864	LUBRI	-26185.517	1000	8.23936389	47A	11.826518
XXX	666.81128	28AR	41557.272	RUGT	202011420E-2	ROTR	
XXF	-1406.0141	. ×	-114.59514	X I X		PSIUME	8
AMA	2.6	-	108.39432	MAA	1513.4874	BTR	- 1
7112	-289.45535	17	1674,2927	218	-514.46836	HADU	5
F 14 4	2.0	- 1	-194,53481		6069.1757	XADD	60
T T	-7556.5058	r F	47549.750	α 1	-15939,944	TADU	
ARE	3.	11	2975.9843	Z - Z	3194.55	ZAUU	
THX	-81.527534	TAX	-31.975027	ALFHTI	-8.2958516	NADU	3.
LH.	6.3		-126.39sac	1	9.0	008	6.5
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EIGHT	16937.2	F3C6	300.2000		1.00-2	PSITRE	9.0
	5064.4	310	251.11400	UELS	9.5-	VXSTR.	5.6
	39946.0	RHO	D.237803672-2	ASDOND	1117.0	VTSTR.	9.0
	31265.1	114	8.200voubbe-1	ULLSHK	2.0	VESTR.	9.0
DREGAR	27.21 1477	NSSS		TRSIME	-16.6	PSTR.	6.6
OMEGIR	104.04.05	NS55	5.6	TASITE	-16.8	*4.7	273.0
1	15.0	PASCAT	965.1	REMI	234.0	PSVT	643.8
FSHT	100.40000	SHT	45.6	SvT	34.300000	USTR.	6.9
LATSTR	*P.13915#24	817	20000226.20	IHI	21.291733	*×	44.885311
LNUSTR	4.70043-11	618	2.1156757	13	9.5.	Đ×	35.279681
COLSTR	17.576478	THETAE	17.576974	TH73MK	7.5989743	×C	84.993546
PEUAL	125050	THEITH	51.093297	THISTH	17.595247	¥	19.348724
HIVE	0166326.0	KRIN	3.3879681	ACIN	0.000000	MIGK	1:2446839
XBALTP	54.910054	KBACTI	3.9488839	RUIK.	5.5	PSTR	9.9
	8.1062712CE+1	THETAB	5,2619956	AABP	3.2894545	DSTR	60
	8.9	PHIB	-4,7172532	4414	-4.4552748	RSTR	0
¥.	B. 15557 5552 *2	BETANF	6.3	9011	-1.0836428	T111	6.0
	9.0	GAAL	6.3	AABL	-5.6143462	HITE	9.0
1	2.2	DAGRAT	1:0	ARIC	8. 38883895 E	3114	8:0
	2.2	PSIDOT	9.9	HOIL	0.1424419	HHITE	8.8
AFER	-9.0000160	ERTX	-8.6446694	ERMPX	8.74945393E-5	LHITE	0.0
CHITPP	2.463675	ERTZ.	6.200527¢	EKHP Z	0.11899350E-3	0H116	8.0
ERTH		EPSHT	6.4466640.0	SIGNT	9.9	XIT.	9.6
	.350055	-	0.87177979		6,84852813	4114	8.6
ĺ	*** とうかっからかかしゅ	61818	#:65739c76E-1	100	-11:58908b	2114	8:0
	٥.0	CHSIG	-6.47175314E-2	1010	24.784324	LITE	0.0
	6.4c6: 51cet-6	Cumbig	#.52038753E-6	114	1454.0428	HITH	0.0
LAMBAR		7,	19695946.8	HPH	1535.4383	ALIA	9.0
DROHME	6.3554704E-1	·	0.11055025E-3	RIFFLE	9,7959999	AXP	2.9737894
	1441.1441	T T	-644,43386	VXBDOT	W.14415914E-1	AYP	1.5520523
İ	-633, 44700	4000	253.94740	1000+1	0: 54055065E-1	474	-31,997351
	-15141.347	TBAR	15255.712	VZOUUT	-N.17886564E-3	d×>	8.16827128E-1
ì	-3645.0180	LOAKH	-4705.2632	Pucit	-8.24936917E-3	YYP	6.0
	13406.504	HABE	-6254.2357	1000	0.34140/58E-4	474	W-15557353E-2
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5 2		۰	6	50 00	3	2.00	34.500000	20.628237	9.5	5.9752877	8.8124624	3.4845548	9.6	3,1774036	-2.1669967	-2.37848587E=1	-3,4847695	0.17702205	0.939036796-1	6.74962454	1.0191791	6.8	0.64852415		344 14848	1294.4951	S	0.84385659E-2	0.28990786E=1	-0.20250166E-2	2.63636565E=3	0.63204447E-3	0:16542886E=2	0.0	107.19965	-257.61849	4040.6161	-1981.8797	-21924:491	200220000
	8 1417	ANDONA	DEL SHK	MENON	THSLIK	MI MT	. L > S	INI	15	TA7UAK	74/574	MEIN	RSTH.	AABP	ALIF	5516	AABL	AAIL	BBIL	REAT X	EKAP Z	LEGIO.	- ×	1010		I	RIRDLR	VXBUUT	-Yebbot	10007	Putt	DUCT	RUUT	X	YTE	214	L 1 K	Œ	N T E	ALFHIT
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13.0	PASUNT	1547.0	MUNI	2.86.5	PSVI	
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-		41567.8	DIA	9.175PAE2PE-2	VSUUNU	1077.8	VYSTR		
7		20700	7.145	6 - 200000000000000000000000000000000000	UELSMR	2	V7818	0 6	
ž	MEGME	21.619919	SSHL	0.0	TRUCKE	8	PSTR	6	
Ĩ	HE 61F	164.000.4	SSAZ	2.0	TESTA	-10.0	111	213.0	
1		15.5	PASCHT	1135.8	- NERT	234.0	PSVT		
n	~ 1.0	140.4.4.0	Pr.S	42.6	S v T	32. 505568	USTA		
4	ATSTR	42442416.8-	A15	-1.0240701	Int	11.251416	d X	44.2536.58	
2	NESTA	4.CASSED	918	3.6427668	15	2.5	D	35.672554	
2	ULSTA	17.100,500	THETAB	19.125528	TENULE.	9.8465286	ŭ	54.25.2886	
4	EUAL	10.476215	THETTH	26.040.02	TETUTE	13,546262	¥	35.944813	
*	MIN	4.4245436	MEIN	3.5872554	MEJM	5.4252886	MIGE	1.9489262	
4	BACTE	51.0kx306	XDALT1	3.7000000	PETE	0	0 - S G	N .	
×	X	155.195#5	THETAB	-2.11521816	AACP	4.1964279	DSTR	8	
-	D	11.005001	7110	.0	A & 1 F	1.1215078	1201	5	
2	20	-6.671Fab15	RETABE	4.4136718	5615	8.26238415	TITR	6	
		× • • • • • • • • • • • • • • • • • • •	6446	2.2	124	-4.8593529	*11"		
1		2.2	DMERAT	1.3	ANIC	2,17135486	JITH	65	
		2.0	F\$1001	2.0	8616	-R.10569161	RHITE	8	
_	L+ ×F	-1.0419418	ERTX	1.2437349	EKHPX	2.95467785	LHITE	69	
1	77.1	66.357055	EXTL	1.9512148	ERMP Z	1.3895481	Call	50	
*	¥	2.2	FPSKI	B. 4941367E	SIGHT	B. 77658612	KITE	6	
4	*	47.77.10	-	2.87177979	YO X	8.54746034	AIIA	3	
3	DXS -	2210220122	CT816	42892211.B	1101	*6:3126864	2178		
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•	570	-7.1215#523E+1	ひてのとうひ	8.915972126-6	118	948:85849	1111		
•	AIGHT	-E. 5551 7406E-1	7:4	1.2566774	ITI	1324.2941	3-12	3	
*	RETEN	8.236451968-1	, AC	U.44773485E=0	RTHBLR	6)	AXA	-6.53651421E-1	
2.	Œ.	416.401/0	I do	591.03517	VKGUOT	8.10556180E-1	AV	8.29664851£-2	
5	K	-153. 43575	JUAN	-523. # 38£#	-10001	-C.1778789Pt=P	47#	-32.175665	
*	ĭ	-19169, 355	TOAK	17100.398	V28001	-N. 55439591£-2	0 × >	135,19585	
5	3	-5515.21ez	LBAKE	-975.72561	P001	E.126018c1f-2	4 + >	11.683827	
3.	¥	107.50.00	X D A C.	3e57.1462	2070	-0.12116565E-3	474	-0.27184676	
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•	<u>.</u>	-410.1160	_	33.181445	a l x	8.9	PSIUME		
•	_	+238,9386B		-185.51e19	414	858,63838	878	2.5	
•	<u>+</u>	138.104.7	7.7	den. Se944	812	-314.7852	MADU		
•	*	466.35155	L.1	-225.49391		5261,8167	MADO		
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ŗ	710.000mg	SriT	47.6	2 v T	34.384460	LSTR.	S	
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מ	12.2316-1	FrIB	3.9	AA1F	-6.68216248	X 10 X	9	
æ	2.8721538	DETANF	3.6164485	4149	6.24742626	TITE	2.0	
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	, , o	PSIUUT	2	9616	-E.71892743E-1	MHITE	5	
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1777	84.1443/4	EKT 2	1.6661750	ERMF 2	1.0078116	STIND	9	
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•	24.3173ER	13	4.67117979	KGVI	6.65053271	YITR	9	
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-	-531 -3414	400	531,536194	-**FUB !-	0151458935E-3	424	52,149913	
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~	-5164.5154	LOARD	-1463.6665	PLOT	-8.27118501E-4	d A A	14,237827	
~	2440.6301	HHER	-1657.8695	GUGT	-K.17465638E-3	424	2.6921558	
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		*	-134,51956	# # #	966,46366	818		
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٠.	-6540.0011	ī	15177.425	ĭ	-11285,783	YADU	9.0	
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# HOR		128.8	5	3077.8		-16.6	3.0	234.6	32.588888	1.000400		10.755939	16.872256	6:4712119		4,1757697	-8.64485518	8.24656512	-6-9727763	8.33964228	-8-12796733	8.93457266	1.0073405	0.61653264	0.63202816	1.8475987	24,273559	1329,8872	1673.6374		-0.15454646E-1	2426466429		9-144135785-2	-8.53617964E-3	9.0	1249.7701	-454,88293	7634,5286	-14594,168	-46097.048	-6.4889887	3.4048726	8.69856341
		A	DELS	VSOUND	DELSMR	TESTE	THSTTR	WENT	1	IHI	1.5	TEVUER	THISTR	TCIN	RSTR	AABF	AAIF	6617	AABL	RAIL	8611	PREFX	EXMF 2	SIGHT	KOVI	LTOT	prot	TIR	E L	ストのローズ	VXBDOT	1000		1000	Root	XTR	APR	21R	112	M T	Z L	ALFHTT	ALPVTA	AABB1F
38-AUC-17		347.9	246.29999	0.17586080E-2	6.28849898E-1	4.0	2.0	1426.0		-2,5186297	0.2156732	20,633939	30,372256	1.9204075	1.6199871	-8.17211994	0.0	3.4676661		1:0-		1,1934919	1,7967266	8.46467600	0.67177979	•	o-	0:011449672=6		0.13411045245	166.47393	19696 200	77.7.5	-1467.5361	36142,510	35,104295	-185:1955	669.78173	-144.18583	19790,698	5313,7578	-	1905	1,5331625
1-51-17		F 3C6	907	E HO	TIME	2000	ROSE	PASCNT	PH.	A19	618	THETAB	THETTR	- NIGH-	XBACTI	THETAB	PHIA	BETANF	GAMC	OMGRAT	P81001	EKTX	EKTZ	EPSHT	KOHT	_CT818_	CIBIO	COMBIG	2		H O C		BABH	HOVE	00 A R	×	- **	11	-	F	*	¥×1	444	1 1 2
UTTAS(876) 1		19900.0	6268.0	41567.8	36224.8	27,819999	2	15.8	89.400	1,45157	0.7152465	8,83393	17,947529	.092760	4.14987	82.7987	13317	0.00915	•	0.0	3	3.	6594	•••	40.563635	,2792826	.1797811		-0.32771971E-1	0,17293321241	020.00	10000000000000000000000000000000000000	-7869-8628	3217,4670	37649.441	-957,68977	- 200,44471	134,18066	644,78863	-1362,0562	-2061,5333	46.697696	8:333	. 246
		MEIGHT	*	14	7	HEGHR	OMEGTR	KFR	FSHT	LATSTR	LN681X	COLSTK	PEUAL	MYY	XBACTP	O X A	0 × ×	82A	•	٥	•	ALPHP	CHITPP	EX TR	DEF	HUXS	MC 4 S	30X0			E 1		T .	e I	W I	XHF	ANA	342		E E	1	×	- I	7 M J

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1,986,1	6266.0	41987.0	30224.0	27,619999	124,62000	15.1	100.40000	-2.7937671	13,317537	23,401544	29,436888	-	-0,27401334	236,43833	_	-6.4364413		1 9	•	*** 5966884	e.		3,667619		2017	14277	19901	1795	-1120,0393	2		•	56667.744		271.30555			17.50	
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									1																		-		E-1				1									
9	8.0	8	5	60	73	695.0		35.289831	1.8145745	74.877839	20.757316	1.3568682	8.0	•	5 5	5		6	6	6	9.6		6.0			0.	8	8.59273494	8,33010ce3E	-32,199134	256.53886	16.465121	4,5634664		0.0	2.0		•			6	
2 H 1 S d	VXS1R.	VYSTR.	VZSTR.	PSTH	HLVT	FSVT	COTE	× ×		U	Q.	MPIN	PISTR	S 18	RSTA	TITE	HITE	JITE	MILLIA	LHITR	GHITA	X L I X	-	ZITR	11	MITE	NITE	AXP	AYA	AZP	Q × >	440	- d2A	ROTE.	PSIUME	818	MADO	XADD	YADU	ZADU	NADO	LADU
146.6	9.4	1677.8	0.0	-18.8	-16.8	234.6	32.360000	0.22549525	8-5-	12.588454	21.382665	7.4877839	6	4.23.0485.2	-3.7852726	-8.41644849E-1			0.15021995	8.92386154	1.8682268		0.85851793	0.04100982	24,344582	1780.1776	2498,8873	1.0	-8.39691521E-1	0.16585572E-1	-0.25867362E-1	0.77547262E-2	-0.48355222E-2	-8.96696682E-3	6.0	1672.9441	-688.98586	16219,515	-18865,846	-51833,386	-3.6121586	3.9848995
-	DELS	A SOUND	DELSMK	TENTEL	THOSE	I	100	InT	13	THYSHR	THINTE	XCIN	PSTR.	4884	PAIR	8616	AABL	AA1L	8816	EKHTX	EKMP2 .	SIGHT	KOVT	LTOT	DIOI	718	HOI	KIRBLK	VXBUOT	vrBuut	VZBUOT	Puor	1000	RUOT		YTR	218	LIR	œ E	Cr Z	ALFMIT	ALFVIT
22200000	246.29999	0.17500000E-2	Ē.	2.0	5°.0	1511.0	42.0	-5.5876137	12.694658	22.468454	34.862885	0.10145745	-0.16057374E-1	1.1100015		. 621	9.0	e	3.0	1.3518751	1,7159271	0.46621515	0.67177779	0.11162811	-8.36940166E-2	U.85411682E-6	1.4605863	d.71525573E-5	-636.77676	867.81197	19861.268	-2641,5031	-16155.432	54849.299	-14.677223	-264.67474	525.46245	-574.77540	14969,965	7312,9838	-14,785688	-259,50767
	ra				5000	PASCAT	SHT	A15			THEITE		XBACTI	THETAB	PHID	BETAMF	CAMC	DMGHAT	PSIUUT	EKTX	-ERT2	EPSMT	FEH	61818	918HD	CLHSIG	74	۷,	HUAR	JUAR	TOAK	LBARH	I	AR		1	12	_	ī	-	X V T	T A A
9.004.1	6258.0	41567.6	30664.0	67.614449	124.66000	15.4	788.42888	-2.3556269	15.650475	24.466454	35.282.85	5.5289831	-0.16051575	236,53606	16,465121	4.3034604	5	9,	2.	-1,4063510	61.175965	3.	53,548646	0.34613079	0.22769427E-1	-0.167616991-1	-8.25529c01t-1-	8.14507502E-1		3	112002.200	4	11495.696	49266.254	-1264,5118	9	-15.664666	817.10445	7	-4/48.4320	0.1076533	-2.5636646
_	×			~	~								<u>a</u>							ALFEF							LAMBAR	DASHMR	TIN	X X	ZHR	715	r I	Y I	XBF	4 2 4	442	L 1	I a f	ZEL	XH1	- II

	U11AS(S/6) 1	-21-17	16-5EP-77		RUN 34.			
HE I SHT	19968.6	FSCL	3	>	9.951	PSITES	5	
×	6,000.0	4106	246.29999	DELS	3.0	X L SX A		
~	41567.6	014	0.175##BBBE-2	VSDUND	1677.8	VYSTR	6	
21	36629.9	TIME	6. 20000000E-1	DEL SMR	8.8	WZSTR.		1
HEGER	21.019929	NESS	0.0	TEN TE	-10.0	PSTR.	5	
HEGTR	144.64505	2000	5.6	TASTIE	0.00	LVT	273.0	
KFK	15.6	PASCNT	1666.0	IL I	234.0	FSVT	10	
FSHT	766.46566	SHT	2.04	SVI	32. 5888888	DSTR	6	
LATSTR	~	418	-4-9166241	THI	2.1586894	•		
NEGATE	1	613	11.681756	- 81	614	4		i
COLSTK		The TAK	24 A255 K	127127	14 7455.61			
	22 20 20 20 20 20 20 20 20 20 20 20 20 2	416110) (6/6460.40	
1			•	E	A4.6	A.	9.	
274	A. 544. A64	ZIOX		ZHUX	8.9659512	ĭ	9.0	
BACTP	- Casana	XBACTI	-0.4246414	Z LOZ	9.0	3 L O L	8.0	
V X D	255,15674	THETAB	-2,9746592	AAOT	4.1587468	S - 80	. 7	
40	12.924649	PHID	100	4144	9.474ABF 44.9	25.40		1
7.1	-14.15567	AP TAME	S S S S S S S S S S S S S S S S S S S		7		•	
3	0055		6.6636367	1100	2	*		
	9 1	PAS	F .	AAGL	2.92007	I		
	20.	DMBMAI	7.0	A A 1 L	0.63384445	JITE	0.0	
	2.	PSIDOT	2.	33	-6.29419435	MHITE		
ALFNF	-5.6664699	ERTX	1.1617869	ERNFX	6.92365205	LHITR		
ddl TH	81.019596	ENT	1.8439836	ERMP 2	1.9661686	DHITE	•	
KIR	9	EPSHT	478436	SIGHT	41070700	WITE.	6	
4	25.00.00	KUMT	•	2	2104010		•	
7	200000000000000000000000000000000000000		•			H - 7 - 1	•	
	0 1110100	0 1	00/40411		•	HLT7		
2	0.1/6664136-1	1010	. 43/86283E	1010	4.16694	LITE	o. 0	
8704	-v. 56395857t-1	STREET	•	₹	2626.65624	MITR	0.0	
ANDHE	-0.30367734E-1	24	#.99582696	HEAL	3411.7000	NITE .		
CABRER	8.13472647E-1	2	-:	KIRBLK	1.0	AXP	-2.8793936	
X	200.21903	HORK	747.5/467	VXBUOT	-B-15755469	AYP	-0.7459755a	
¥	-1232,1346	JUAR	1632,1542	VYBUOT	-0.27855698t-1	AZA	-30.756137	
**	-19536.136	TBAR	19619.440	VZBUOT	0-11415029	•	C 1 M 1 . E 5 C	
721	-12509.565	LUARH	-1643.5708	Puor	6	. 0.	12 924449	
- XE	31362.931	HUBER	828.500	- 1000	9.5.5.5x8x32		4	
**	61266.162	HANG	-4-4974	1000	2001486	2		
4	534 5 97 5 9 1	:					9	
	065000000000000000000000000000000000000	- 1	00.03736	¥ .	2	PSICHE		
K :	144/ • 60363	- ;	-245.62438	¥ 1 €	1905.9157	91K	8.5	
A (3/6.54645	17	956.12723	218	-693.69795	HADU	9.0	
	664.41484	-	-531,53638	LTR	11642.668	X A D U		
L	-15729.556	-	27243.413	- 111	-21493:674	- YAUD		
	-4610.5769	Į	6781.6784	¥ _ Z	-59051,623	ZADO		
×II	81.134581	- ×	-21.646646	ALFHTT	-6,1673489	NADO		
F	-4.8894504	1 4 7	-248.13485	ALFVTT	2.8984779	1 ADD		
ī	954,95734	ZVI	3.1592999	AABBIE	9 445 4V.S.C.			
	0	000011			١.			

TE TOTA	199661	FSCG	366.2000	>	150.0	PSITRE	0.0	
< ;	9.0000	2	640.8444	DELS	8.01	VXGTR.	5.	
- :	2.70CI	RHO	4.17598888E-2	ASOCNO	1077.6	VYSTR.	8.8	
77	30554.6	TIME	W. AUBUNDONE-1	DELSAR	9.0	VZSTR.	6	
OMEGME	27.014999	NON	9.0	THOUTH	-10.0	PSTH	2	
OMEGIR	124.62000	NOON	5.6	THOITE	200	N V	273.8	
XFX	15.6	PASCAT	1557.0	FLHI	254.0	FSVI	695	
FSHT	100.4000	THO	45.6	SVT	32.300000	DSTX.	5	
LATSTA	-5.4386555	A1S	-4.8659008	H	1.3546781	×	26.508482	
LNGSTK	16.6	619	15.510009	13	-3.6	D		
COLSTK	24,536749	THETAB	24,336749	THISHR	14,256749	X	86.684684	
PEUAL	24.4368BF	THEITR	37.5	TH75TR	24.0	ď	9	
ZTVX	7.00000CD.	XIOX	9.0	XCIN	8.6504684	MUN	5	
XBACTP	-2.4578545	XBACTI	-8.29378544	RSTR.	5	PSTR	3	
A X C	255.46965	THETAB	0.44956497	AABP	4.0285314	OSTR	9	
- 844 -	10.549719	PHIB	9.0	- 414A	4.1308915	FSTH	60	-1
474	1.9888267	BETAMF	3,6053156	8611	-0.14074197	TITE	5	
Q	9.9	GAMC	9.9	A A OL	-12.037575	HITK	9	
	\$	UMGRAT	9.	AA1L	8.75648394	2118	6	
œ	9.0	PSIUUT	9.0	9816	0.24360257	MHITE	5	
ALFAF	-1.1268325	ERTX	1.4665106	EKMFX	0.98654886	LHITE	3	
CHITPP	60.503760	ERTZ	1.7167620	EKAP 2	1,0094962	OHITE	0.0	- 1
EKTR	\$.	EPSET	0.46605315	SIGNI	0.63453555	XIIX	0	
i i	66.531652	T T T	0.87117979	KOV.	0.63155861	YITH	5	
RUXO	0.54927631	CT316	0,11127502	LT01	0.36992740	ZITR	8	
MUYS	0.22328666E-1	CHSIG	-0.39601074E-2	1010	24,501063	LITE	6	
870H	-0.15536783t-1	CUMSIG	4.87932465E-6	118	2040.2778	HITE	0	
LAMBMR	-0.29113717E-1-	7N	1.0047198	HPHH	3201.4266	NITH		1
DESTRE	0.15555014E-1	۷.	0.89465967E-6	KTRBLK	9	AXP	0.72012783	
TEX	1669.6552	HUAR	-676.21544	VXBUOT	0.99003665E-1	AVP	-8.34380698	
X N X	-1051.7110	JUAR	1031.7110	VYBUOT	0.56826017E-1	4ZP	-32,924968	
X 1 2	-10458.619	THAR	19608,915	VZBUOT	-0.15570784	4×V	253,46965	
LER	-12595.565	LBARH	-2631.2354	PUOT	0.11854068E-1	ΥΥP	16.549719	
TI	4519,6291	MONH	-12775.825	0001	-0.24523881	d7A	1.9888267	
¥	63220.461	CHAN	65165,975	RUUT	0.17076685	RSTH	0	
X	-1457.8668	×	-43.279671	XTR	9	PSIUMS	6	
AME	-571.66156	¥ 1	-286,62103	YTR	1924.8947	BIR	8	
2×5	21.061179	7.7	439,46363	2TR	-700.60574	MADO	6	
J. H. J.	955.04934		-624.97055	7.7	11756.605	XADD	6	
	-9421.8754	HT	12493,775	H TH	-21707.101	YADD		
7 X Z	-5076.8566	7	7997.8613	2 2	-59639,654	ZADD	6	
r x	-4.4906564	⊢ ∧×	-18 789621	ALFHTT	-2,7862592	NACO	3	
THT	-2.56407.79	YVI	-284.23295	ALFVIT	3,7295165	LADU	0	
ZHI	458.13874	1 1 2	1.3248900	AABBIF	4.7329846		•	

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	UTTA9(976)	1-51-17	9-DEC-77		80N 15		
HEIGHT	19908.0	FSCG	369.28389	>	5	041102	6
I.	6268.8	8 LCG	245.89999	2130		- XXX	. 6
I	41507.0	OI	7.17Seesee.	VSOUND	977.9	- A - A - A	6
17		TIME	S. Zenesesene	DEL 3MR		V7.4.TB	6
OTEGER	29.720700	5002	8	CE LONG L	6	010	i 6
DYESTR		2000	6.00	THSTTR	6	1	6
M L		PASCNI	1768.8	I	6	- > > 4	•
1184	707.40700	FIS	45.6		505050	97.50	
LATSTK	-1.0944239	A 1.9	-1.7097.44	-11	20 001123		
LNGSTA	0.00000		4 404044		14 B	E 0	40.10000
CGI STR	200100	THETAD		07867	7 367690	D (17.514205
PEDAI		116110			2000	ر × د ×	46,04,104
***				_	9.5	1	
2 4 4 7	4.0166338	NIOX	1.7319665	Z L	4.2746754	X D I X	6
XOACT	21.993865	X B & C T I	2.1993465	BOTR.	6.0	PSTR	6
9 1 0	101,23162	THETAB	3.2297766	AABF	3,3547941	DSTR	6
644	17,356673	BHIB	6.6	AAIF	-1.8636352	RSTR	6
87A	5,7128564	BETAWF	8,0633511	BBIF	-0.24623458E-1	TITE	
•	6	GAMC	6	1644	-4.1541244	- I	•
0	5	OMCRAT	6	4411	ALT-040- 2	1	. 6
Œ	6	PATOUT			A A D A D A A A D A A A D A A A D A A A D A A A D A A A D A A A D A A A D A A A D A A D A A D	111	•
ALFAF	68.5879999	FETT	1 2621922	> L	1		
4 4 4 8 0	76 36 36 44	C 10	22172020		96/6//400	F ()	•
1 1 1 1 1	13.678564	ER T.	1.4466122	EKHEZ	1.0147493	DHI TR	6.
1 L		# C C C C C C C C C C C C C C C C C C C	7. 15463558	LMSIS	1.2598555	KITR	6.
A. E	13.415961	× 0 ×	0.47117979	K 0 × T	0.75479755	YITR	8.
۳ :	9.12714544	CT316	8.9148A253E-1	LTOT	-8.8799858	Z17R	6.0
10 × 01	0.21769438E-1	CHSIG	-N.28799346E-2	0101	27,471151	LITA	6
NO78	6.51P01512E-3	COMBIG	9.68525924E-6	118	893.24377	HITR	8
LAMBHR	-8.29085651E-1	24	0.99833767	I O I	1512,7996	RITE	6
ax Loro	0.295956565-1	ر د	9.35762786E-5	KTPBLK	1,0	AXA	1.7989645
e L	1417,3374	HOAR	-429,68821	VXBDOT	P. 16676607E-2	AYA	-8.92339456E-1
0 1 >	-453.05495	JB 4 R	453,40485	VYRDOT	9.54645530E-3	424	-32,128793
272	-18833.958	78 A P	18463,792	V2800T	0.12188816E-3	0 × >	101.23162
ب د د	-5689,2498	LBARM	-1421,4396	PDOT	8.33823854E-3	4	17.358673
Z I	16341,307	HOARH	-6238,9834	1000	-0.16798942E-3	424	5.7128568
e i Z	27131,914	0848	27995,956	1004	0.37654558E-1	PSTR	6
M A M	-311,73696	T X	-29,194732	a L x	6	PSIONG	58.0
407	-287.34946	* 1	-97,959674	× 1.8	839.43699	BIR	6
J=2	156,58572	7.	-73,265627	270	-305.53867	MADD	5
<u>د</u> د	584,2644	-1	-215,61415	LTR	5155.8588	XADD	6
MAR	-4759.8981	Y .	-2122.9414	a L	-9466 35AS	YADD	6
F . X	-2418,6657	Z	2733.8349	2	-26888.545	7400	. 6
LIX	-34,188615	X V T	4.9918423	ALFHTT	1.7730267	NADO	6
414	-1.7261026	4 \ 1	-96.232891	ALFVTT	00000	400	
1+2	-71.5452AB	TAZ	-1.7483468	AABBIF	1.8637979		
				•			

	UTTAS(S76)	1-21-17	9-0EC-77		RUN 16.		
					1		
HE 16HT	19968.8	FSC6	369.28888	>	80	PSTTRA	6
×	6268.8	ALCG.	245,89999	DELS	0	VXSTR	6
7	41567.0	OI	P. 17599000E-2	VSOUND	1077.0	VYSTR	6
11	36224.0	TIME	P. 20000000E-1	DEL 3MR	5.	VZSTP	6
OMEGME	29.720000	SOUZ	6.4	TENTER	5.0	315	6
OMEGTA	137.07999	0007	8	THSTTR	6.00	WL VT	273.9
KFR	15.0	PASCNT	2031.6	KLH7	2 4 2	FSVT	695.8
FSHT	786.4668	PHE	45.0	SVT	32.396060	DSTR.	5
LATSTA	-0.86986941	A18	-1.5669727	THI	8,7391549		44.553316
LNBSTA	16.74941	818	6,0416779	13	8	×	11.672997
COLSTK	17,239255	THETAP	17,239255	THISHR	7,1592553	Ų,	42.245346
PEDAL	29,436060	THETTR	37.5	THYSTR	0.40	d ×	6
MAX	4.4563314	XBIX	1,1072997	XC12	4.2245346	MPIN	6
XBACTP	15,139236	XBACTI	1,5139236	BSTR.	5	PSTR	6
Vx6	134,95994	THETAB	3,3575525	AABF	3,3103565	OSTR	6
478	17.445405	PILE	5	AAIF	-2.8342951	RSTR	6
87A	7,9182832	BETANF	6.5668983	9915	-0.87259753E-2	TITR	. 60
•	6.0	GAMC	6	1644	-4.1746602	HITE	6
0		OMGRAT	6	AAIL	22458936	JITE	6
•		PSIDOT	6	6816	0.16237964	FILE	6
ALFAF	-3.9388427	EKTK	1.4147717	EXZEX	9.89859596	LHITP	. 6
CHITPP	79.664179	EKT2	1.7989198	EKWF2	1,2171358	AL IIIO	8
EKTR	6.0	EPSHT	9.51566898	SIGHT	1.0742954	XITR	0.0
4.0	20,343912	KOT	0.07177979	KOVT	P.78967819	YITR	6
S X O X	0,16952722	CT916	F. 91527454E-1	LTOT	-1.4132319	211R	8
MUYS	0.21878398E-1	CHSIG	-8.29416346E-2	ntor	25.520138	LITR	6.
820M	8.18561784E-2	COHOIC	8.59724199E-6	77.2	963,09224	MITR	6.6
LAMBAR	-8.21798429E-1	7 Z	8.99811819	a m	1588,5396	N I A R	6.0
DEGREE	0,227665986-1	<u>د</u> د	6.6	KTRBLK	1.0	AXP	1.9338594
E X	1597,4458	HDAR	-607.79460	VXBDOT	8.104962895-1	AYP	P.43461586E-1
Z ×	-435,64047	7878	435,84847	VYBDOT	-0.24149638E-2	AZA	-32,115199
212	-18869.859	TBAR	18929,074	VZBCOT	0.34192703E-2	a×>	134.95994
ر د د	-5944,7889	LBARH	-1865,9883	Poot	-0.18927268E-2	d >	17,445485
a 1	12125,547	HO ARI	-9432.6397	1000	-0.51885313E-3	424	7.91A2832
£ I Z	27859.681	GBAR	27917,121	PDOT	-0.19284157E-1	PSTR.	6.6
X	-479,96389	- ×	6.5568422	MIN	6.0	PSIONG	-150.0
¥ = F	-352,03201	¥T	-137.43266	414	923,67322	B T R	6
2=2	61,935217	7.	92.187388	270	-336,26398	MADD	6.6
	669.14684	<u>_</u>	-385.17834	170	5574.4606	XADD	6
A.Z.	-4341,5589	F	2612,9741	a L	-10416.548	VADD	6
L Z	-3864,5163	<u>-</u>	3835.6988	O L N	-25624.671	ZADD	6.0
I	5,1201052	- ×	1.5367369	ALFHTT	-1,5274318	DOV	6.6
L L	-1.5970551	* * *	-135,03561	ALFVTT	7.9560775	LADD	6.0
1 4 2	92,385647	2 / 1	-0.27834549	AABBIE	2,8343985		

	UTTAS(576)	1-51-17	9-0EC-77	•	RUK 17.		
						1	
MEIGHT	19968.	1306	369.20000	>	6.00	PSTIRE	6
Ix	6266.8	PLC6	245.89999	DELS	65	VXSTR.	6
14	41507.9	CHE	9.2370000E-2	VSOUND	1117.0	VYSTR	5
71	36224.0	4 I ME	P. 20000000E-1	DELSMP	6	VZSTR.	6
DHEGHR	29.726000	880N		SEL SEL	6.0	PSTR	
OMEGTR	137.67999	NON	F. W	STLOST.	-13.9	E < 7	6.27
KFR	15.0	PASCNT	1873.6	E I	200	FSVT	600
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LATSTR	-8.67282837	A18	-1.3969302	THI	13.111709	×	44 SAAR72
LNGSTR	6.4713916	818	4.4065899	8	- F	E	20 20 20
COLSTK	15,510192	THETAB	15.516192	THISHR	5.4301929	, u	31.449794
PEDAL	24,697869	THETTR	30.933320	THISTR	17.433320	Q.	12.582919
NIVX	4,4544672	X B I N	2.8965753	KCIN	3.1436795	X L d X	67945999
XBACTP	24.902105	XBACTI	2.4902185	RSTR	6	PSTR	6
AxB	101,19271	THETAB	3.6518479	AABF	3.1166914	G 10	6
V 7 B	15,212541	PHIB	6	ALIF	-2.5952798	4150	6
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713	-455.56431	JBAR	455,56431	VYBDOT	-0.69118836E-2	AZA	-32.079715	
278	-18969.822	TBAR	19032.806	VZBOOT	-0.93508340E-2	Q×>	67.485900	
œ	-5595,1043	LBARH	-1346.8160	PUOT	8.38914651E-3	4 >	5	
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.472930	KOHT	4	KOVT	9.8313P6F3	VITE	•
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16466954E-1	CHSIG	-P. 51501514E	1010		1178	
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19,9587	3648	319,95871	VYBDOT	1598924	4 Z P	
15512	TBAR	15574,169	VZBOOT	3.138338	4×P	64.69628
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Œ	6.	PSIDOT	6.0	991	0.61676479E-1	MHITP	6
ALFAF	-7.5182198	FKTX	1.2873449	RESER	8.86825383	LHITA	
CHIPP	77.113986	FRTZ	1.9163016	2 de x 3	1.0128869	SHITE	
FATE	. e	FEST	0.52867633	BISHT	1,2355864	X I T R	
	12.695079	¥01	9.A7177979	T >0	P.75945846	Y I T R	5.
MUKB	0,12723126	CTSIG	8.74784687E-1	LTOT	-6.4428239	ZITR	6
MCY8	8.20579478E-1	CHSIG	-8.625A0307E-3	DTOT	27,843685	LITA	6
MU28	-0.13517562E-2		9.47339813E-6	410	742,10517	MITE	6
LABBER	-8.25984665E-1	1	6.99869482	2 0 1	1182,8533	ST T	60
DABHER	8.24552989E-1	2	M.35762786E-5	KTRBLK	6	Q X 4	1.3213477
ě I.	975.73788	101	-178.37895	VX9001	-P.2P549174E-1	AYP	-0.23931713E-2
W I A	-337,66928		337,66928	VYBDOT	-8.21651338E-3	424	-32,137753
212	-15391,457		15412,693	VZRDOT	P. SOSASTEBELL	4 × >	181.38966
¥ 1.	-4566,9889	LBARH	-1401,2043	Poor	-P-18574855E-3	٥ >	16.478456
81	15944.185	ISARI	-4657.9277	1000	9.33884256F=3	479	1750010 4
812	21200.251	DBAR	21867.558	#00T	8.73797158E-3	PSTR	
MEM	-296,68329	L ×	-37,548457	X T R	6	PSTONG	•
¥ # ¥	-264,67329	+	-93,2657.87	4 7	A95.5230A	910	
4=2	121,50638	12	-197,69883	278	-253-15019	MADO	
	519,45500	11	-235,06988	410	4271.9263	XADD	6
N. H.	-4884,8329	H	-3103.6187	N I	-7843.4367	VADD.	
FER	-2226.0747	7	2623.4525	8-2	-21549.623	7450	6
XIX	-41.763245	K V T	4.2227878	ALFHTT	V. A.S. BOLA.	0042	. 6
THA	-1.7263883	* * *	-91.559779	ALFVTT	644044		6
1							

84.8	F 9CG	360,26900	>	8.00	PSITRE	6
60	9214	245.49999	STEE S	1 N	VXSTR	6
E	CI	9.1750000E-2	ON TOSA	1877.8	××××	. 6
37363.0	TIME	F. POSPOSOSE-1	DELBMR	6	VZSTP	6
20000	NBOR	6	EL LORL	2.011	PSTR	
137, 68499	26.36	E .	TESTA	-18.0	HLVT	273.0
5.8	PASCNT	1133.9	FLH	234.9	FSCT	
. 40808	PH.	45.6	377	32.300000	DSTR	
9.56688974	A18	-1.1933575	THI	-8-6948749A	XX	46.461919
1.662833	919	7.9755529	13	5	T X	9.000.0
16.569978	THETAE	16.569978	TIVSIR	6.4899789	i Li	38.862369
6,153759	THETTR	33,461733	-	19,951733	d.×	9.8991378
.6461939	ZIOX	9.98885882	XC IN	3. A862369	Z H C X	6.49134862
772750	YBACTI	1.0772750	PSTR.	6	PSTR	
166.71469	THETAB	3.2627225	AARF	2.6115622	æ en co	6
.392487	0110	5	AA1F	-4.3927164	200	6
. 518486P	RETAME	4. 4868741	981F	9.32969894E-1	TITE	6
	PAMC	6.00	AAPL	-3.8138951	III	6
	CHGRAT	1.0	AAIL	9.20262485	JITR	6
6.	PSIDOT	6.6	991	0.28571359	BILITA	6
.87610721	EKTX	1.3594662	EKMFX	0.92228996	LHI14	6.
61.714614	FKTZ	1,6972815	EXMF 2	1,9882853	SHITE	8.8
	FE SET	0.49584874	FROIS	9.4698986	XITR	8
575914	THO X	A. A7177979	¥0×	2,82515968	YITR	6
0,21205176	CTSIG	P.76163873E-1	LTOT	1.9034212	ZITR	8
93166446-1	PERC	-0.36147522E-2	DTOT	24,597374	LITR	6.0
7255781E-3	COHSIG	0.47180295E-6	414	898.22485	X I I	0.0
.144347766-1	7 N	0.99832785	TATA	1373,5913	01 H Z	6
.15407334E-1	ر د •	0.14305114E-4	FTRBLK	2.0	AXP	1.844898
1567.0244	HOAR	-745,75743	VXBDOT	9.18681947E-1	AYP	0.20342417E-1
-306,69643	JBAR	388.89643	VYBDOT	8.87911120E-2	4 Z P	-32,121002
651,540	70 AR	15713,324	VZBDOT	-8.53269903E-3	d×>	168.71469
-5362,3461	LBARK	-2163,1667	POOT	P.62930623E-2	4 × ×	15.392487
1.6491	HOARI	-14614.737	1000		424	9.4184869
24772,351	200	25436.977	POOT	-0.13285207E-3	ASTR.	6
-673.72858	×	1.9772588	XTR	8.0	PSIDHG	-150.0
-365,86192		-155.48613	ATA	844,11727	BTR	
-44.967869	17	390.99174	218	-307,23415	MADD	6
574.96908	-	-355.65588	178	5154.596A	MADD	6
-3596,5530	H	11100.077	0 - I	-9519,1381	YADD	6
-3244,7002	⊢ 2	4606.8950	œ ⊢ ?	-26153.566	ZADO	8
6,3009285	F ^ N	-4.3235685	ALFHTT	-5,1247551	MADD	6
-2.2841971	¥ v T	-142.68193	ALFVTT	5.2703628	LADD	6
. 55677	112	0.33497198	AABBIF	4.3926400		

of dispression of the same of	UTTAS(S76)	1-21-77	9-0EC-77		RUN AB		
MEIGHT	16459.8	900	368.20980	>	120.0	PSITRZ	6.
×	5130.0	9011	205.8999.	DELS	-5.0	VX8TR.	5
1	39815.8	OH	9.175PRJBPE-2	ONDOSA	1077.0	VYSTR	6
11	37363.9	7.1 ME	0.2600000E-1	PEL 3MR	6	VZSTR.	6
OFFICER	29.700000	SOON	6.4	RILOXI	5	PSTR	6
OMEGTR	137,98499	5000		THOTTE	80.00	K VT	273.0
KFR	15.0	PAGENT	2001	ı	6. 210	17.7	5
FBH	700.42900	THE	40.00	L > 6	000000 NM	0.00	
LATSTR	-8.59259338	A19	-1.3015991	THI	-2.2389183	×	46.358791
LAGSTA	13,365371	P1.5	9.9128524	IS	- N	C ×	2,7725191
COLSTA	17.461688	THETAB	17-461666	THYSHR	7.3816988	, U	45.635056
PEDAL	28,251935	THETTR	36.629866	TH75TR	23,129866	×	3.2825822
MAN	4.6356798	KBIN	0.27725393	KCIN	4.3635856	Z L d ×	6.17725851
XBACTP	2,6939957	XBACT1	F. 26939956	PSTR.	6	PSTR	6
0×0	202.64568	THETAB	2,2494494	AAGE	2.6291242	081R	
478	16.782781	PHIB	6	AAIF	-5.8593568	RSTR	6
87A	7.9698765	BETAME	4.5024199	9816	-0.54581383E-1	TITE	
۵.	8	SAMC	6	1644	14.6387554	TITE OF THE	6
o	6	OMGRAT	6	AASL	9.25629389	AL IT	. 6
æ		PSIDOT	5	F81	9.37982537	HILL	. 6
ALFAF	-0.65755128	FRTX	1,4167120	FKEFX	9,91669353	LHITA	6
CHITPP	61,284119	EKTZ	1,6666835	EXMF2	1.9087158	DHITE	6
14 X 14 X		EPSHT	P.49502420	PERM	8.19242593	XITR	6
W O	39.610794	KONT	0.67117979	KOVT	0.82703121	YITR	2
MUKB	0,25446263	CTSIG	8.76590534E-1	LTOT	2.0119965	ZITA	6.0
MCYS	8.28968984E-1	SISHO	-8.35879733E-2	DIOT	24,483483	LITR	8
MUZS	-0.3334106AE-2	COMSIG	9.47334975E-6	TTP	1099,7205	R H Y	6.0
LAMBMR	-0.162789226-1	2 N	0.99929199	KMAI	1665.8584	NITR	8.0
Or OIL	Ø.12936816E-1	U	#.35762786E-5	KTRBLK	6.	AXP	1.2792694
E H	1547,1941		-740.23269	VXBDOT	0.1556P934E-1	AYP	9.37064974E-1
Z X	-336,83437	JB 4 R	336.83437	VYBUDT	0.219167538-1	47 V	-32,148737
Z Z Z	-19760,327	TOAR	15821,979	V2900T	-8.17022975E-2	4×>	202,64568
-	-6549.9864	LBARH	-2969,4609	POOT	P.11288855E-1	440	16,702781
Z I	-27,869384	INTOI	-16784,418	DOOT	-8.89411943E-3	424	7.9600765
Q I	30092,179	DBAR	30849,082	P007	P.19095656E-2	PSTR.	6
M. W.	-936.71244	TX	-9.1132628	XTA	6.	PSIDMG	-158.8
# T >	-467,25286	14	-219.17378	YTR	1033,4761	978	
342	-64,926989	11	578.48395	219	-376,15526	MAPO	6.0
	731.43766	-1	-485.55995	L18	6347,6452	XAPD	
A.E.	-4769,7281	H.	16416.542	a L	-11654.543	VADD	6
L 3	-4142,5914	- 2	6116,2497	a L	-32828.534	ZAPO	0.6
H	-1,1846532	K > 1	-1.9286896	ALFHTT	-5.5807371	NADD	8.8
T.	-2.8915421	X V T	-216,28223	ALFVTT	4.7385911	LADG	6.8
712	578,82896	1 1 2	8,46299716	AABBIE	5.8596512		

i		II WE ON	!	9-DEC-77		RUN 41.		
	HE IGHT	16450.8	FSCG	369,29900	>	6	PSTTRO	6
	X	5130.0		245, A.9999	DELS	- N	- 1	. 6
	IY	39615.0	OHE	0.1752000E-2	VSOUND	1077.0	STR	6
	. 71	37363.0	TIME	0.20000000E-1	DELSMR	9.6	VZSTR	6
	OMEGHE	29.746966	800 N	6.0	THOUSE	6.01	2	6
	DHEGTR	137,80499	SSS	9.0	TESTIE	-16.9	1	273.9
	K P	15.0	PASCNT	3000	KLHT	234.0	FSVT	
	TOH!	700.40000	SHT	45.0	SVT	32,390000	DSTR	6
	LATSTK	-0.60557824	A 1.5	-1.4586410	IHI	-2.9279675		46.215136
	LNGSTK	14,14999	818		13	0.5	69 X	
	COLSTK	16.659250	THETAB	16,659250	TH75MR	4,7792502	Ų ×	52.370315
	PEDAL	29,436888	THETTR	37.5	TH75TR	6.9	ď	
	ZIVX	4,6215136	x BIN	6	XC1X	5.2370314	XHAX	6
	XBACTP	-2,2892616	XBACTI	-0.22892616	RSTR.	5	PSTR	6
	0 × >	236,58176	THETAB	9.26934432	-	2,7050066	S TR	6
	VYB	15,219806	PHIB	6	AA1F	-4.6548219	RSTR.	6
	824	1.1946927	BETAMF	3,5528986	9816	-8-95242330E-2	TITE	. 65
	Q.	6	GAMC		AABL			6
	0	6	OMGRAT	6.7	AATL	できたいという。		. 6
		6	PSIDOT	8	9811	0.43714846	TI I	. 6
		-1.8636789	EKTH	1.4682427	EXEFX	9.90272122	HITE	. 6
	CHITPP	80.209326	EKT2_	1.7215207	EKMFZ	1.9097906	SHITE	6
		5.	FEBRE	B.48552898	LEGIS	P.62529291	X	6
		52.689486	K G H I	0.87177979	KOVT		YITR	5
		9.29654438	CISIG	0.78912872E-1	LTOT	1847054	ZITR	6
		8.19895845E-1	CHSIO	-0.23192228E-2	DTOT	24,290560	LITA	6
		-0.148486966-1	COMSTG	0.48868454E-6	118		MITR	6
	LAMBAR	-0.25449039E-1	ZN		a z	2093,1228	Z L I Z	6
	RILDRO	P. 11408342E-1	ر د د	9.49486967E-6.	KTRBLK	6.	AXA	-6,33278907
	r I	1329.0127	K 401	-470.47751	VXADOT	-0.181703796-2	AYA	-0.29573595
		-381.34571	JOAR	301,30571	VYROOT	0.246242798-1	AZP	-31,583815
	MH 2	-16231,851	104R	16260.468	V2900T	-0.51759589E-2	V × P	236,58176
	E (-7453.7950	LOARH	-3129,2777	1004	0.15742325E-1	4 >	15,216543
		2899.0574	HBARH	-16040.734	1000	0.23610282	424	1,1947452
		37721.548	38 A R	38576,358	P 001	0.14967960	ASTR.	6.0
	×	-1251,4863	×	-0.24323272	XTR	5. 6.	PSIONG	-150.8
	L X	-490,33837	11	-246.59914	418	1132,2927	BTR	6
	M X N	31.386352	7.7	973.57546	218	-412,12164	MADU	6
	ر. د	605.01173	-1	-549.58561	LT.8	6954,5882	XADD	6.
	FAF	-6382,3895	-	27552.566	a F	-12768.902	YADD	6
	3 2	-4356,9589	- z	6937.5548	۵- ۲	-35282,282	ZAND	6
	i i	15.929387	× > 1	-16,172628	ALFHTT	-7,1179811	NADO	5
	L I	-3.6412724	>	-244,95787	>	3,6724348	LADD	5
	1 H 2	972,35537	112	1.2110010	AABBIF	4,8848312		

			1																			-						1															
6.0	6.	6.0	<i>c.</i>	5. E.		-		48556				7.4513969	10	6	2	6	2	6	6	0	6	6	0	 	50	60	ě.	9,45826824	-0.8282935AE-1		253,44898	28,427700	3,3433339	5	-150.0	0.4	6	6	2	15.	6	6	
PSITRE	CXSTP.	V × S + V	VZSTP.	PSTR.	HLVT	FSVT	DSTR.	KA	Ø2	UX	Q.	XI dx	PSTR	STR	RSTR	TITR	HITR	JITR	KILLE	4111	BHITE	FITE	YITR	2178	LITR	œ	NITR	TXP	•		Δ.		424	ď	E	4	MADD	MADD	VADD	5	OCAN	LADD	,
150.0		1017.0	6.	-16.0	1.5.9	234,9	32,300000	-3.7988859	-3.6	9.6270495	18-632787	5.7669369	5	2.6850514	•	-0.35307699	9	P. 28942532					٠.	•	24,475008	•	2430.2480	1.0	8.39764544E-1	-0.47056247E-1	8.25432997E-1	-8.24548428E-1	8.84535446E-2	-0.41262738E-2		1520,7361	-553,59377	9348.4132	-17149.391	-47117.475	-7.8293717	4.6289654	6,6773781
i	0513	A SOUNC	DEL 3MR	TERLER	TESTTR	E L E T	3 4 1	IHI		H75MR	STR				AAIF	9616	AABL	AA1L	RBIL	FEF	EKMFZ	SIGHT	X 0 × 1	LTOT	DIOI	TIR	a i	KTEBLK	VXBDOT					i		414	218	LTR	2	2	ALFHTT	ALFVTT	AABBIF
369.2000		7-1725686-12		8.	E . W	1217.9	45.0	-1.6324957	2.14922	6	4		9.6	0.75574496	5		6.0	6		1.5297625	1.7102374	0.49475044	B. A7177979	8.78358968E-1		8.48587945E-6		P.17881393E+5	-014,62689	583,99743	16164,541	-4439.2006	-22055,029	45964,594	-15,165101	-336.479AB	1647.3787	-743,22357	30854.566	398.091		331,353	0.74471134
(5)			,	888		ASCNT	SH4	413	A1S	THFTAE	THETTR	KO IN	XBACTI	THETAB	PHIR	BETAWF	GAMC	OMGRAT	-		EKT2			CTSIG	9181	1816			•	JBAR	TOAR	LOARH	MOARI	DBAR	L X	+	7.7	-1	I	+2	X V T	► ^ →	2 v T
16456,0	1 20 0	•	2		137,00499	15.0		-0.68238961	611358	9,707649	3.050248	4485564	1290514	53	9.4277	343333				1.13887	9,199581	5.6		.3176221	-25635532F-	.1245603	0,230?1352E	565319E-	65	9974			5293,818	43669,137	418,641	13	96664	1116,3496	-8874,7843	200	02438	12587	86.63
HE I UHT	4 3	- !	- 3		DIFETR	# X	FBH	LATSTK	LAGSTR	COLSTR	PEDAL	KAIN	XDACTP	e x >	8 4 7	97A	•	•	~	ALFAF	Crithp	ERTH	P * 0	ACK B	SAOM	5204	LAKBER	DEGILER	Z X	X 7. W	212	Ī	Œ I	e I	N X	A 2 A	2×2		F	Nak	HIM	+ I +	242

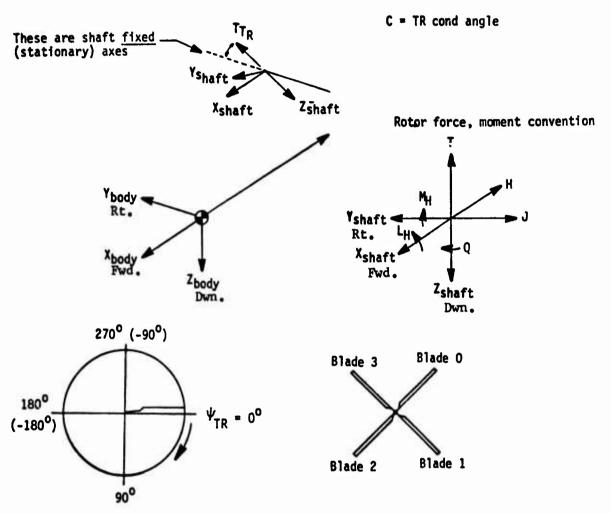
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	5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5 5				\$ 1 N & & & & & & & & & & & & & & & & & &
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SUN PER	1.00 1.00 1.00 1.00 1.00 1.00	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	-6.53:04:04:04:05:04:04:04:05:04:04:04:04:04:04:04:04:04:04:04:04:04:		
	0	1	######################################	***	XX 2 X X X X X X X X X X X X X X X X X
9-DEC-77	69.28888 44.49899 17588886 208886886	9-9-9-9-9-9-9-9-9-9-9-9-9-9-9-9-9-9-9-	2 2 2 3 3 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	20.00 00.00	10 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
-21-77	# 3 # F Z		E G G G G G G G G G G G G G G G G G G G	TI HOR WORLH	X
UITAS(576) 1	M		112751E-2 329242 529737		5593 593 128 128 575
1		.<************************************			

APPENDIX C

AIRCRAFT "HANDS OFF" RESPONSE

This Appendix contains the time histories, (Figures C-2 through C-13), of the aircraft for a period of six seconds following blade loss. The axis system, the parametric definition, and the sign convention used are shown in Figure C-1. Shown below is a definition of the symbols used.

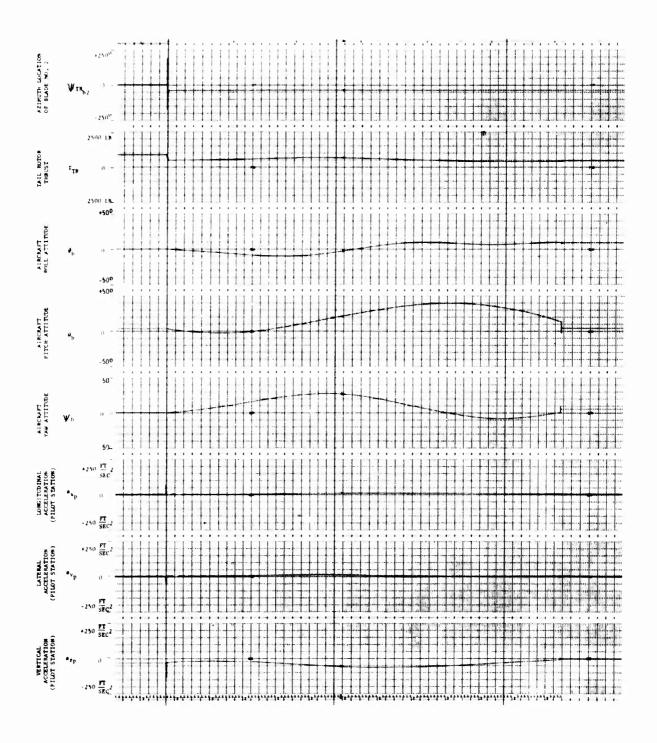
Symbol	<u>Definition</u>
ψ TR $_{ m b}$ 2	Tail Rotor (TR) Azimuth Position of Blade #2
TTR	Tail Rotor Thrust
φь	Aircraft Roll Attitude
heta b	Aircraft Pitch Attitude
ψь	Aircraft Yaw Attitude
a_{X_p}	Longitudinal Acceleration at Pilot's Location
ayp	Lateral Acceleration at Pilot's Location
a _{zp}	Vertical Acceleration at Pilot's Location



Notes:

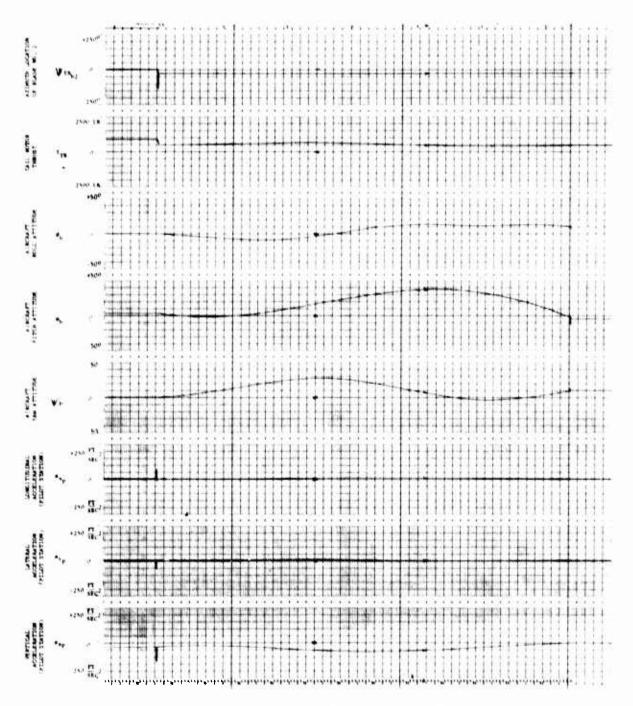
- 1. Tail Rotor Severance Simulation assumes Blade 0 is entirely lost due to ballistic damage (worst case); then, Blade 2 is completely severed when it reaches the jettison envelope.
- 2. Computer program required defining $T_R = \pm 180^{\circ}$ rather than 0° to 360° .
- 3. Jettison window used in the simulation study was $45^{\circ} \le \Psi_{TR} \le 25^{\circ}$ to examine performance characteristics for a blade loss range that exceeds the required 42° window discussed in part 1 of this report.

Figure C-1. Axis System, Parametric Definition, Sign Convention



G.W.: 16,450 FSCG: 360.2 ψ Damage: 0° SAS: OFF

Figure C-2. Aircraft "Hands Off" Response (0^0)



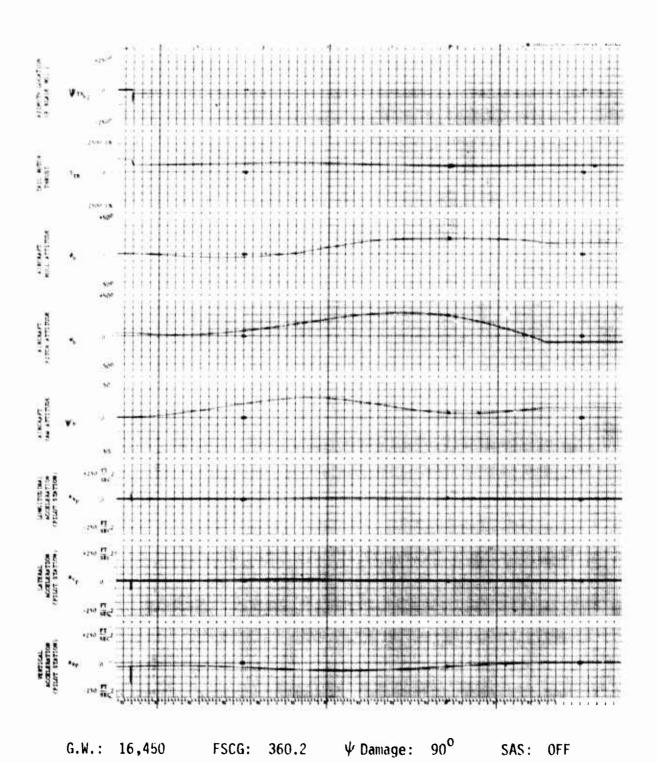


Figure C-4. Aircraft "Hands Off" Response (90°)

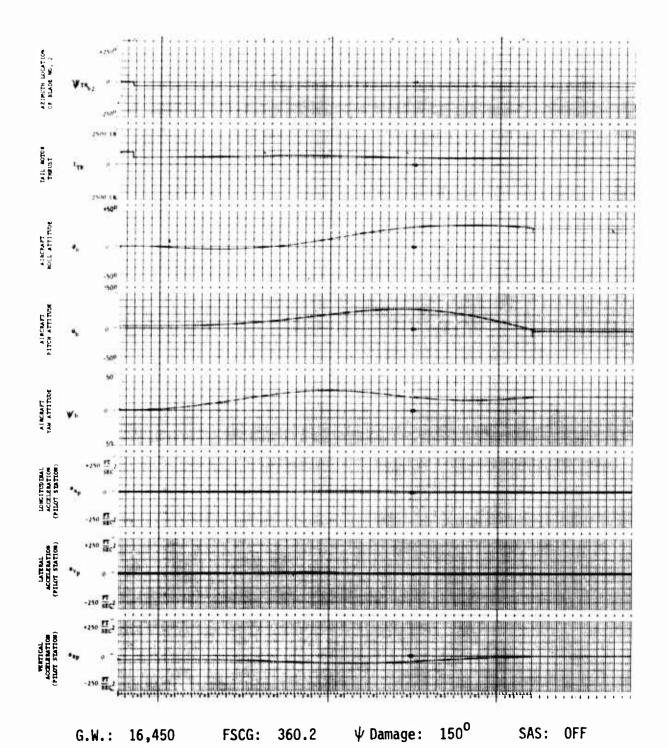
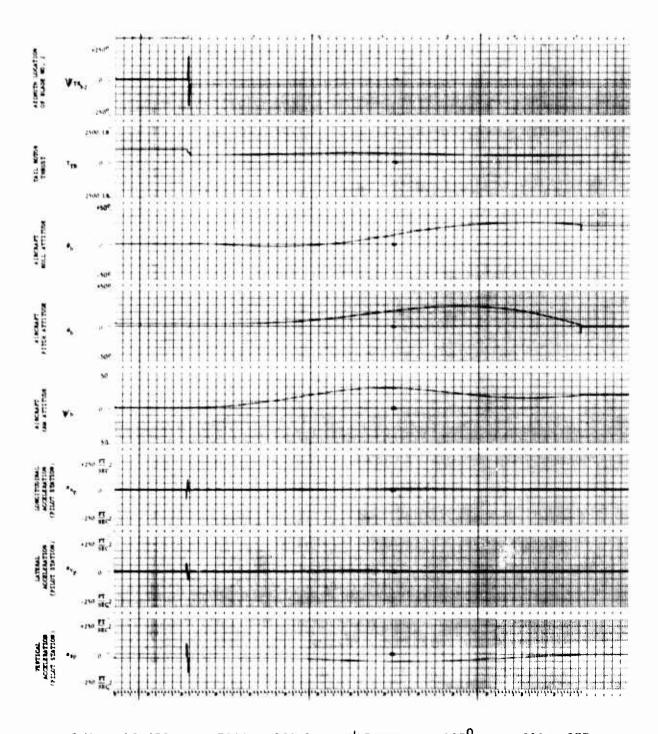
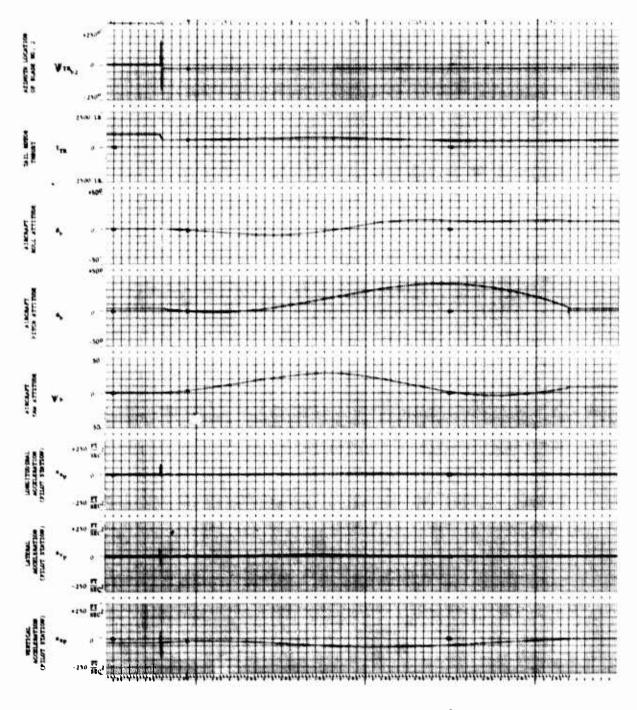


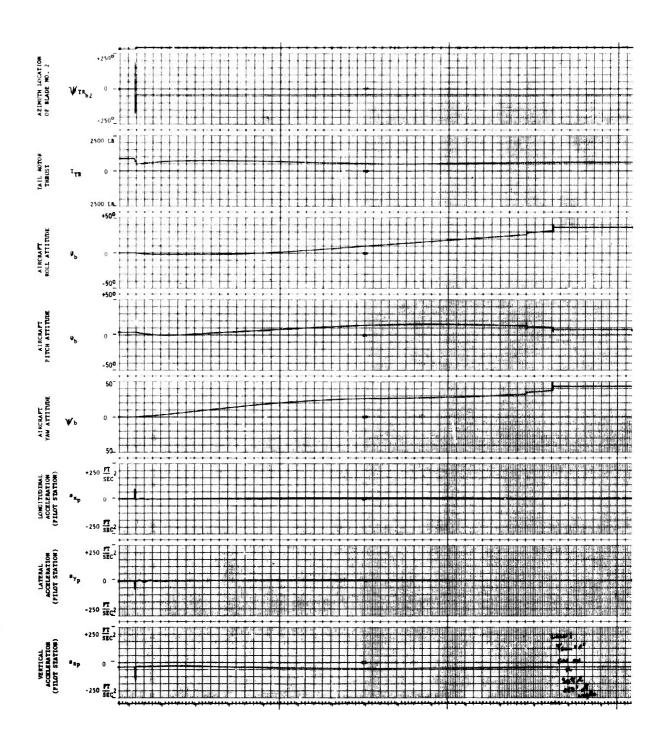
Figure C-5. Aircraft "Hands Off" Response (150°)



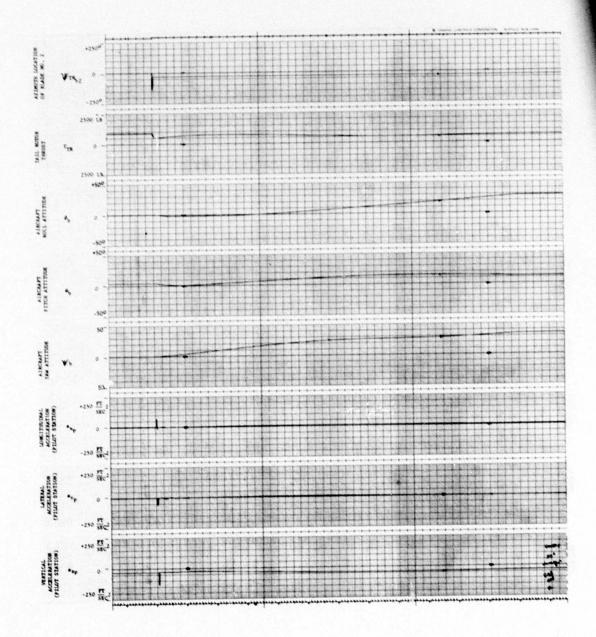
G.W.: 16,450 FSCG: 360.2 ψ Damage: -135° SAS: OFF Figure C-6. Aircraft "Hands Off" Response (-135°)



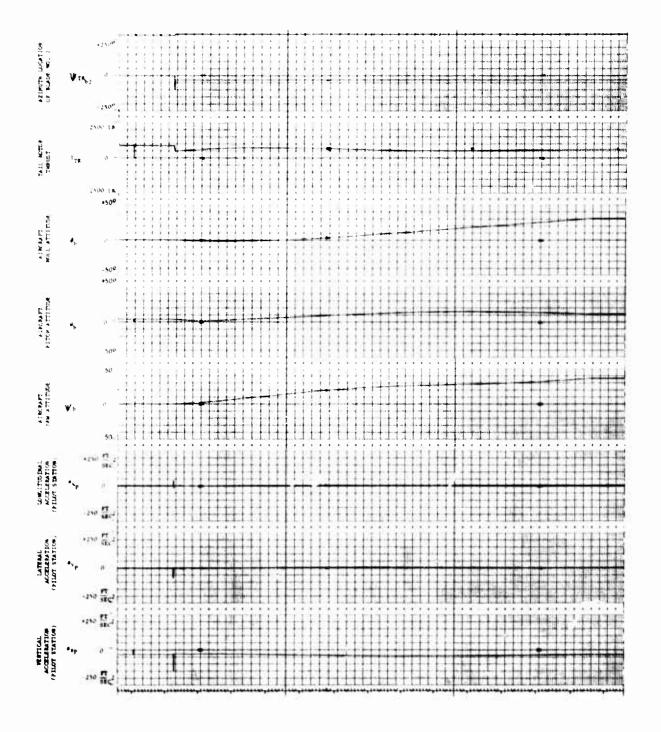
G.W.: 16,450 FSCG: 360.2 Ψ Damage: -50° SAS: OFF Figure C-7. Aircraft "Hands Off" Response (-50°)



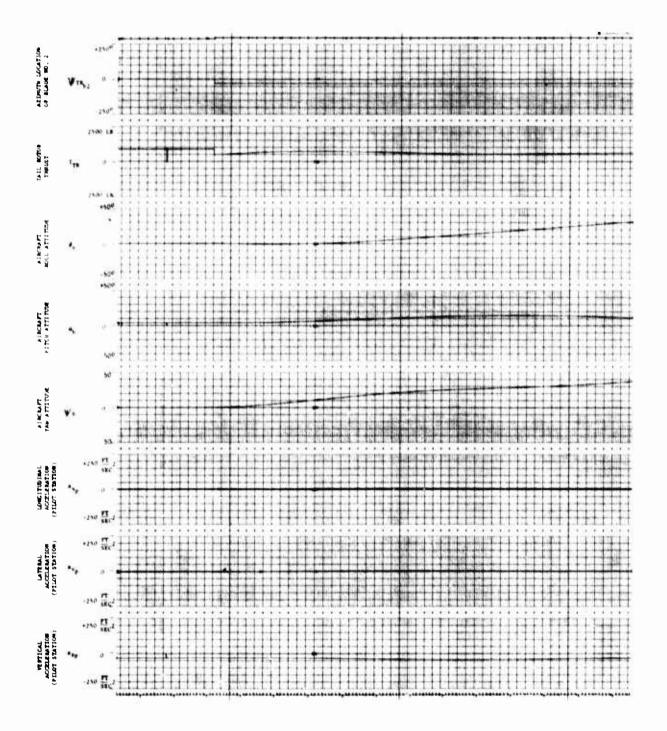
G.W.: 16,450 FSCG: 360.2 ψ Damage: 0° SAS: ON Figure C-8. Aircraft "Hands Off" Response (0°)



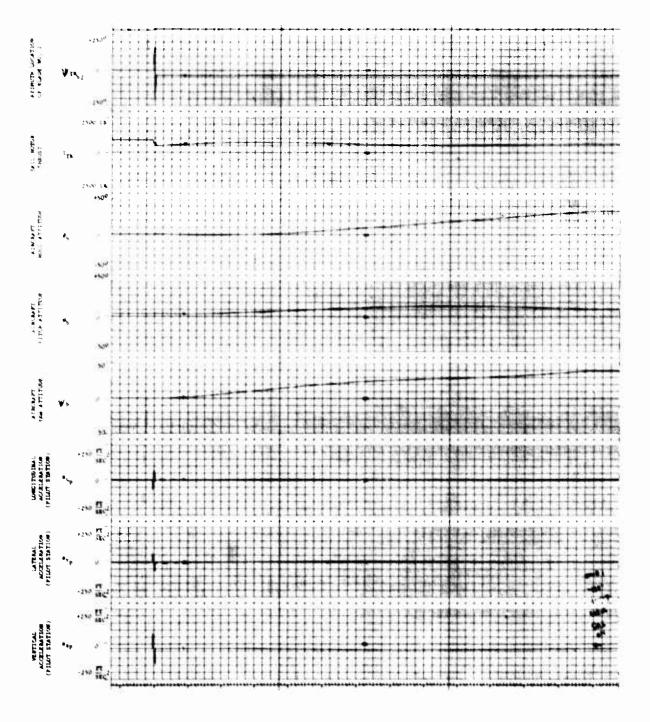
G.W.: 16,450 FSCG: 360.2 ψ Damage: 50° SAS: ON Figure C-9. Aircraft "Hands Off" Response (50°)



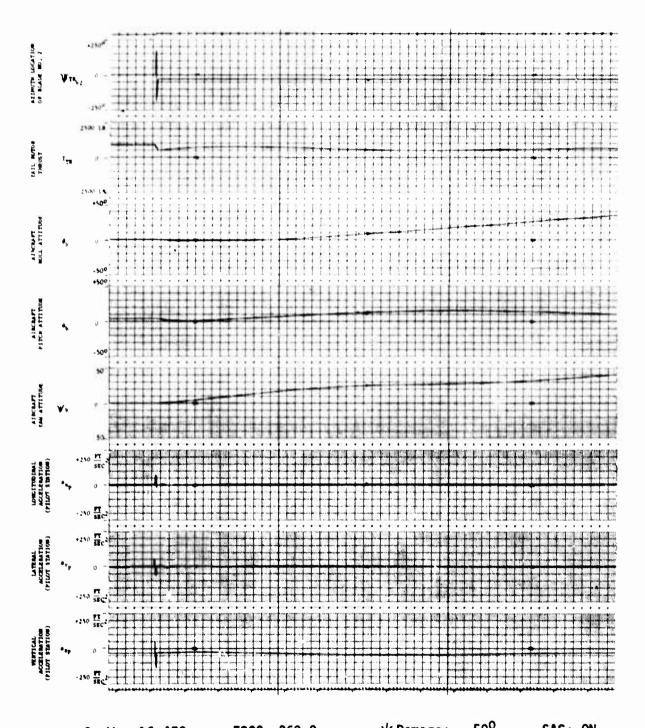
G.W.: 16,450 FSCG: 360.2 ψ Damage: 90° SAS: ON Figure C-10. Aircraft "Hands Off" Response (90°)



G.W.: 16,450 FSCG: 360.2 ψ Damage: 150 SAS: ON Figure C-11. Aircraft "Hands Off" Response (150)



G.W.: 16,450 FSCG: 360.2 ψ Damage: -135° SAS: On Figure C-12. Aircraft "Hands Off" Response (-135°)



G. W.: 16,450 FSCG: 360.2 ψ Damage: -50° SAS: ON Figure C-13. Aircraft "Hands Off" Response (-50°)

APPENDIX D

STEPPED TRANSITION TIME HISTORY TRIM DATA

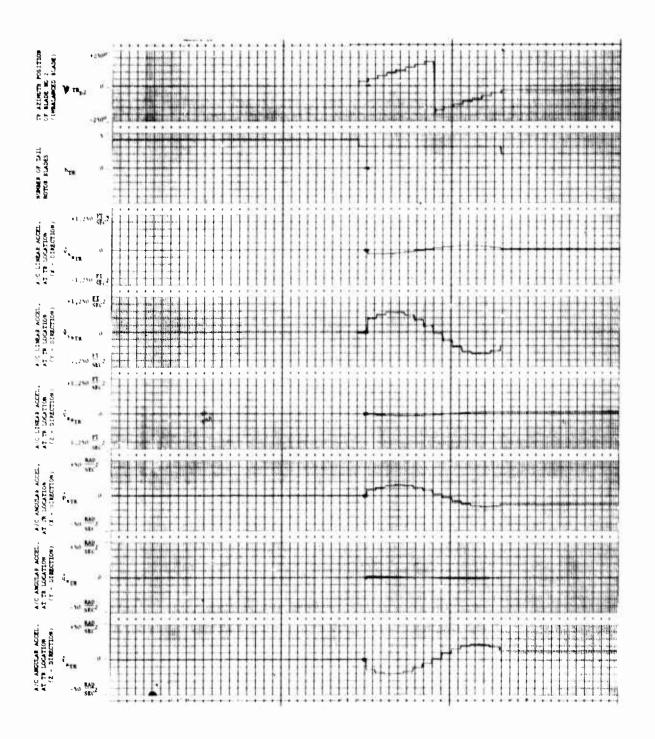
Figures D-1 through D-12, included in this Appendix, are the time history cases performed to determine the helicopter response to the unbalanced load during transition from four to two tail rotor blades. With the exception of Figure D-8, computer printouts have been provided for reference. Definitions for the key symbols and abbreviations used are provided below.

Symbol	<u>Definition</u>
TR b2	Tail Rotor (TR) Azimuth Position of Blade #2 (Imbalanced Blade)
bTR	Number of Tail Rotor Blades
^v xsTR	Aircraft Linear Accelerations at TR Location (FSTR, BLTR, WLTR) Measured Along TR Fixed Shaft in X-Direction
V _{ysTR}	Aircraft Linear Accelerations at TR Location (FSTR, BLTR, WLTR) Measured Along TR Fixed Shaft in Y-Direction
v _{xstr}	Aircraft Linear Accelerations at TR Location (FSTR, BLTR, WLTR) Measured Along TR Fixed Shaft in X-Direction
v zstr	Aircraft Linear Accelerations at TR Location (FSTR, BLTR, WLTR) Measured Along TR Fixed Shaft in Z-Direction
P _{STR}	Aircraft Angular Accelerations at TR Location (FSTR, BLTR, WLTR) Measured About TR Fixed Shaft in X-Direction
q _{sTR}	Aircraft Angular Accelerations at TR Location (FSTR, BLTR, WLTR) Measured About TR Fixed Shaft in Y-Direction
r _{sTR}	Aircraft Angular Accelerations at TR Location (FSTR, BLTR, WLTR) Measured About TR Fixed Shaft in Z-Direction

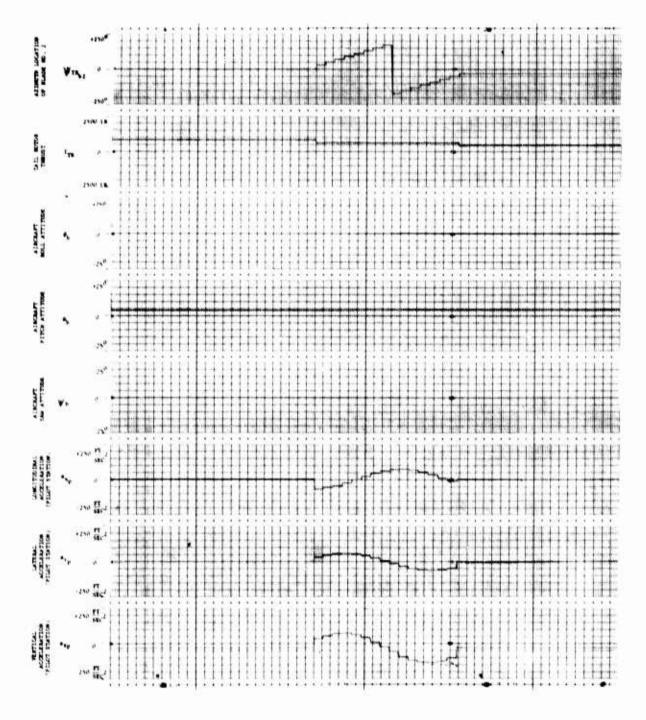
<u>Symbol</u>	<u>Definition</u>
T _{TR}	Tail Rotor Thrust
φ b	Aircraft Roll Attitude
θ b	Aircraft Pitch Attitude
ψb	Aircraft Yaw Attitude
^a xp	Longitudinal Acceleration at Pilot Location (Along Body Z-Direction)
a _{yp}	Lateral Acceleration at Pilot Location (Along Body Y-Direction)
^a zp	Normal Acceleration at Pilot Location (Along Body Z-Direction)
F _{yITRb3}	Inertial Force Along TR Blade #3 (Spar) in the Y- Direction
*T _{ITR}	Total (Sum of All Three Blades) TR Inertial Force Along TR Shaft (-) Z-Direction
*HITR	Total (Sum of All Three Blades) TR Inertial Force Along TR Shaft (-) X-Direction
*JITR	Total (Sum of All Three Blades) TR Inertial Force Along TR Shaft (-) Y-Direction
*MHITR	Total TR Hub Pitching Moment Due to Blade Inertial Loads (Moment About Shaft Y-Axis)
*L _{HITR}	Total TR Hub Rolling Moment Due to Blade Inertial Loads (Moment About Shaft X-Axis)
*Q _{HITR}	Total Moment About Shaft Due to Blade Inertial Loads (Moment About Shaft Z-Axis)
P	Aircraft Roll Acceleration
• q	Aircraft Pitch Acceleration
ŕ	Aircraft Yaw Acceleration
p .	Roll Rate
q	Pitch Rate

Symbol	<u>Definition</u>
r	Yaw Rate
٧	Airspeed

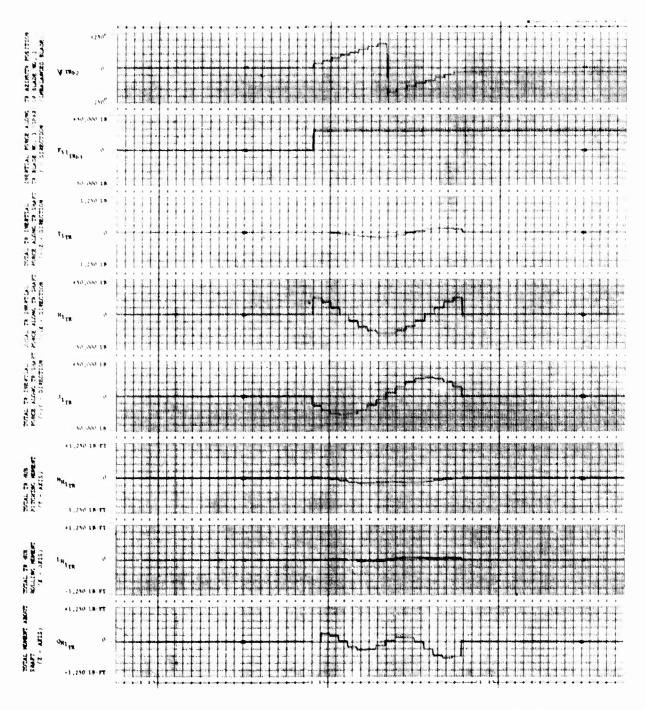
*These total forces and moments imply the summation for all three blades and are the inertial contributions without aerodynamic contributions.



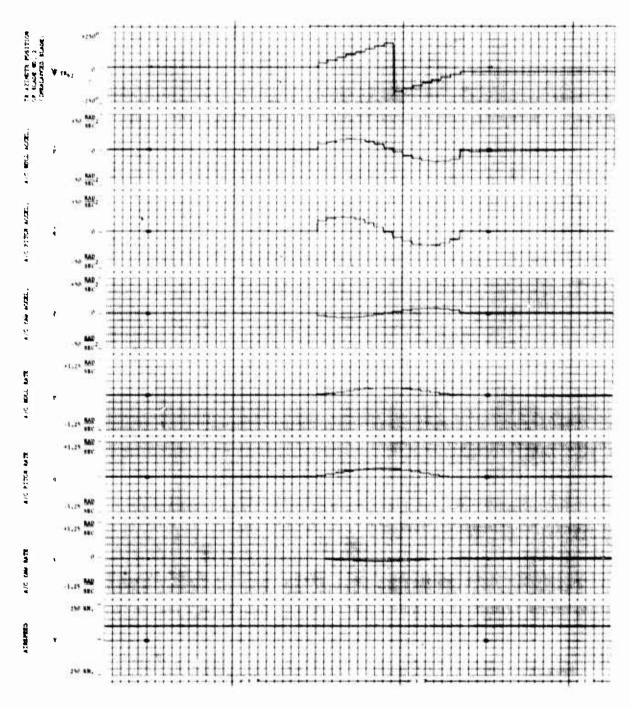
G.W.: 16,450 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS Figure D-1. Stepped Transition Time History



G.W.: 16,450 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS Figure D-1. (continued)

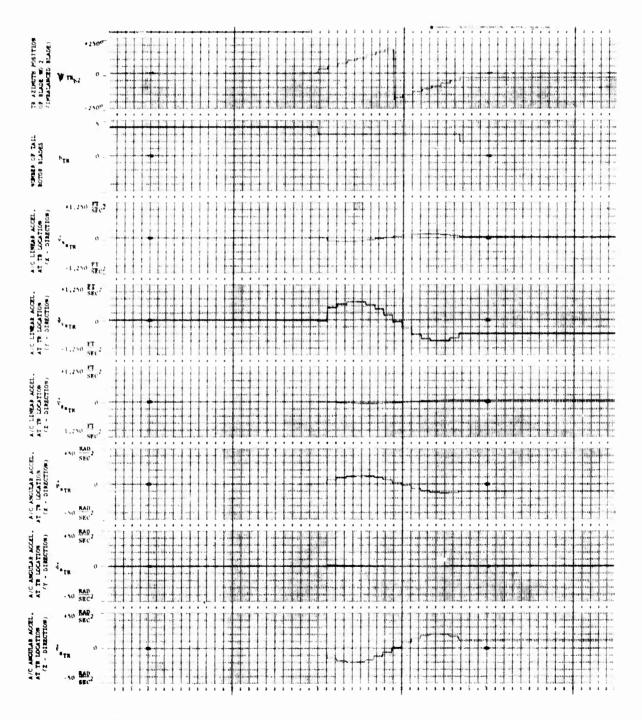


G.W.: 16,450 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS Figure D-1. (continued)



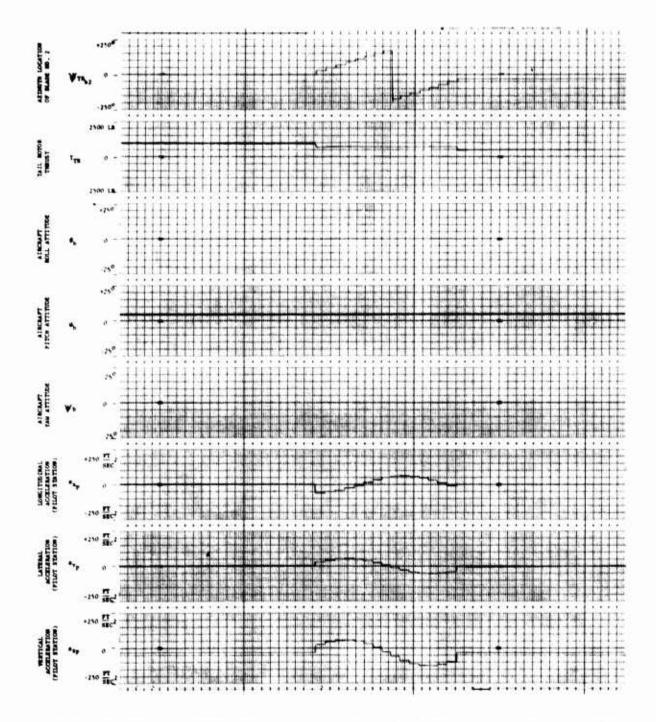
G.W.: 16,450 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS Figure D-1. (continued)

Figure D-1. (continued)



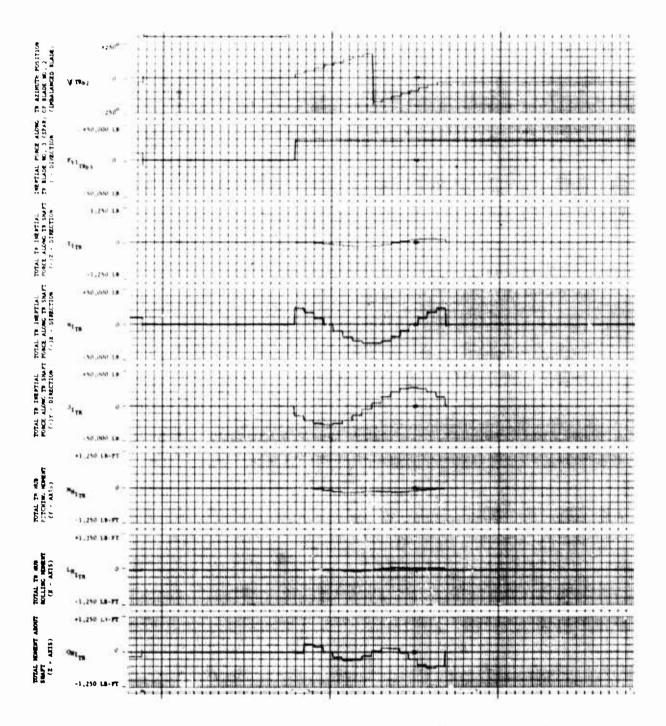
G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS

Figure D-2. Stepped Transition Time History

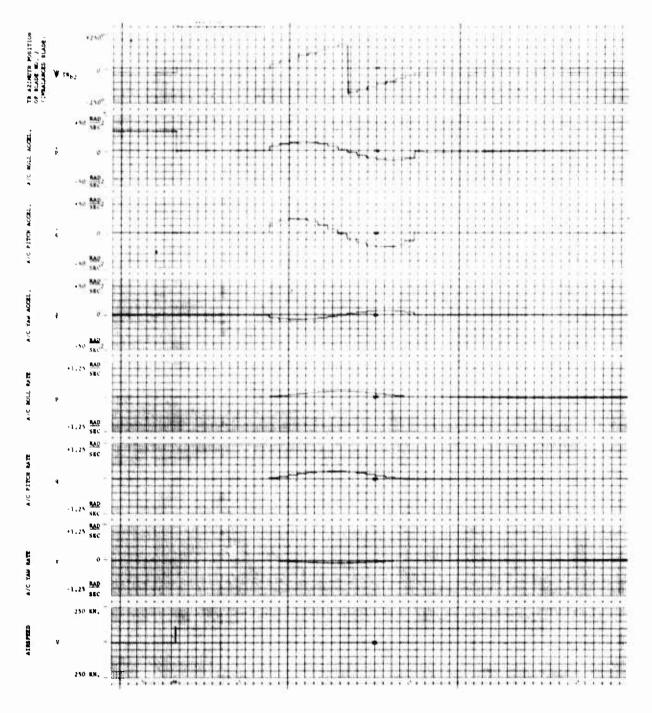


G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS

Figure D-2. (continued)

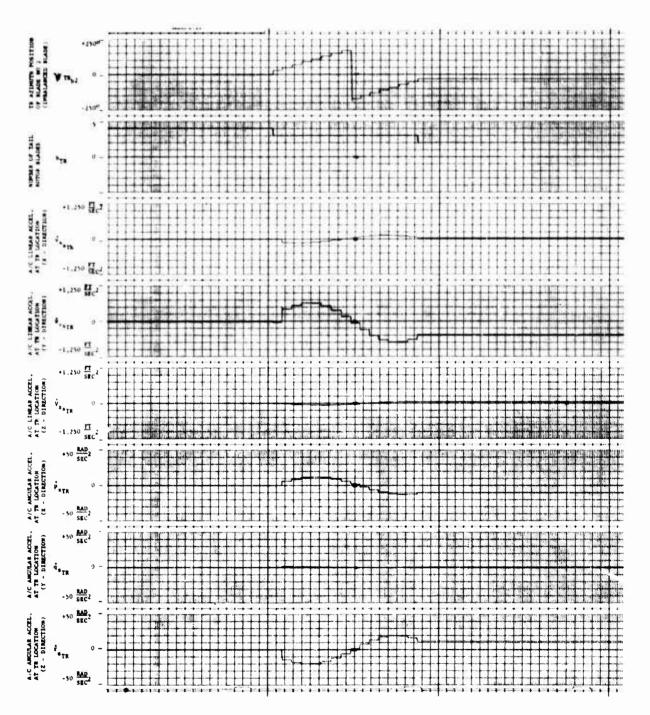


G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS Figure D-2. (continued)



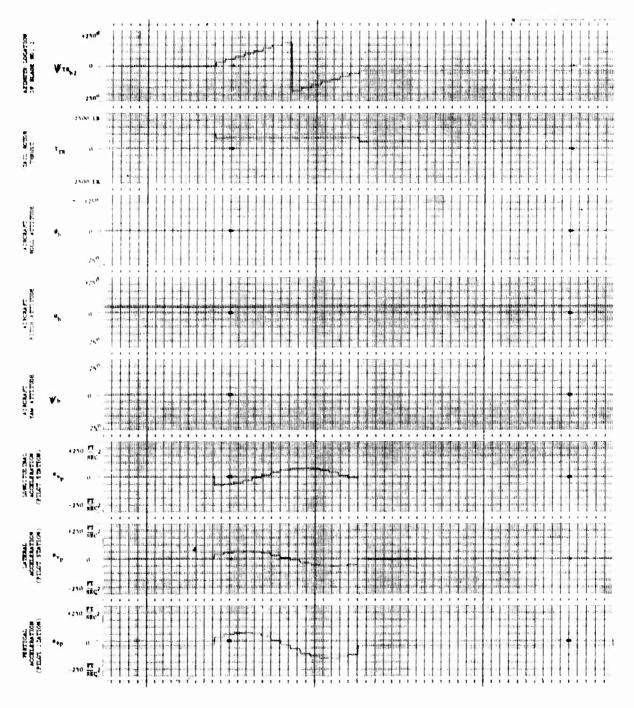
G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : SLS Figure D-2 (continued)

Figure D-2. (continued)

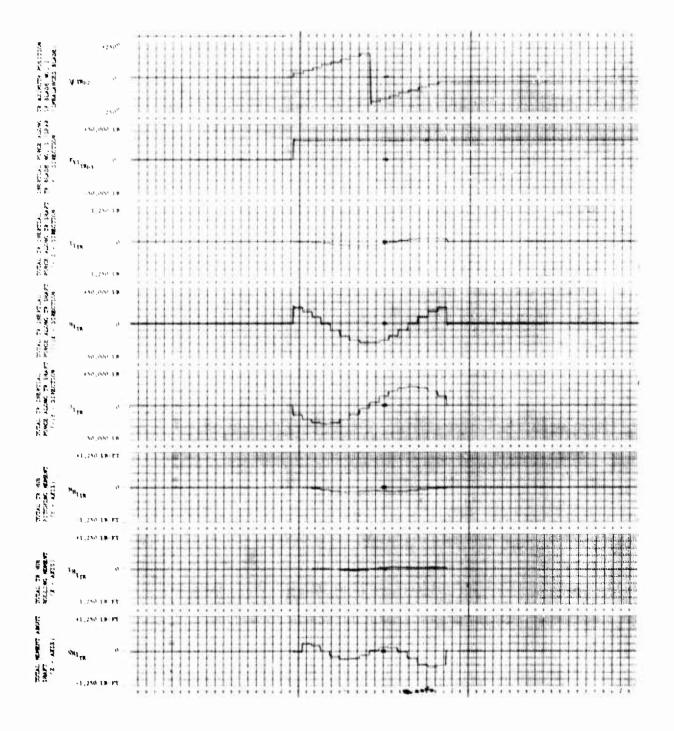


G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : 10,000 Ft

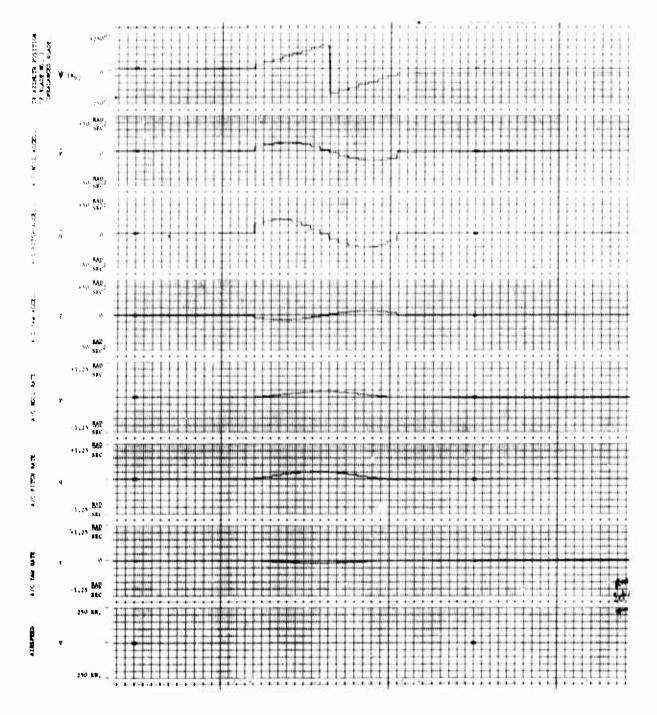
Figure D-3. Stepped Transition Time History



G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-3. (continued)



G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-3. (continued)



G.W.: 19,900 Lb. FSCG: 360.2 V: 100 Kts N_R : 100% SAS: 0FF H_D : 10,000 Ft Figure D-3. (continued)

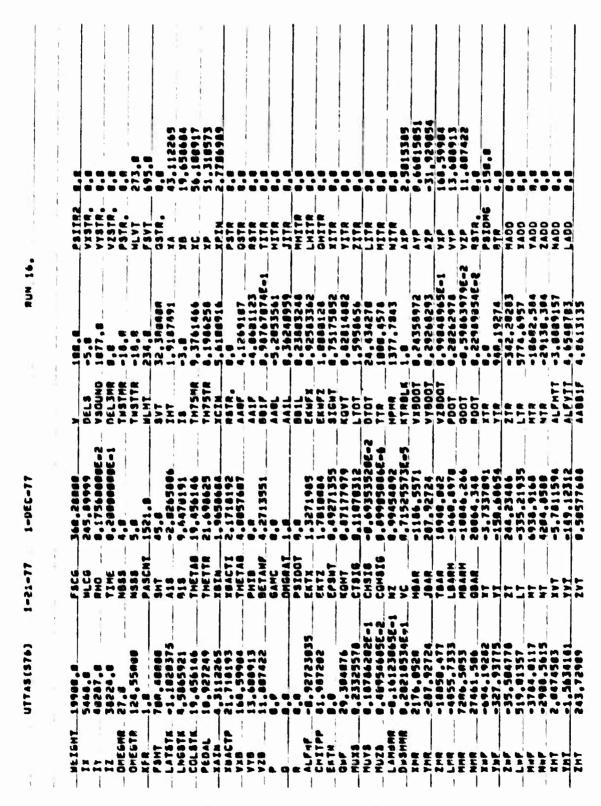
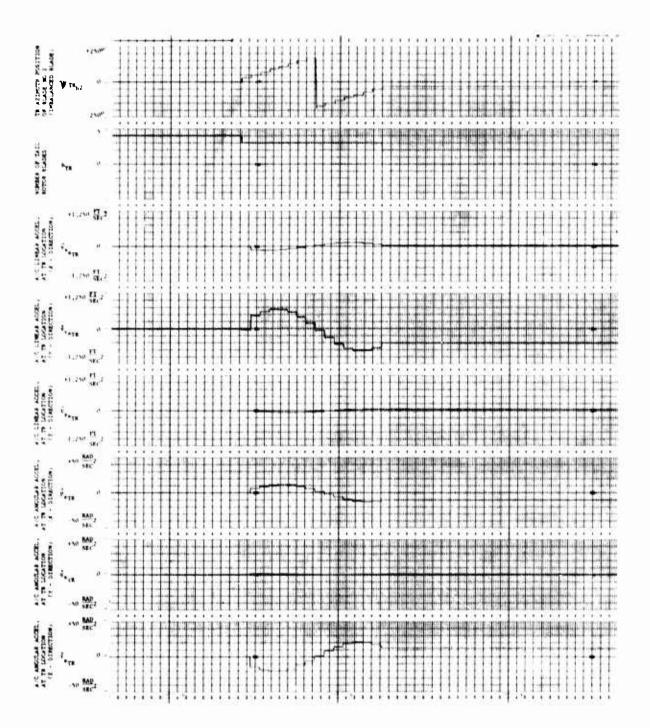
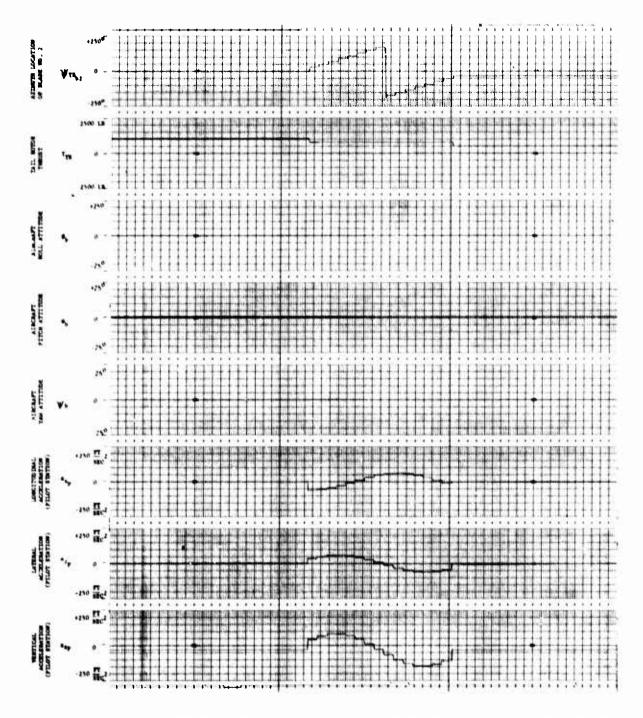


Figure D-3. (continued)

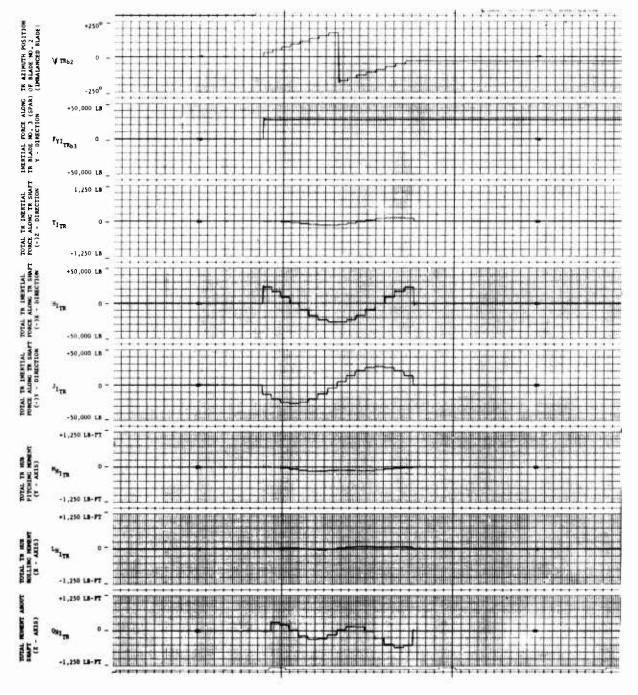


G.W.: 19,900 Lb. FSCG: 347 V: 100 Kts N_R : 100% SAS: 0FF H_D : 10,000 Ft Figure D-4. Stepped Transition Time History



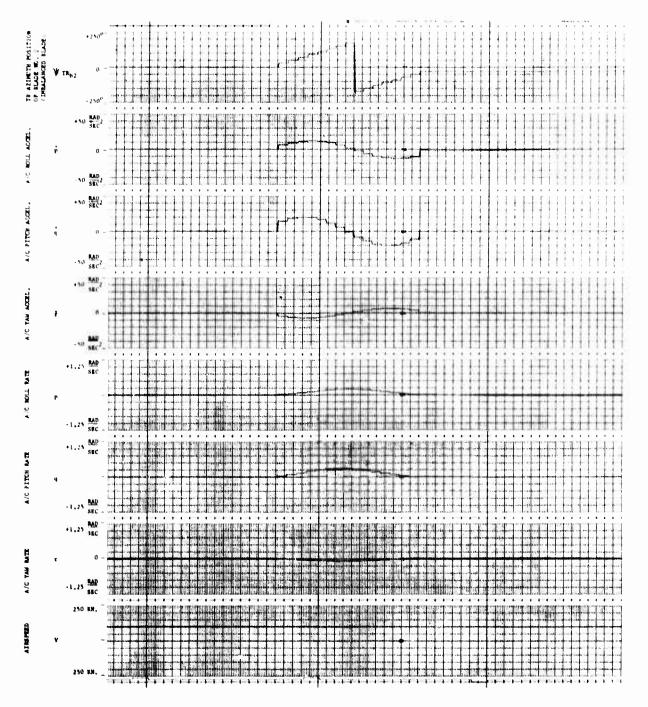
G.W.: 19,900 Lb. FSCG: 347 V: 100 Kts N_R : 100% SAS: 0FF H_D : 10,000 Ft

Figure D-4. (continued)



G.W.: 19,900 Lb. FSCG: 347 V: 100 Kts N_R : 100% SAS: 0FF H_D : 10,000 Ft

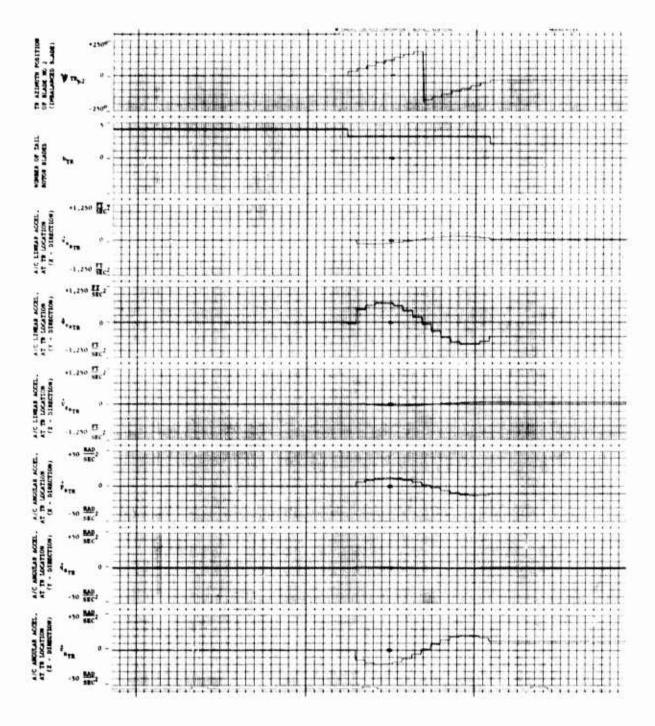
Figure D-4. (continued)



G.W.: 19,900 Lb. FSCG: 347 V: 100 Kts N_R : 100% SAS: OFF H_D : 10,000 Ft

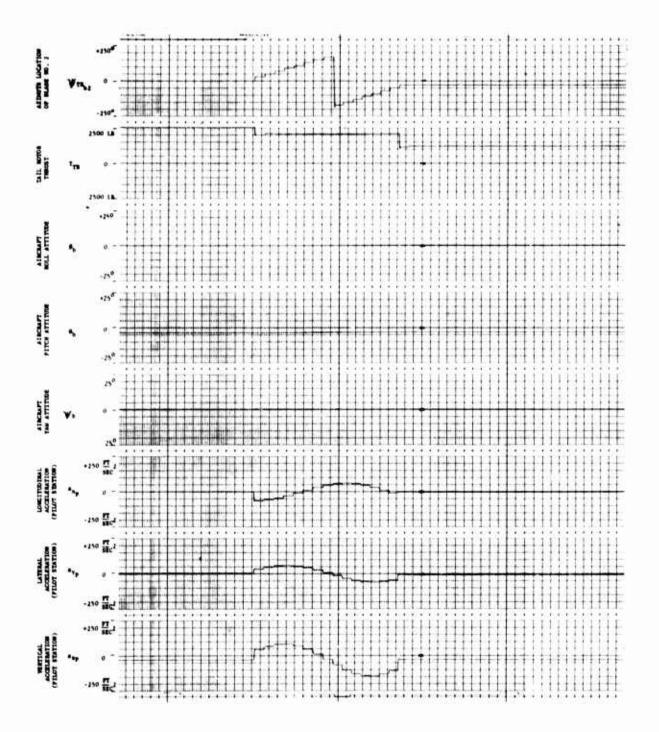
Figure D-4. (continued)

Figure D-4. (continued)

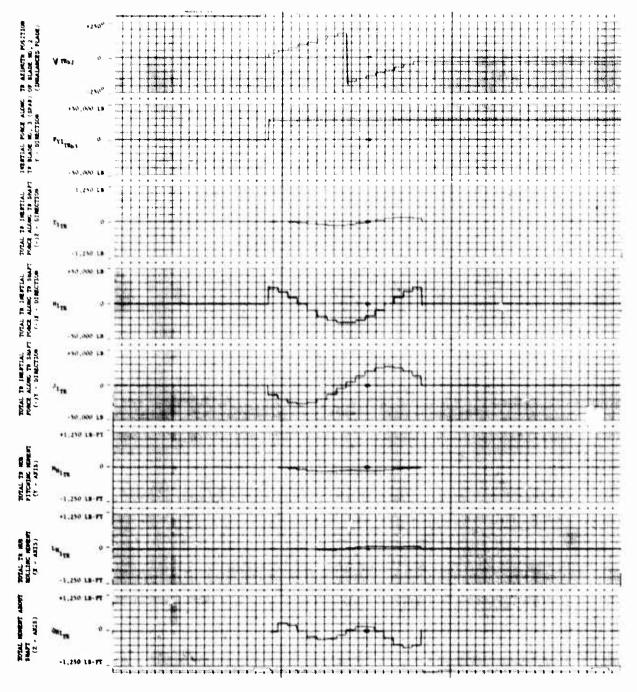


G.W.: 19,900 Lb. FSCG: 347 V: 150 Kts N_R : 100% SAS: 0FF H_D : 10,000 Ft

Figure D-5. Stepped Transition Time History

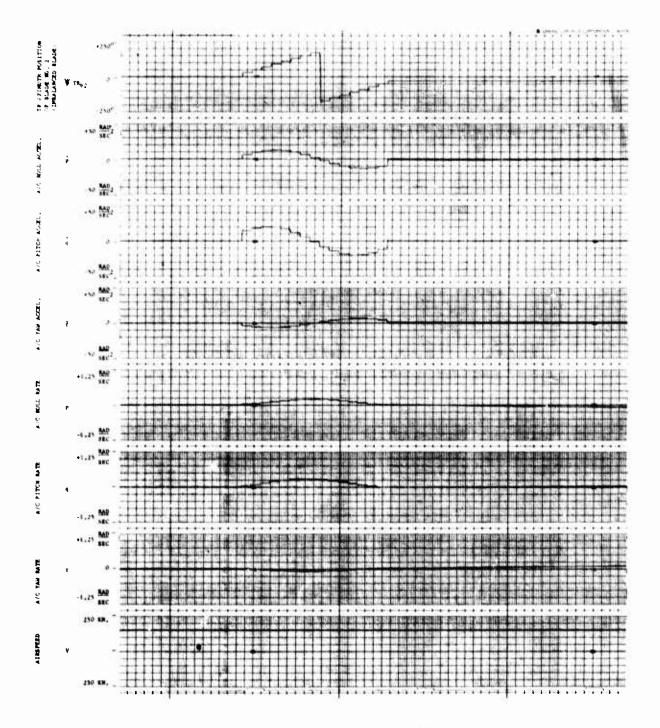


G.W.: 19,900 Lb. FSCG: 347 V: 150 Kts N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-5. (continued)



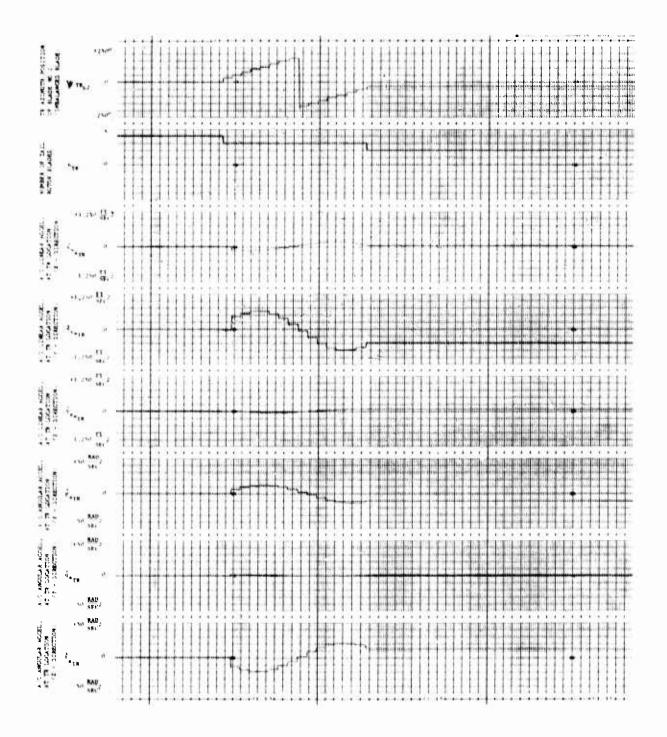
G.W.: 19,900 Lb. FSCG: 347 V: 150 Kts N_R : 100% SAS: OFF H_D : 10,000 Ft

Figure D-5. (continued)



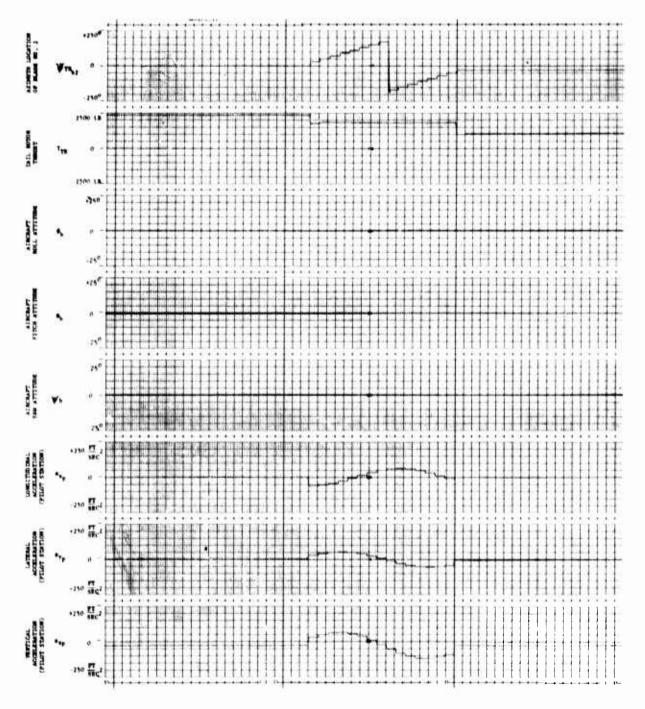
G.W.: 19,900 Lb. FSCG: 347 V: 150 Kts N_R : 100% SAS: 0FF H_D 10,000 Ft Figure D-5. (continued)

Figure D-5. (continued)

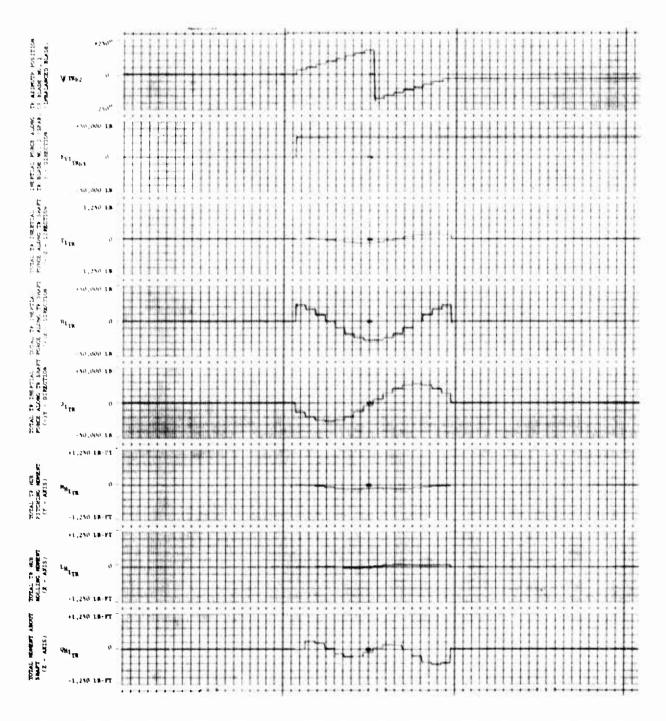


G.W.: 19,900 Lb. FSCG: 360.2 V: 150 Kts N_R ; 100% SAS: ON H_D : 10,000 Ft

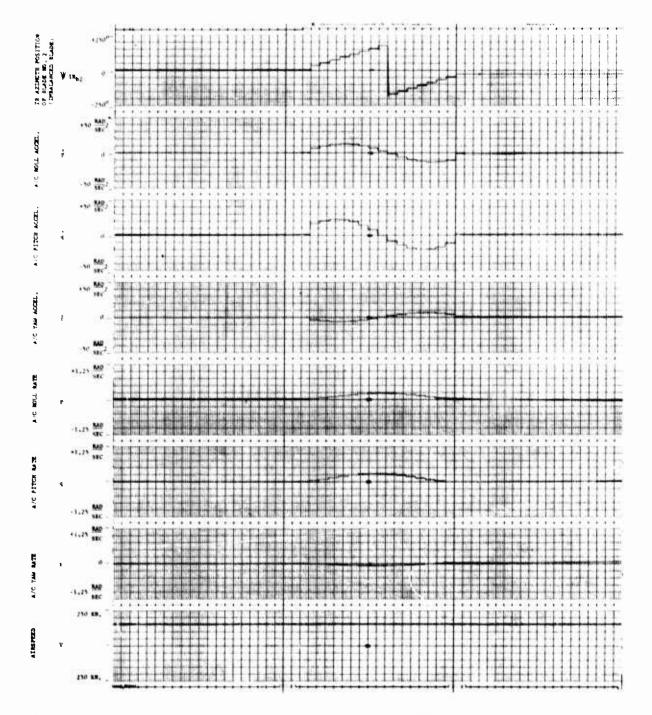
Figure D-6. Stepped Transition Time History



G.W.: 19,900 Lb. FSCG: 360.2 V: 150 Kts N_R : 100% SAS: ON H_D : 10,000 Ft Figure D-6. (continued)



G.W.: 19,900 Lb. FSCG: 360.2 V: 150 Kts N_R : 100% SAS: ON H_D : 10,000 Ft Figure D-6. (continued)

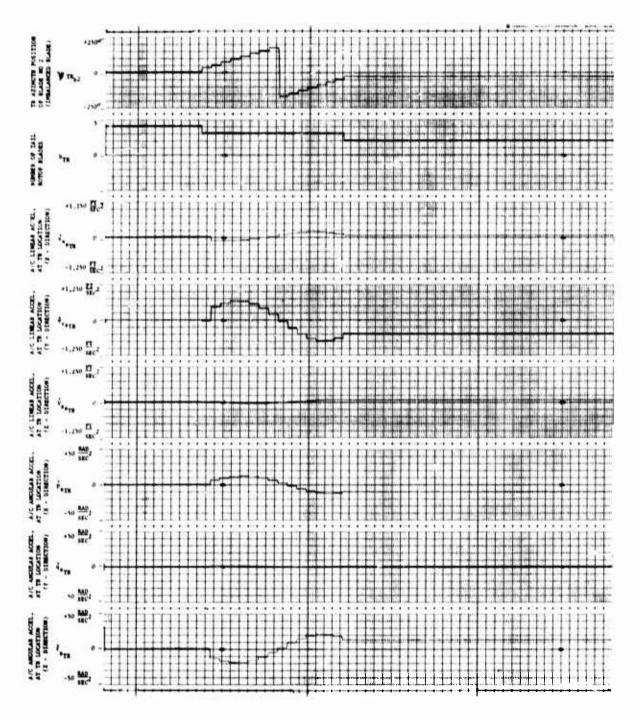


G.W.: 19,900 Lb. FSCG: 360.2 V: 150 Kts N_R : 100% SAS: ON H_D : 10,000 Ft

Figure D-6. (continued)

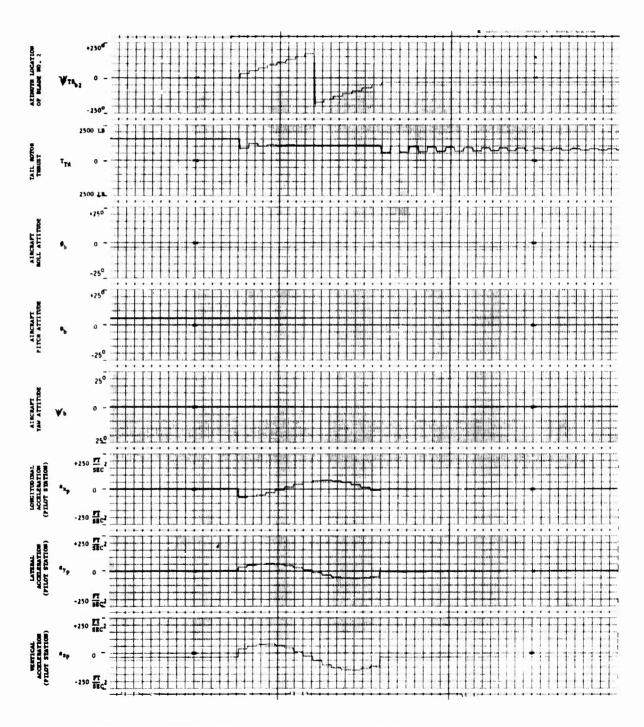
MEIGHT	19900.0	P8C6	364,2000		150.0	PSITRE	
1	5466.0	9074	245,8999	DELS	-5.6	VXSTR	
	49267.6	DIE	9,175600006-2	ASOUND	1877.0	VYSTR	0.0
	36224.0	7146	POSSOR	DELSHR	0.0	VZSTR.	
OPEGHR	27.0	1683	#*#	TERTER	-10.9	PSTR.	6.0
DHEGTR	124.55806	2002	5.0	THBITE	-18.0	HLVT	273.0
	15.	PASCNT	2163.0	MLHT	234.0	FSVT	695.0
	701, 1000	PH8	45.0	BVT	32.300000	DSTR	
LATBTK	Ň	A18	-4.6536977	LHI	9.48978681	XX	29.614630
LAGSTK	12.389178		14.658177	-		*	4.2219444
COI STK	210700 10	THETA	21 904077	THYSHE	410710 11	2	44 44460
PEDAI	11 442814	746770	29 480741	41744	14.189741	0	44 24.10.
XAX	2.41.44.5	W. B. T.	A2219AAA	XC7h	87777	×0 ×	5 106120
XBACTP	44156	YAACTT	A 4444584	97.50	6	0110	
1	261 46244	THETAB	4.48484	4466	A 66430 A	2000	
					4 4 4 4 4 4 4		
	1266660	05.00	200000	1100	9393137646	HOLE I	
	389C311C364	DE FAMP	4.4364893	1190		MATA	
1		SAMC		AABL	-11,633179	HITE	
	***	DHERAT	1.0	AA1L	#. 6638424S	3110	0.0
		PS1007		961L	0.16593356	RITE	5
ALFHF	-2.2709382	EATX	1,3658546	EKNEX	0,91466265	LHITA	0.0
CHITPP	61.144775	EKTZ	1,7418915	EKHFZ	1.0666552	DHITA	
EKTR	9.0	EPSHT	4.49038468	\$16HT	6,71077046	XITH	1,0
	60.986626	KONT	6,87177979	KOVT	9.82927212	VITE	9.6
MUXS	0,34938521	CT816	0.11199662	LTOT	-8.11933851	2178	0.0
MUYS	D.25613297E-1	CH 3 16	-B.22562351E-2	DTOT	24,307693	LITA	
MUZS	-8-19822189E+1	COMSIG	0.08488535E-6	118	2205.3241	MITA	6.0
LAMBHR	-8.32642178E-1	N2		TLAI	3168,9815	_	
DISTRE	0.13619981E-1	VC	8.89466967E=6	KTRBLK	0.1	AXP	-0 - 168697 ISE
	1386.2511	HBAR		VXBOOT	-2.296496B1E-1		-8.82419843E
	01135.7785	JOAR	1135.7785	VYBDOT	-8.35528698E-1		Ш
	M	TBAR	19161.365	VZBOOT	-0.32233896E-1		253.40214
	324	LBARH	-3119,5636	1	-8.27689234E-1	AYP	16.554529
	13818,483	TOVEL	-9485,6962		-8.33457210E-2	424	-0.51425602
	61199,689	DOAR	63329.476		-8-164986475-2	RSTR	
	3	T.X	-2.3355596	XTR		PATONE	-50.
	-644_44657	**	-313,32935		2072.4869	BTR	
		7.7	696.36215	278	-754-32283	MADO	
1	1947, 7641		-494-42751		12729.248	MADO	
	200 - 200	. F	10706.444		12117. 444	004	
	-4700 BAT-		A764 TARK	04.3	471 61677	7 A BB	
	13.517461	XVT	-15.853020	ALPHTT	-4.4348439	MADO	
1	7776biz 27						
		1 4 4		ALFVTT	4.1915888	LADO	6.0

Figure D-6. (continued)

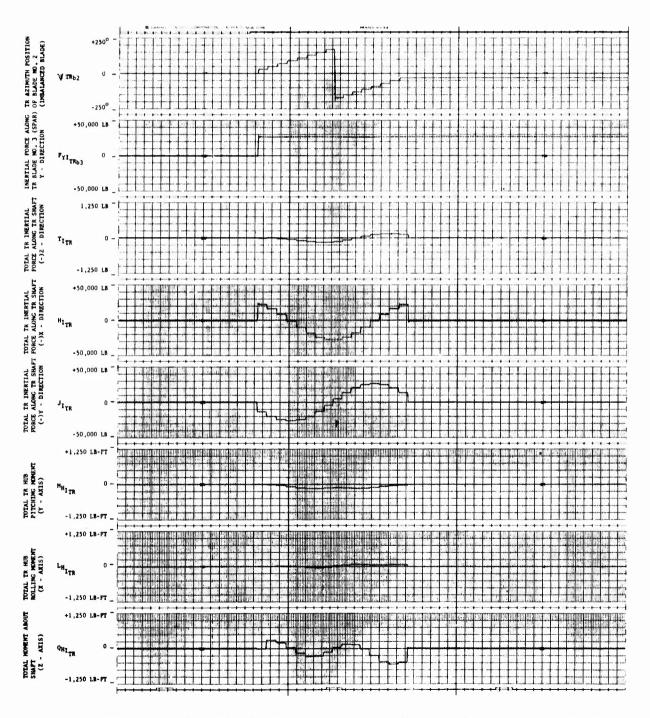


G.W.: 19,900 Lb. FSCG: 360.2 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft

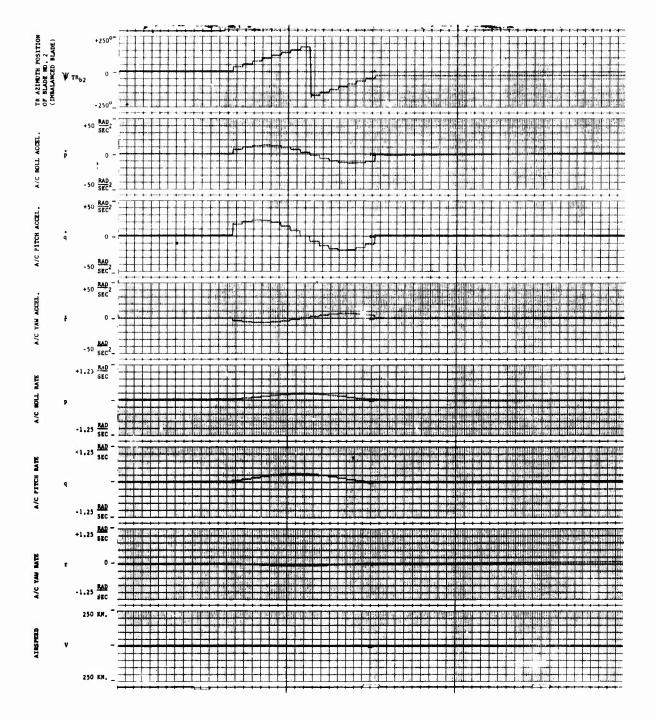
Figure D-7. Stepped Transition Time History



G.W.: 19,900 Lb. FSCG: 360.2 V: Hover N_R : 100% SAS: 0FF H_D : 10,000 Ft Figure D-7. (continued)



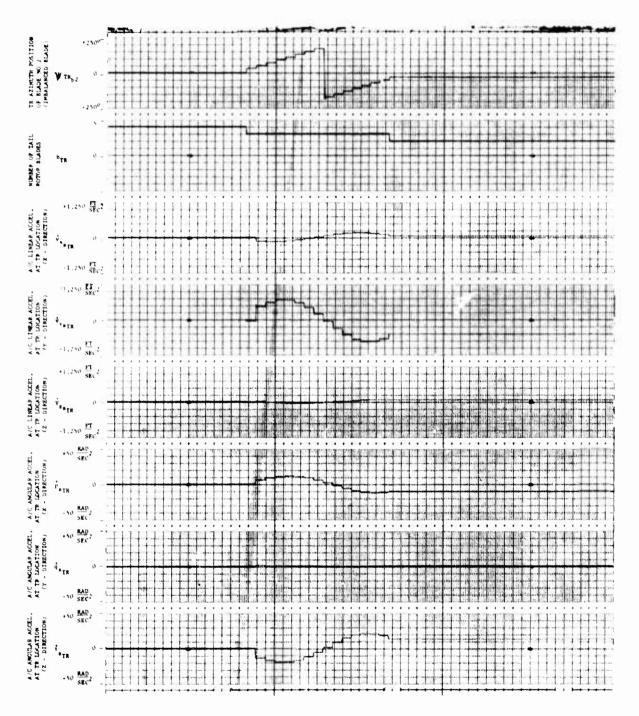
G.W.: 19,900 Lb. FSCG: 360.2 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-7. (continued)



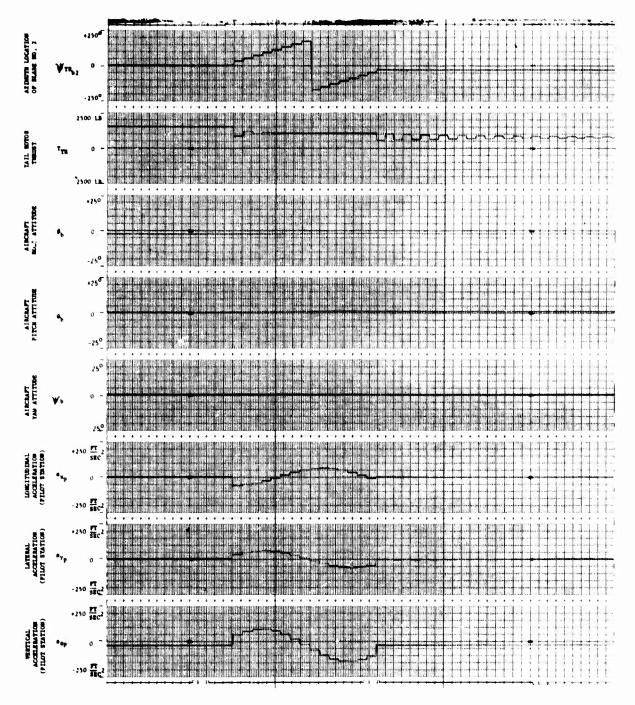
G.W.: 19,900 Lb. FSCG: 360.2 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft

Figure D-7. (continued)

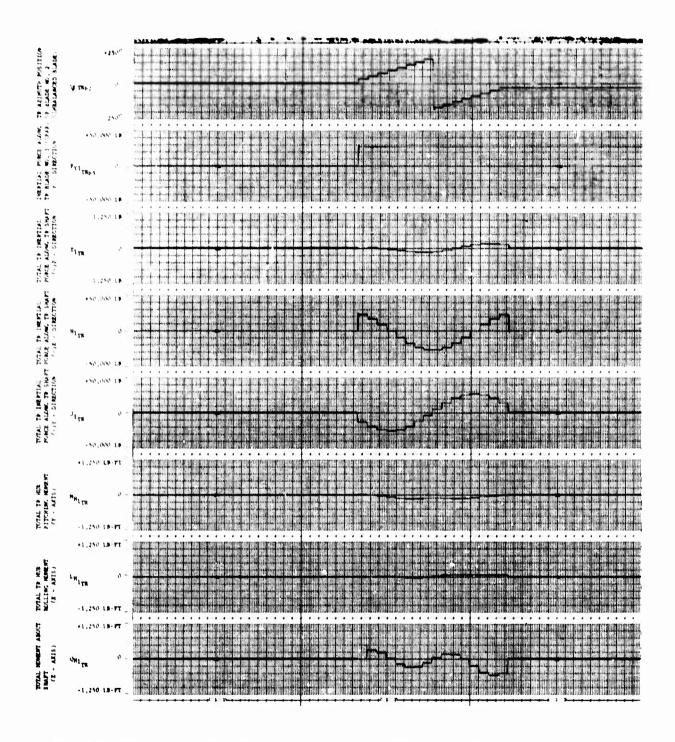
Figure D-7. (continued)



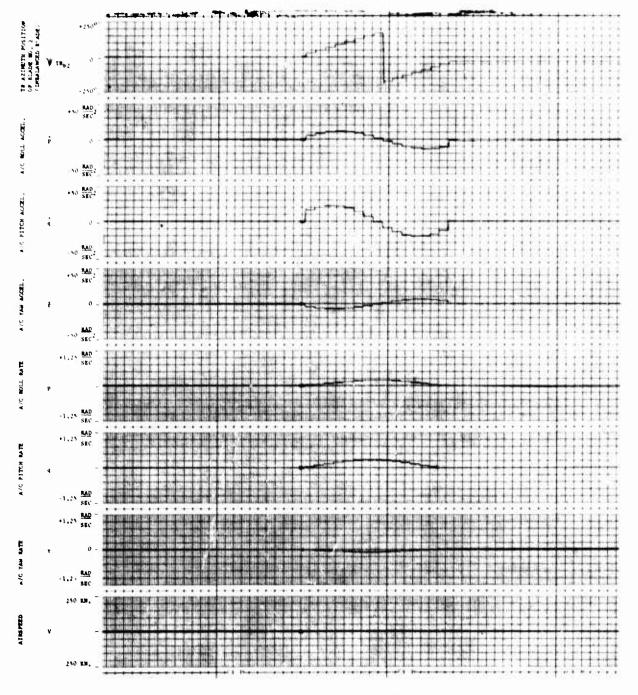
G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-8. Stepped Transition Time History



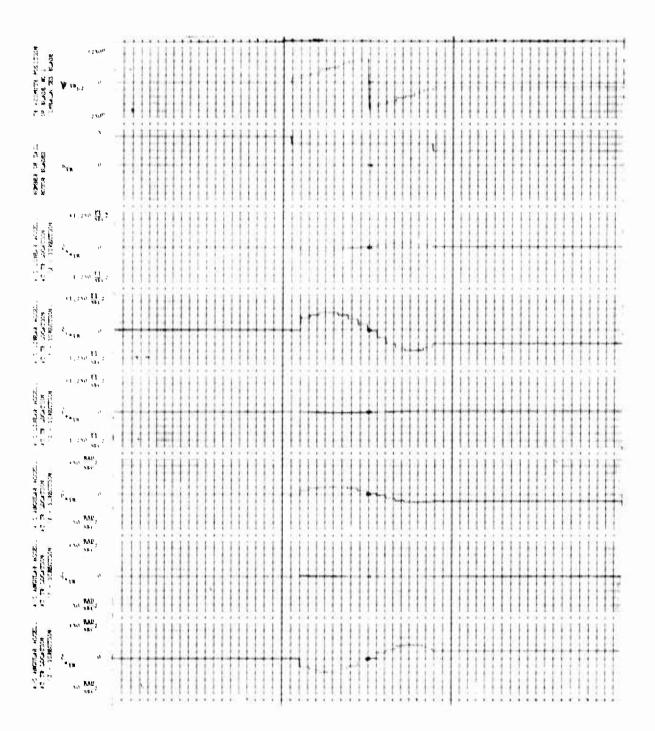
G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-8. (continued)



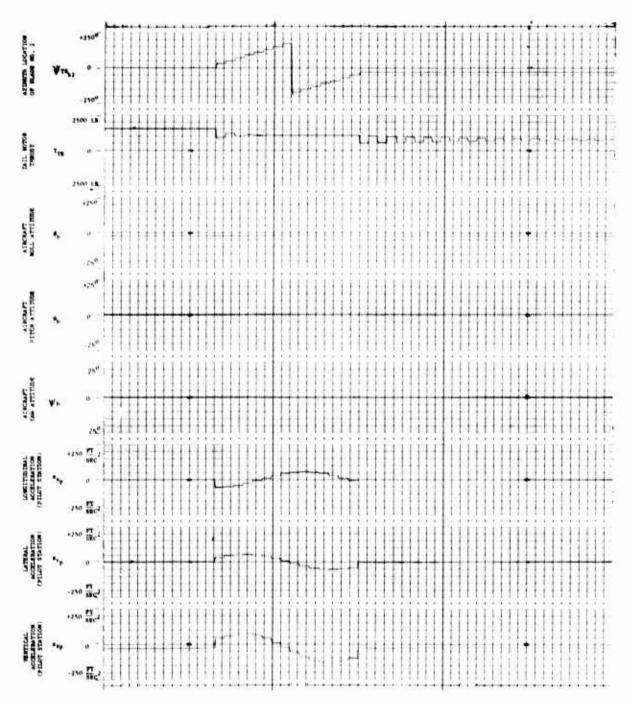
G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: 0FF H_D : 10,000 Ft Figure D-8. (continued)



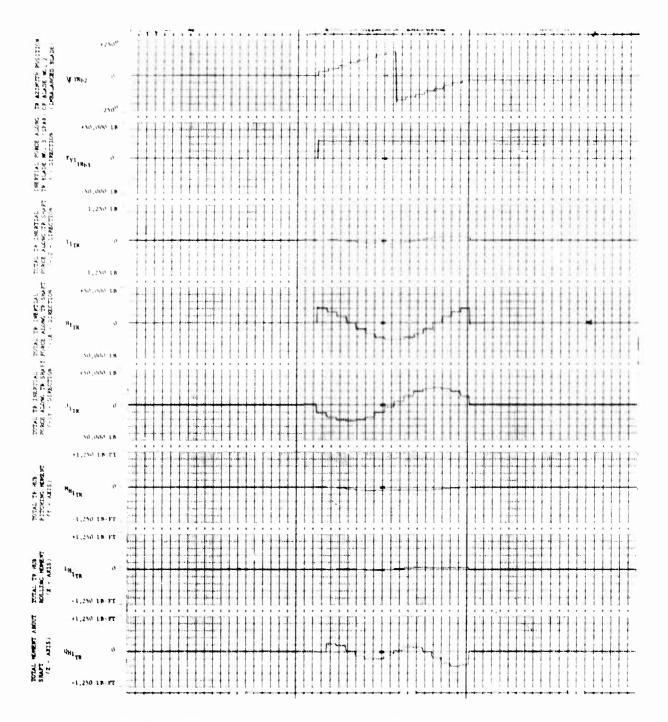
G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: 0FF H_D : 10,000 Ft Figure D-8. (continued)



G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 95% SAS: 0FF H_D : 10,000 Ft Figure D-9. Stepped Transition Time History

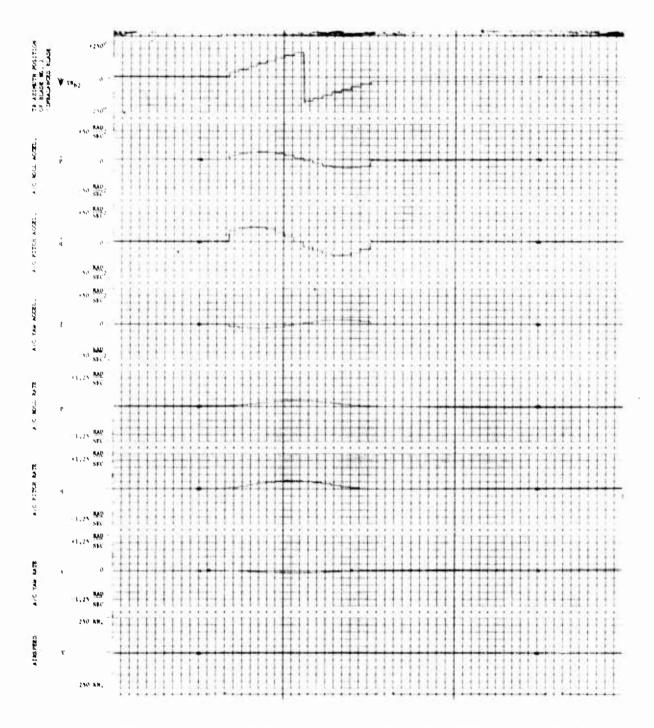


G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 95% SAS: 0FF H_D : 10,000 Ft Figure D-9. (continued)



G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 95% SAS: 0FF H_D : 10,000 Ft Figure D-9. (continued)

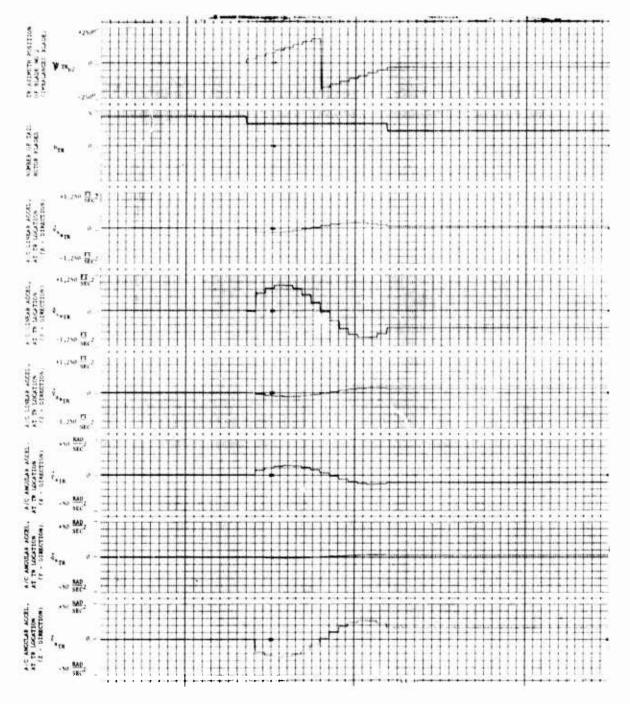
" " 1 " 11 "



G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 95% SAS: OFF H_D : 10,000 Ft Figure D-9. (continued)

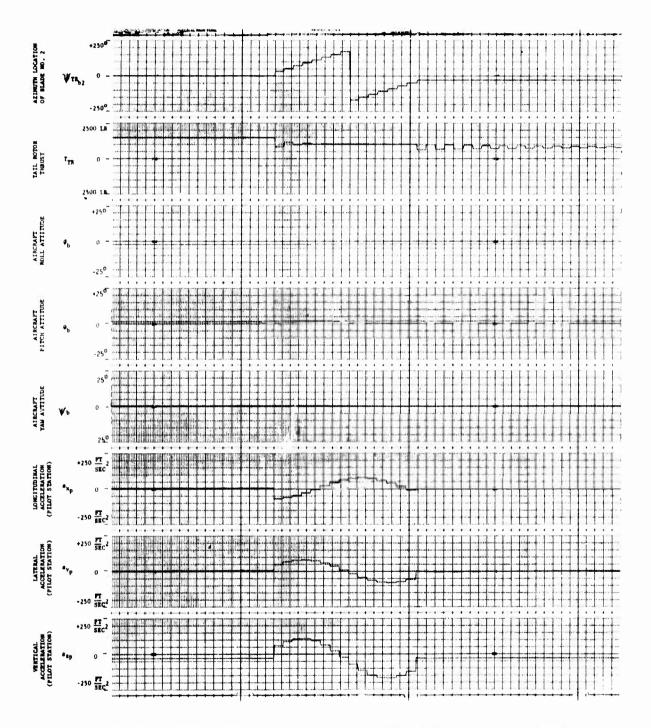
ME16HT	19900.0	FSCG	347.0	>	1.86-2	931189	6
	5468.0	90	245.89999	DEL 3		VXSTR	
	44287.9	OHO	9.1756999E=2	VSOUND	1977.9	VYSTP.	6
	38224.0	TIME	O. SUGURBORE	DEL 3MR	6	V797R	6
	25.649999	NBSS	6.4	TESTER	8	PSTR	6
	118.32250	N888	6.5	THSTTR	8	> 1	273.0
	15.0	PASCNT	1075.0	HH	234.0	F8VT	695.0
	789.48888	SHT	2	T > 8	32.300000	DSTR	
i	0.33622503	A18	-0.68736193	THI	34.8	XX	52.181496
	-1.3654426	818	-2.4827418	15		8	54.82485
	22.361426	THETAB	22.361426	THISMR	12.281424	X	74.258917
	21.748664	THETTR	36.006376	THISTR	22.586376	d ×	21.311988
	5.2181486	XBIN	5.4824889	XCIN	7.4258917	XPIN	1.1507688
	69.987916	XBACTI	6.9687816	RSTR	6	9160	6
1	O TABORASE	THETAR	70308194	4406	A ACOLOGI	94.00	0 0
	111111111111111111111111111111111111111		200000000000000000000000000000000000000		2 4172541	0	
1	200410410	2014	600000000000000000000000000000000000000		1153,1013	- 0	200
	8, 21878868E=3	7 7 7 7 7	9	100	1004100	*	
- 1	8.8	GAMC	50 · 50	AABL	-9,4808691	HITE	6.0
	8.0	OMGRAT	6.1	AA1L	-0.16935888	JITR	9.0
- 1		PSIDOT	6	881L	-0.16069345	MHITR	8.0
	-85,813533	EKTX	-0.53023976	EKNEX	0.21619007E-1	LHITR	8.8
- 1	2,7012518	EKTZ	0.72024825	EXXF2	0.29983885	OHITE	0.0
	8.0	EPSHT	66666675 8	SIGHT	0.0	XITE	0.0
	0,19399675	KOHT	1.0	KOVT	0.84852813	YITR	6.0
	8.24533431E.4	CT316	0,12049727	LTOT	.78.0	ZITR	0.0
	8.8	CHSIG	0.516252235-2	DIOT	45.079999	LITR	0.0
	-0.96721173E-6	COHSIG	8.11739630E-S	TTR	1526,2310	HITE	0.0
	-0.71846474E-1	71	0.99891798	HPIR	2170.2562	NITR	0.0
	8.71845507E-1	2>	0.11546071E-4	KTRBLK	8.7959999	AXA	0.39500075
	177.73051	HBAR	797.13925	VXBUOT	-0.22286597E-1	A Y P	1.3585385
i	-628.64868	JBAR	628.64868	VYBOOT	0.23042536E-1	AZA	-32.139716
	-18620.412	TBAR	18605.638	V2800T	0.17653953E-3	VXP	0.16896658E+1
1	-8508.0047	LBARH	-2602,3943	Poot	0.25993833E-3	d A A	0.0
	14896.892	HBARH	6581.1971	1000	-8.17789459E-3	VZP	0.21896820E-3
•	46015.352	OBAR	46535.709	ROOT	-0.26210617E-4	PSTR.	0.0
	12.904958	×	43.499392	XTR	6.0	PSIONG	-158.8
8		11	-0.34973258E=6	414	1434.2946	BTR	4.9
	9.7135899	7.7	65.393978	218	-522.04158	MADO	6
ŧ	6.6	[1	-0.78992567E-6	178	8609.4865	MADO	6.6
	-124.21451	ī	1968.9895	Œ	-16748.833	VADO.	
1	6.6	-	9.10143694E-4	AT.	-46016.954	7400	5
	43.499392	× ×	-0.621921892-7	ALFHTT	-19.652797	OOV	
							2
	6	404	A-38264604 P-	A1 5 0 7 7	6 6	001	

Figure D-9. (continued)

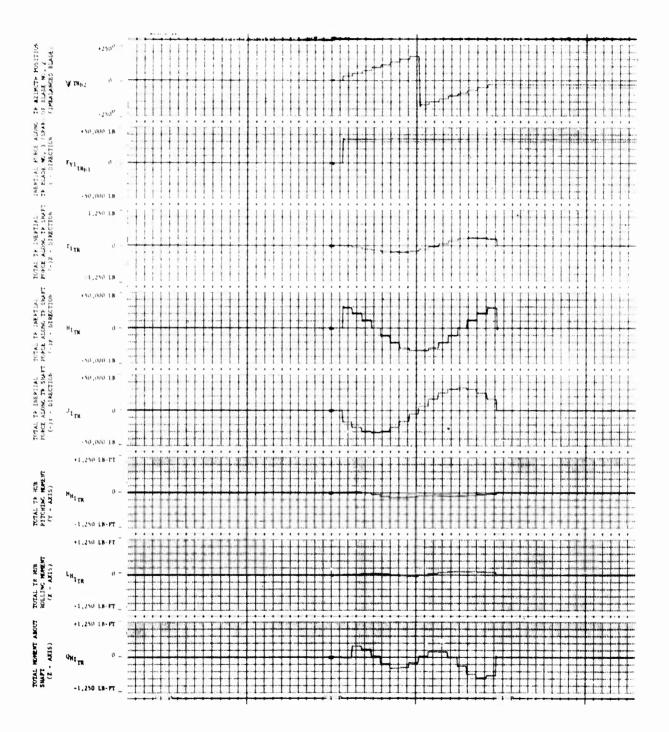


G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft

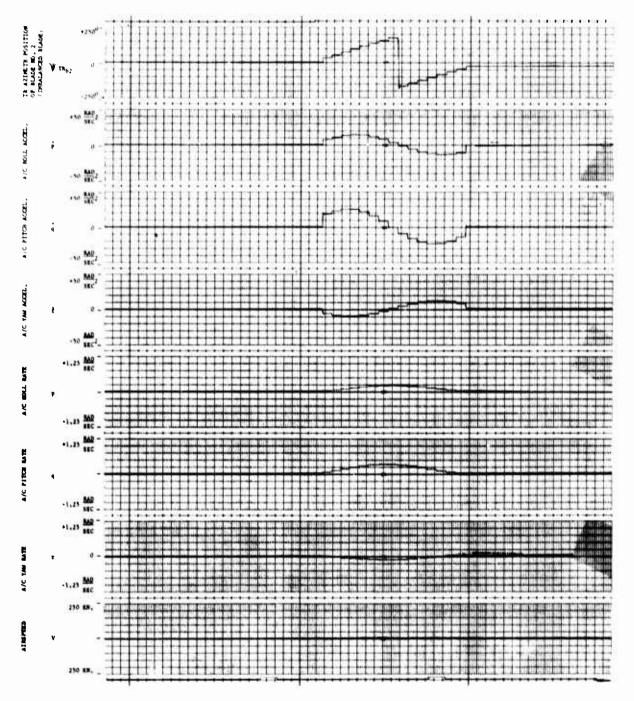
Figure D-10. Stepped Transition Time History



G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft Figure D-10. (continued)



G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: 0FF H_D :10,000 Ft Figure D-10. (continued)



G.W.: 19,900 Lb. FSCG: 347 V: Hover N_R : 100% SAS: OFF H_D : 10,000 Ft

Figure D-10. (continued)

94000							
	•	FSCG	347.0	>		PSITRE	9.6
THO	_	MLC6	1666	DELS	5.0	VXSTR.	6.6
Colored Colo	2.0		SOCOE	323000	67	VYSTR.	S .
Factor F	6800	NA SA		TELSOR		943184	
E-1 A34.0 THE A8 19 489481 A	80499	2000					
E=1 A18		PASCNT	797.0		234.8	FSVT	
E=1 A18	46689	SHT	. S. B		32.390908	OSTR	
THETAS 19.889481 TATSMR 9.329481 XC THETAS 19.889481 XC THETAS 19.889481 TATSMR 9.3294811 XC THETAS 19.889481 TATSMR 9.3294811 XC THETAS 19.899881 TATSMR 9.3294811 XC THETAS 19.899881 XC THETAS 19.899881 AAIR 19.899898 TATSMR 9.9199888 TATSMR 9.9199888 TATSMR 9.9199888 TATSMR 9.9199888 TATSMR 9.9199888 TATSMR 9.9199888 TATSMR 9.9199888 TATSMR 9.91998 TATSMR 9.91998 TATSMR 7.9199888 TATSMR 7.9199888 TATSMR 7.9199888 TATSMR 7.9199888 TATSMR 7.919989 TATSMR 7.9	P86578E-1	A13	. 6427217		30.095614	V.X	58.481291
THETAS 19, 489481 THYSAR 9, 3294811 KC KREIN S. 1594698 KCIN S. 5088758 KPIN KRACTI S. 6989523 THYSTR 16, 960423 KPIN KRIN S. 1594698 KCIN S. 5088758 KPIN E-3 EFTAMF S. 6 OHGRAT 1.8 OHGRAT 1.8 EKYTX -0.8 4969599 KCIN -0.9 50886875 TITR EKYTX -0.8 4969599 KCIN -0.9 50886875 TITR EKYTX -0.8 4969599 KCIN -0.9 50886975 TITR EKYTX -0.8 4969599 KCIN CONSTRUCT	14422245	818	P1.8489368		0.54	E ×	A
### ### ##############################	109401	THETAB	19.489481	THISHR	9.3294011	×	55.000758
Colored Colo	111142	THETTR	30,426423	TH75TR	16.926423	d X	26.959573
Colora C	16751	X DIN	5,1569698	KCIN	5.5686758	KPIN	1.4557791
E-1 THETAB 1-1575311 AABF 3-5198866 057R E-3 GETAMF 8-2-514668 AA1F 1-9984993 TITR GARC 8-8 BAC SA AA1C 8-8-92866475 TITR GARC 8-8 BAC SA AA1C 8-8-92866475 TITR GARC 8-8-92866475 TITR FRIT 8-8-92866976 FRIFZ 8-8-929668-1 HITR FRIZ 8-8-92966999 BIGHT 8-8-929668-1 HITR FRAIT 1-8-929592792E-2 DIOT 45-87999 FRIET 8-8-9296676-1 HITR FRAIT 8-9996999 FRIEZ 8-8-929999 FITR FRAIT 8-9996999 FRIEZ 8-8-929999 FITR FRAIT 8-9996999 FRIEZ 8-99999 FITR FRAIT 1-9-2792712 VX8DOT 8-299999 FITR FRAIT 8-9996999 FRIEZ 8-99999 FITR FRAIT 8-9996999 FRIEZ 8-99999 FITR FRAIT 1-9-2792712 VX8DOT 8-299999 FITR FRAIT 1-9-2792712 VX8DOT 8-299999 FITR FRAIT 8-999999 FRIEZ 8-99999 FITR FRAIT 8-9996999 FRIEZ 8-999999 FITR FRAIT 8-9999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-9999999 FRIEZ 8-9999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 RIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-99999 FRIEZ 8-999999 FRIEZ 8-999999 FRIEZ 8-99999 FRIEZ 8-99999 FRIEZ 8-99999 FRIE	188691	XBACTI	5.6989551			2100	6
E-S GETANT B.8 GANC B.8	16	THETAB	1,1575311		3.5198866	-	
E-M GETANT B.0 GANC GANC GANC GANC GANC GANC AAAI -0.0025066475 ILTR FRICAT -0.0025066475 ILTR FRICAT -0.00250666475 ILTR FRICAT FRIC		PHIB	-2.5714688		1,9989193		6
CAMC 8.0 ONGRAT 1.0 PRIOR 1.0 PRIOR 2 PRIOR 2 PRIOR 2 PRIOR 6.0 PRIOR	~	BETANE	9.0		-8.98486475		9.0
Cheral 1.0		GAMC	0.0	į	-6.3684695		8.8
ENTX		CHERAT	1.0		-0.9258UPB7E-1		8.8
EXTX -0.4966364 EXHFX 8.161192485.1 LHITR EAST 0.0 22362431 DAILTR EAST 0.0 22362431 DAILTR EAST 0.0 22362431 DAILTR EAST 0.0 22362431 DAILTR EAST 0.0 22362431 DAILTR EAST 0.0 22362431 DAILTR EAST 0.0 23952495 EAST 0.0 235623449 EAST 0.0 24693449 EAST 0.0 246		PSIOOT		- 1	-8.93195589E-1		
E-4 CHEL B.4499999 BIGNT B.84852431 XITR XONT B.84852413 XITR XONT B.84852413 XITR XITR XONT B.84852413 XITR XITR XONT B.84852413 XITR XITR XONT B.894894 XITR XITR B.894894 XITR XITR B.894894 XITR XITR B.894894 X	•	EKTX	-8.49686364		E.16119244E-1		8.8
E-4 CT816 F.99999	149856	EKTZ	8.63757979	- 1	0.22365451		8
E-4 CT816 F.986266E-1 LT07 -78.5 CT816 F.986266E-1 LT07 -78.5 CT816 B.29592492E-2 D107 45.87999 L17R LS CDM816 B.295632E-6 TTR 139.8.9199 L17R LS CDM816 B.295632E-6 TTR 159.8.9199 LT B.295632E-6 TTR 159.8.9199 LT B.295633E-4 KTR8LK B.79599999 AXP LBAR 18637.849 VX800T B.25959999 AXP LBAR 18637.849 VX800T B.25959999 AXP LBAR 18637.849 PD0T B.396986-2 AXP LBAR 18637.849 PD0T B.396938E-4 VXP LBAR 18637.849 PD0T B.396938E-4 VXP LT B.88.8776572E-6 VTR 1310.8687 BTR LT 152.49398 ATR 1310.8687 BTR LT 152.49398 ATR 1310.8687 BTR LT 153.7598986E-6 LTR 8851.5233 XADD NT B.1843235E-7 ALFMT -22.834167 NADD VVT -8.34976672E-6 ALFWTT 8.8	047	EPSET	D.4499999		6.0		9.0
E-4 CTRIG F.98624968E.1 LTOT +78.8 4E-6 COMBIG B.29592792E.2 TTR 135999 LITTR AS.879999 LITTR AS.87999999 LITTR AS.8799999 LITTR AS.8799999 LITTR AS.8799999 LITTR AS.8799999 LITTR AS.8799999 LITTR AS.87999999 LITTR AS.879999 LITTR AS.8799999 LITTR AS.8799999 LADOUR N.T. B.37976672E.6 LTR AS.87977777777777777777777777777777777777	9611699	KOHT	1.9	ł	9.84852813		8.6
### COURTS #### ###############################	1936625-4	C1810	F.98F2946BE-1		-18.5		8 .
### COMBIG #.6356932E=6 TTR 1394.9149 HITR PROPERTY		CHRIG	B.29592792E-2		45.87999		
E=1 N2 B-995448	9-3016419	COMBIG	8.63356932E-6		1394,9144		6.0
Mark	2188338E-1	MZ	0.99963440		2296.3993		8
MARR	E-1	3	9.21618988E-4		0.7959999		8.67521553
JBAR 466,82112 VYBDDT -8,88384862E-2 AZP TDAR 18637.649 VZBDDT 6,5966938E-4 VXP LBARN -3868,9148 PDDT 8,194946E-3 VYP HBARN 6255,914 RDDT 7,1919946E-3 VZP AT -8,34976672E-6 VTR 1310,8687 BTR AT 52,494398 ZTR -477,12541 HADD AT 1575,78896E-6 LTR 8851,5233 XADD AT 8,18143233E-7 ALFHT -22,834167 NADD VV -8,441643E-7 ALFHT -22,834167 NADD VV -8,34976672E-6 ALFVT 8,8		HOVE	612,62941		0.25858996E-1		1,4326575
TOAR 18637.649		JBAR	669.92712		-8.88384862E-2		-32,129302
LBARH		TOAR	18637.649		E.59669538E-4		B.16894791E-1
MBARM 6590.1334 DDD1 F.19199364E-3 YZP DBAR 42525.914 RDD1 B.51938199E-3 RSTR. XI 58.879456 XXR B.8 B PSIDMG VI -8.34976672E-6 YTR 1318.8687 BTR ZI 52.494398 ZTR -477.12541 MADD LI -8.7898986E-6 LTR 8851.5233 XADD MI 1575.7598 MTR -15397.773 YADD XVY -8.6441643E-7 ALFMTT -22.834167 NADD VVT -8.34976672E-6 ALFVTT 8.8 LADD	_	LBARH	-3868,9148		0.184871336-3		8.8
### ##################################	88.949	HBARH	6590.1334		F-19199364E-3		8.34138616E-3
XT 58.879456 XTR 8.0 PSIDMG YT -8.34976672E-6 YTR 1310.8687 BTR 2T 52.494396 2TR -477.12541 HADD LT -8.7698986E-6 LTR 8851.5233 XADD HT 4575.7588 ATR -15391.773 YADD XYT -8.54416438E-7 ALFHTT -22.834167 NADD YYT -8.34976672E-6 ALFYTT 8.0 LADD	-	OBAR	42525.914		9.51938199E-3		9.6
27	_	XT	38.979456	į	9.	DHO	-156.8
ZT 52.444398 ZTR -477.12541 MADD MADD MT 1575.7588 MTR -15397.273 XADD MT 8.181432335.4 MTR -15397.273 YADD XYT -8.64164235.7 ALFHTT -22.634167 MADD VYT -8.349766725.6 ALFYTT 8.8 MADD MADD		¥	-8.34976672E-6		1318,6667		4.0
LT -0.7696966=6 LTR 8651.5233 XADD MT 1575.7866 MTR -15397.773 YADD NT 0.18143235E=4 NTR -42657.661 ZADD XYT -0.64416425E=7 ALFMTT -22.634167 NAOD YYT -0.34976672E=7 ALFMTT -22.634167 NAOD	163690		52.494398	æ	-477.12541		8.8
NT 8.1814323355-4 NTP -42057,661 ZADD XVT -8.644164235-7 ALFHTT -22.634167 NAOD VVT -8.349766725-6 ALFVTT 8.8		-	-9.78988986E-6		8851,5233		
NT 0.10143233E-4 NTA -42057.601 ZADD 0 XVT -0.64416423E-7 ALFHTT -22.634167 NADD 0 VVT -0.34976672E-6 ALFVTT 0.0	.238111	HT	1575.7860		-15397-773		5
XYT -0.64416423E-7 ALFHIT -22.834167 NADD 8		-	8.18143235E-4		-42057,681	ZADD	8.8
YVT -0.34976672E-6 ALFVTT 0.0	179456	XYT	-0.64416423E-7	LFHTT	-22,834167	NADD	
		¥ > T	-8.34976672E-6	LFVTT	9.0	-	

Figure D-10. (continued)

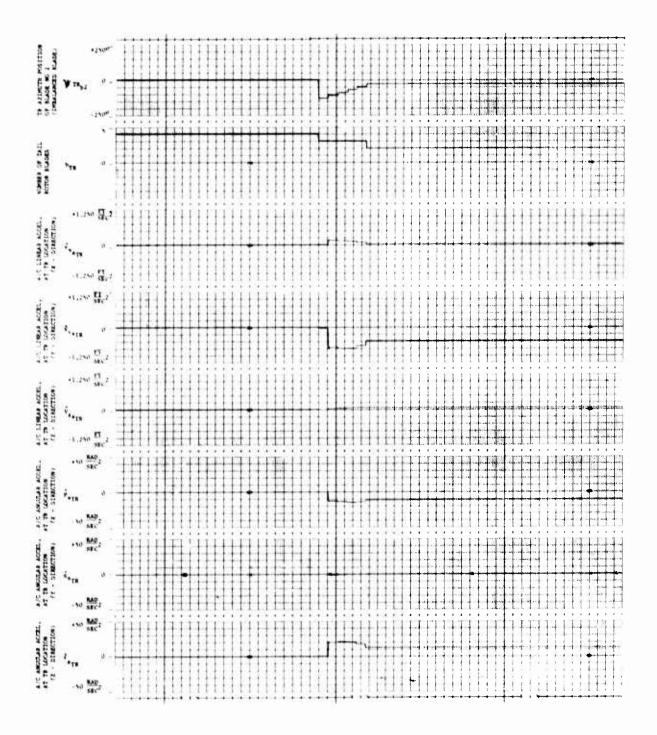


Figure D-11. Stepped Transition Time History, D

Damage = 50°

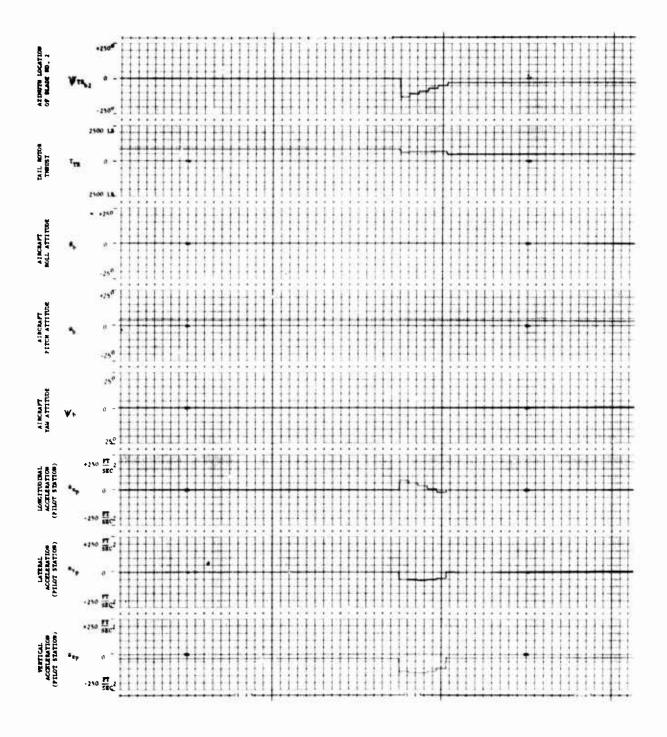


Figure D-11. (continued)

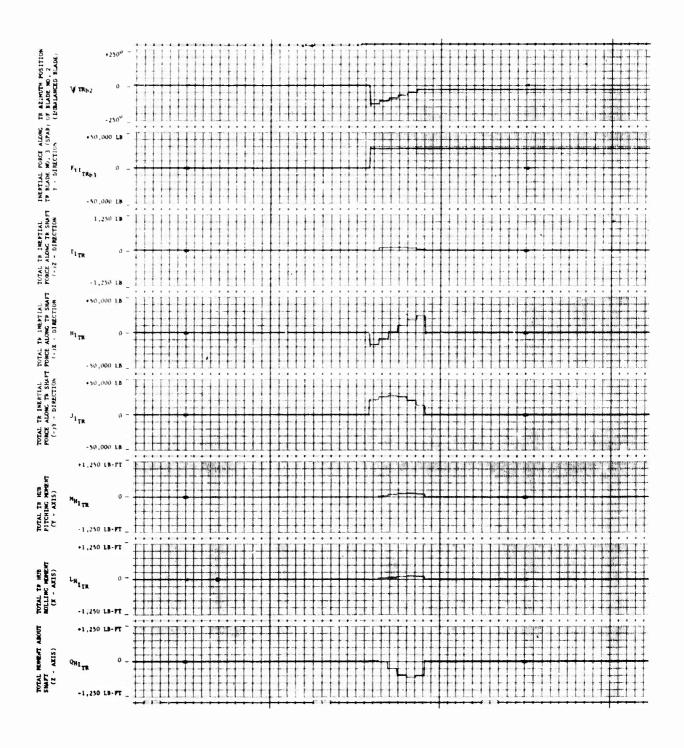


Figure D-11. (continued)

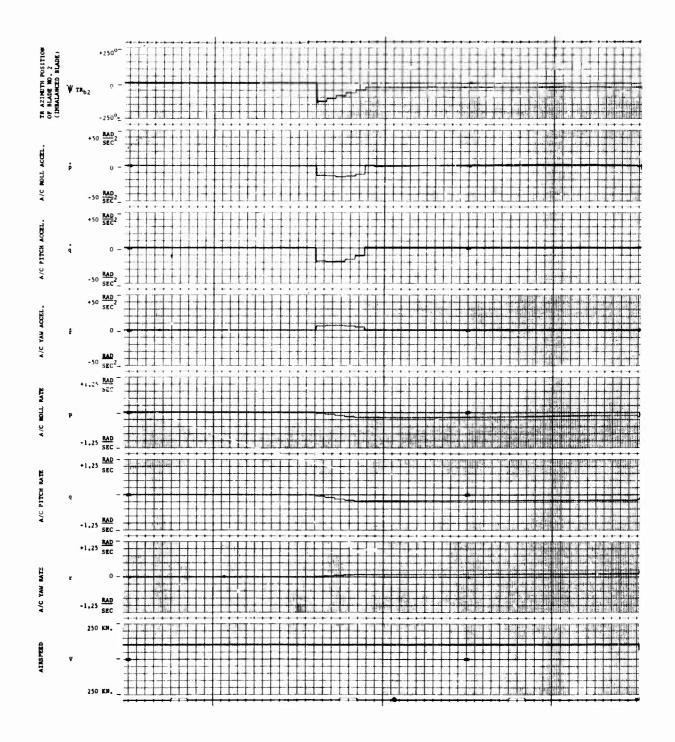


Figure D-11. (continued)

		B + 1																									-									
			9.6	9.6	173.8	195.0	9.0	16,298737	19,04661	37,781963	59,598654	-	6.9										0.0	•	D 6	2.0642155	-8.244862698-	32,183912	100,03003	10.047795		50.6				
PSITRE	VXSTR.							ì				į		DSTR	200	111	HITE	ŀ			X I THE				MITA			1	- [424		9	1	i	AADD	
100.0	-3.3	1116.0	9.0	-18.8	-18.0	234.0	32.366986	-2,7695552	-3.6	6.4451148	4.9412928	3,7701963		3,1814497	-5.6861862	8.58649945Eet	-4.1988659	9.24187694	E,3628885	# 14084118	2,64751979	8,63188377	3,4638934	24,481868	1131,2135	8 1	8,151842486-2	-8,143688178-1	2-216486666	9.20261696E-2	-6:297494806-2		364,62487	4411.4717		
>	DELS	ASOUND	DELBMR	Į.	_	* 7 * 4		LIM		1		KCIN		İ					5516		- SIGHT			5707	# T D	KTRBLK		1			- 1		* ·		a a	
360,29000	245.19999	0.23788888E-2	9.20085989E-1	2		1217.0	45.0	-1.2161287	8.8851668	16,525114	16.441292	1.00000	2,1024486	3.688882		3.6798895				1,586,56	1.48679889	0.07177979	0.67878526E=1	-9.47506950E-2	6.44692652256	B.71525573E=5	-1180,6896	254,36919	2561 6161	-15256,386	23863,979	-23,053153	178,51212	0395,41927	13103,227	
F3C6	#LC6	DH	TIME	Kosa	Ness	PASCHT	DH4	A18	818	THETAB	THETTA	Z I O X	XBACTI	THETAB	PHIB	RETAMP	CAMC	DHORAT	LOGISA	E	FP 547	THON	CTSIG	CHSIG	919W97	AC	HBAR	2048	HOVE	HOVE	DBAR	± X	 		=	
16450.8	4333.0	36515.0	37363.0	27.0	124,55888		700.4098	8.59348	•	16,525114	11.107155		21,024400	599	-	1.017			100000000000000000000000000000000000000	A2 461646		37,429361	0,23325674	0.157251122-1	-8.975394948-2	8,12522958E=1	1922,0494	*254,36815	407	2	22587,512	-093,68486	-361.12768	546, 64727	-3392,2619	
MEIGHT	ž	<u>.</u>	71	KERKE	DACETH		BHL	ATSTR	NGSTK	COLSTA	EDAL	MAK	BACTP	9 × ×		974		.	A	17.10	EKTR	1=0	MUXS	245	2079 2079	DESHAR	KHH			II	MAN	M. H.	414	117	1	

Figure D-11. (continued)

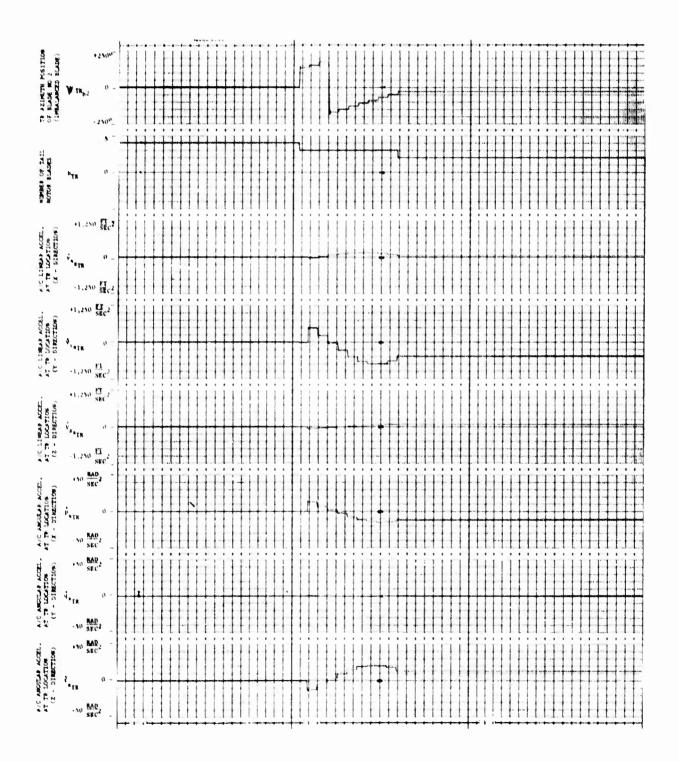


Figure D-12. Stepped Transition Time History.

Damage = -50°

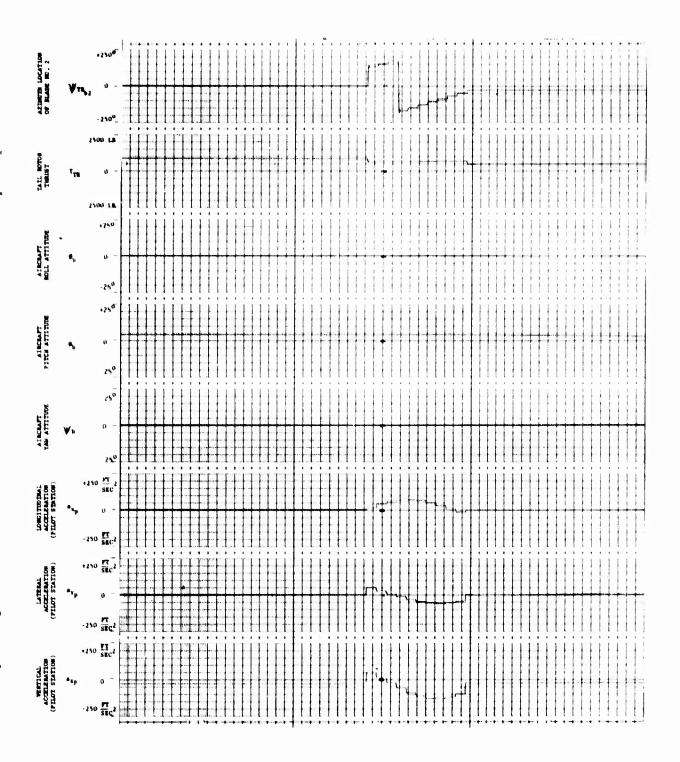


Figure D-12. (continued)

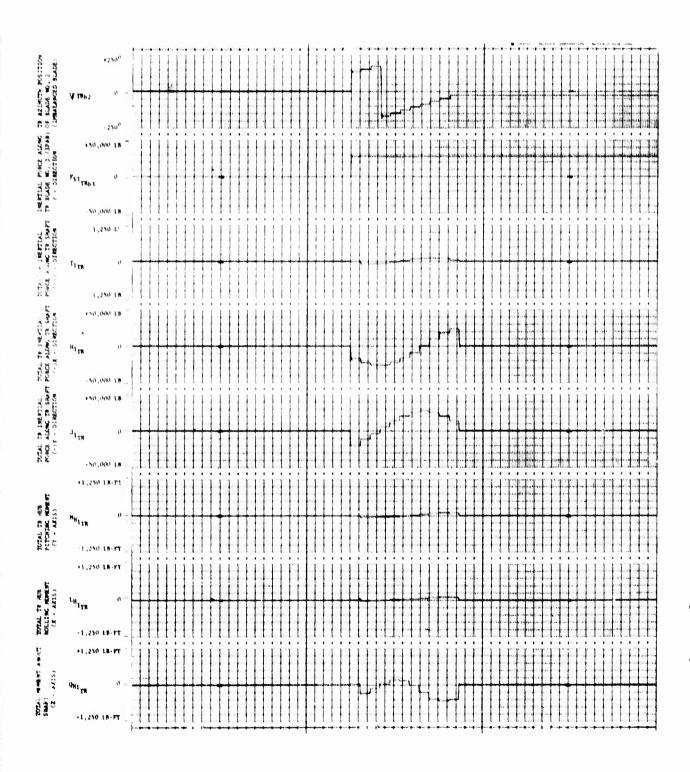


Figure D-12. (continued)

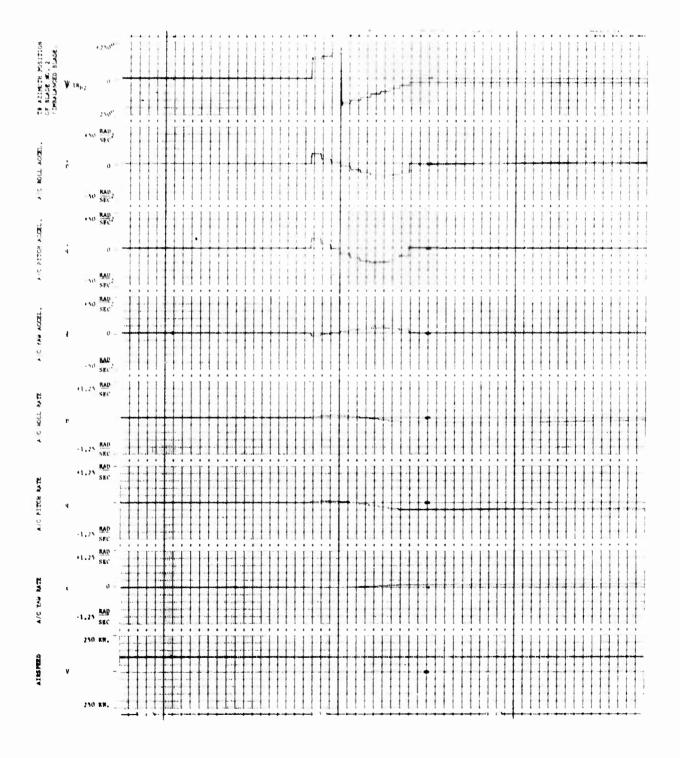


Figure D-12. (continued)

	UTTAS(576) 1	1-21-77	18 27-NOV-77	;	BUN 11.			
WEIGHT	16659	908	342 . Vagase	>	5.66	241182		
=	-	PLC6	245,69999	DELS		VESTR		
<u>_</u>	38515.0	BHO	.23788890E	V SOUND		VYSTR	6	
21	37363.0	TIME	INBUGE-1	DELBAR	60	VZSTR.	6	
DEFORE	27.0	NESS	6.	TENTE .	18	PSTR	8.8	
DHEGTR	124,55660	NOON		TESTA	-18.0	HLVT	273.0	
KFR	5.1	PASCNT	217.0	HLHT	•	FSVT	95.	1
PBHT	B. 48868		5.0	BVT	32,398988	OSTR.	•	
LATSTK	-0.59346201	-4	1.2161287	THI	-2,7695562	XX	46,298737	
LVCSTR	727	810	. 9651668	13	-3.8	×	19.896889	
COLSTR		THETAB		THISHE	6,4451148	×C	-	
PEDAL	11,167155	THETTH	6.441292	THISTR	4.9412920	ď×	50.590054	
MATH	.62987	KBIN	٠.	XCIN	3,7781963	4	2,7317919	
XBACTP	•	XBACTI	.102440	RSTR.		PSTR	0.6	
0×A	•	THETAB	3,6880852	AABF		OSTR		
VYB	11,391428	PHIB		AA1F	-5.6861862	RSTR		
82A		RETAMP	3.6798895	1196		TITA	60	
•		GAMC	2.6	AAGL	-4.1968659	HITR	6	
0		DMCRAT	1.3		2418769	JITR		
•	•	PRIDOT	6.	RBIL	9.3628884S			
÷	.55888	EKTX	1.3847468	EKMFX	9268499	LHITR		
CALTER	. 39384	EK72	1.6490261	EKMF2	1,3879961	-	6.6	
ERTR	•	E SML	6.48579889	SIGHT	6475197	- KITA-	6.9	
¥10	7.42936	KINHT	0.87177979	KOVT	0.63160377	YITR		
MUXS	~	C7916	0.67879S26E-1	101	3.4638934	ZITR		
MUVS	15725112E-	CHBIG	-8.47596958E-2	DIOT	24,481968	LITA	6.0	
820H	.276988	COMSIG	8.49892652E-6	4-1-10	837,84561	FITE		
LANDHR	0.97539494E	74		T Q T	1133,2135	RITH	6.0	
DESTER	125229586=	NC OA	0.71525573E=9-		1.9	AXP	2.8642159	
R I	25.0494	KON	-1187.6898		9.15184242E-2	AVP	-0.24466269E-1	
* XX	361	JBAR	254,36815		-8.14368817E-1	AZV	-32,103912	
ZHR	-15644,553	TOAR	15724,929	V2900T	B.23989519E-2	AXA	168.63883	
CHA	393	LBARH	-2261,9183	b.	-8.958589656-2	447	39142	
a I I	-842,14135	HUARK	-15256,386			424	8.64779	
MEN	•	DBAR	23883,979	POOT	2-3081	RSTR.	9.6	
MEN	5	¥.	-23,853153	X L X		PSIDME	•50.6	
424	1276		-178-11212	410	86,62437	BTR		
4=2	130	7.1	464,63322	278	6,30838	MADO		
-	6.64727	- 1	-395.81927	172	=	MADO	E . E	
LXI	392,261	F	13103,227	2 - I		YADO	6	
N X X	194.718	- X	4978,2169	*	43	ZADD		1
- I	3,1772	X V T	•	ALFHTT	-4.7546483	MADD	6	
- X	114551	↓ ∧ ↓	:	ALFVTT	3.8617782	LA00		
241	464,26319	ZVT	9.37883121	A4981F	686897			
	ı							

Figure D-12. (continued)